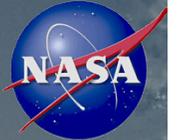


International Space Station External Payload Accommodations



Earth Venture 2 Workshop
4 March, 2011

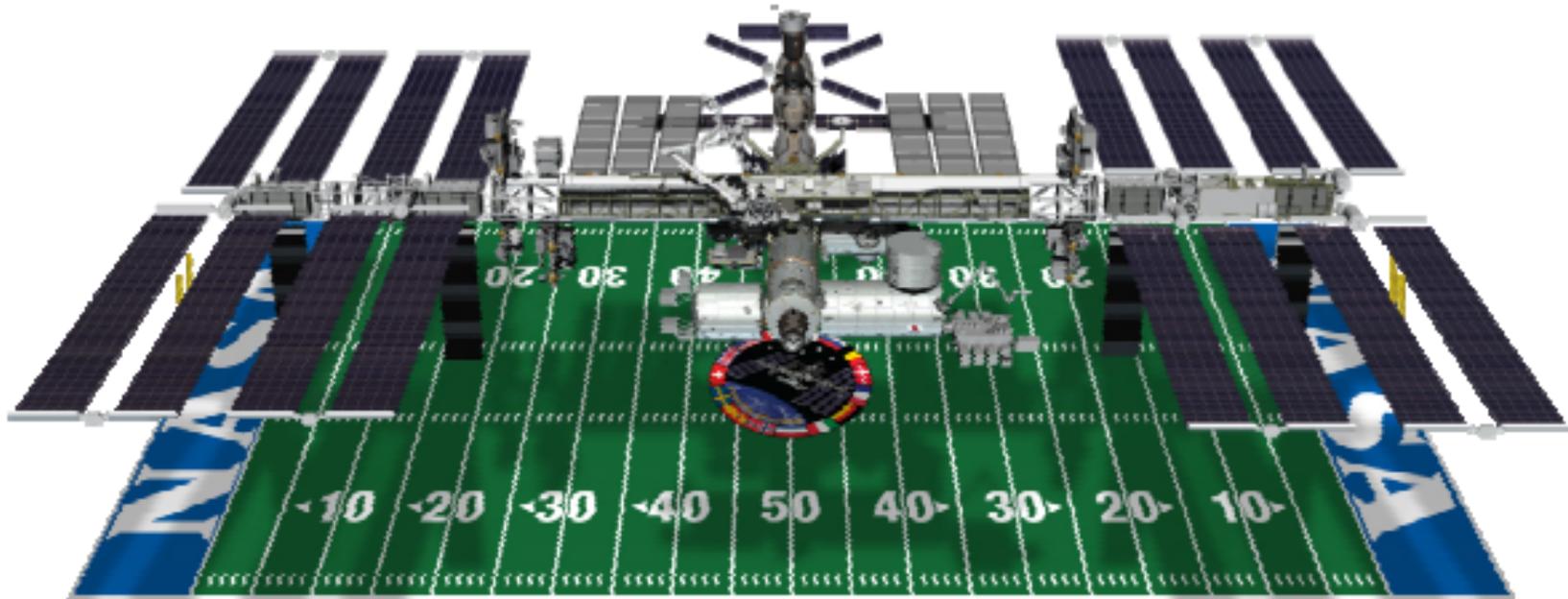


Al Holt

ISS Technology Demonstration Office
Space Station Payloads Office



International Space Station Facts



Spacecraft Mass: 799,046 lb (362,441 kg)

Velocity: 17,500 mph (28,200 kph)

Altitude: 220 miles above Earth

Power: 80 kW continuous

**Science Capability: Laboratories from four international space agencies –
US, Europe, Japan, and Russia**



Window Observational Research Facility (WORF)



US Laboratory Window

50-cm diameter

Telescope-quality optical glass

NADIR view



WORF Rack

Facility to support visual and multispectral remote sensing using Lab Optical Window



Windows on the Earth



Service Module Window
40-cm diameter
NADIR view



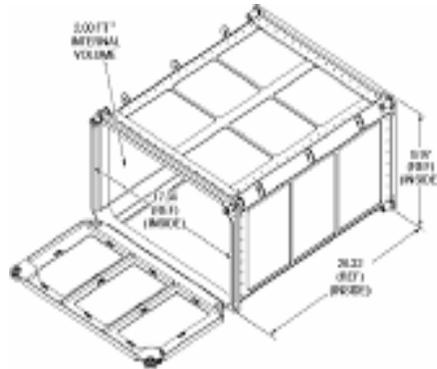




EXPRESS Racks *(Expedite the Processing of Experiments for SpaceStation)*



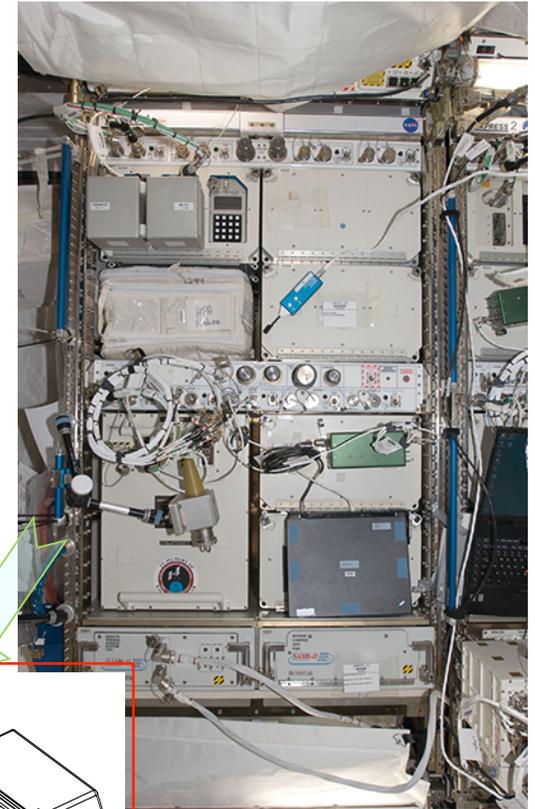
Middeck Locker



Features

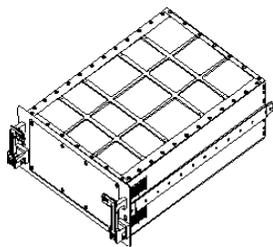
- 4 rear captive fastener attachments
- Friction hinge
- Dual door locks
- Installation tool guides on 4 corners
- Weight – 12 lbs

Sub Rack size payload capability with standard utilities such as power, data, cooling and gases



EXPRESS Rack

International Sub rack Interface Standard Drawer



Features

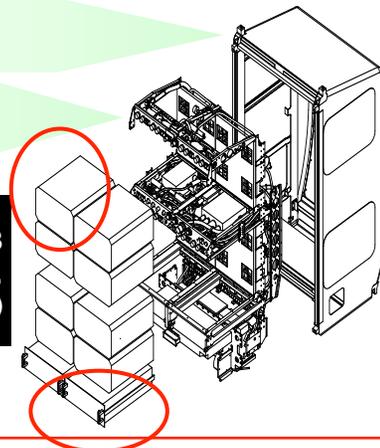
- 4 PU (Panel Unit)
- Blind Connectors
- Locking Handles
- Weight – 27 lbs
- Rated to at least 37 lbs

EXPRESS 8/2 Configuration

International Standard Payload Rack

Secondary Structure & Subsystems

8/2 Payload Configuration (8 Middeck Lockers, 2 Powered ISIS Drawers)





ExPRESS Rack Resources

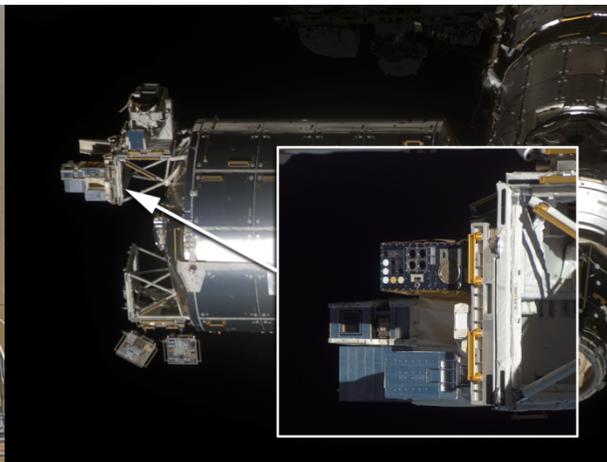
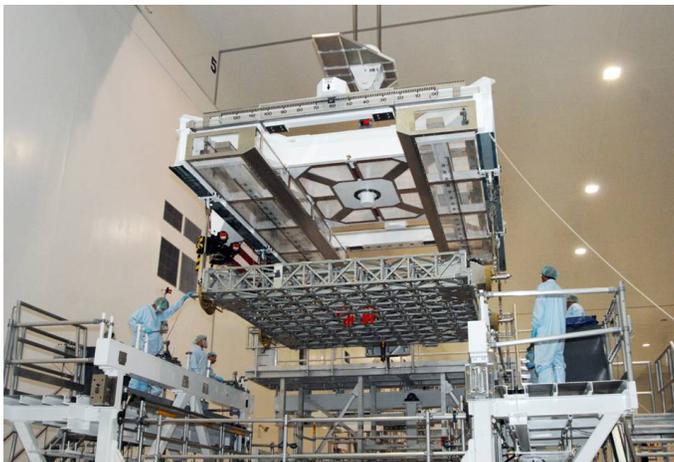
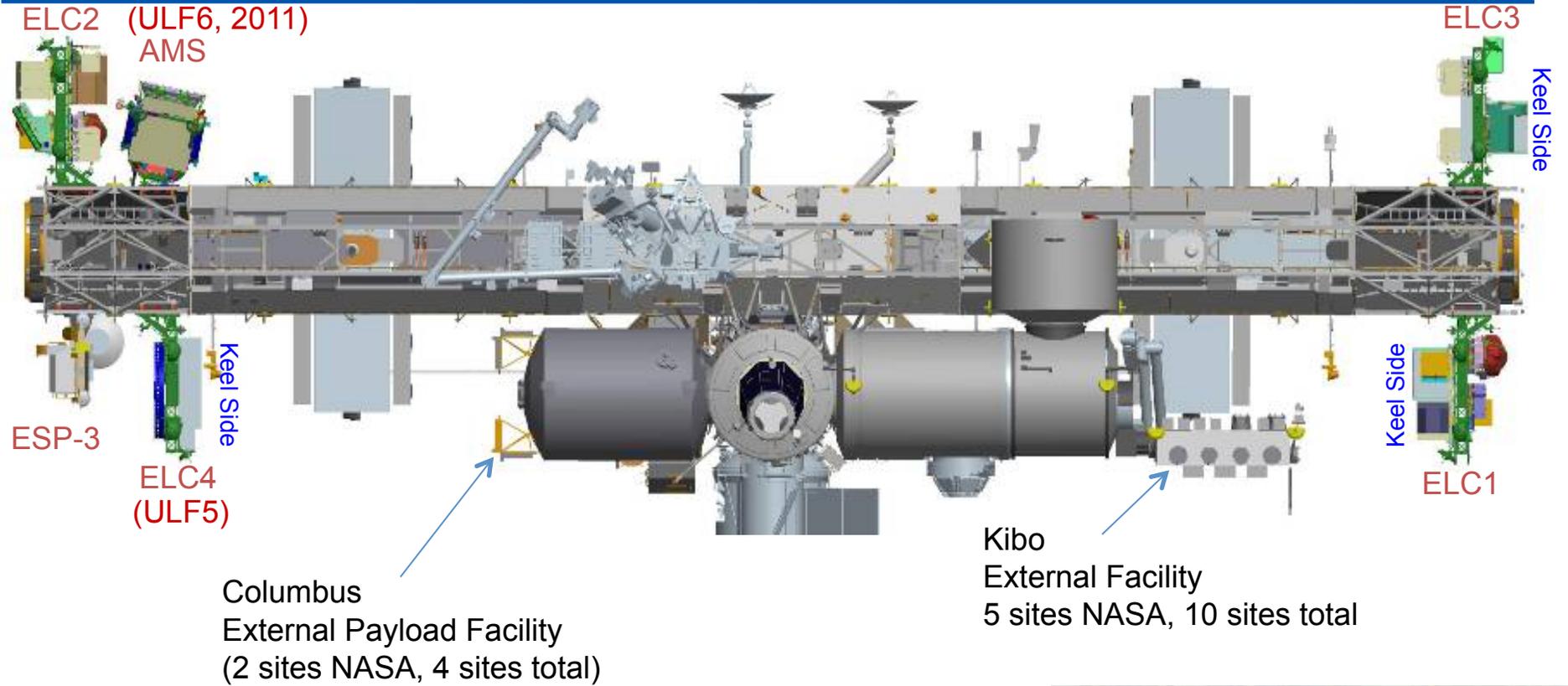
(Expedite the Processing of Experiments for Space Station)

System	Middeck Locker Locations	ISIS Drawer Locations	Rack-Level Accommodation
Structural	72 lbs. within cg constraints	64 lbs. within cg constraints	8 Mid deck Lockers 2 ISIS Drawers (4 Panel Unit)
Power	28 Vdc, 0 – 500 W	28 Vdc, 0 – 500 W	2000 Watts 28Vdc power
Air Cooling	≤ 200 Watts	<100 Watts	1200 Watts
Thermal Control System Water Cooling	500 Watts (2 positions per rack)	500 Watts (2 positions per rack)	2 positions per rack
Command and Data Handling	RS422 Analog Ethernet 5 Vdc Discrete	RS422 Analog Ethernet 5 Vdc Discrete	RS422 Analog Ethernet 5 Vdc Discrete
Video	NTSC/RS170A	NTSC/RS170A	NTSC/RS170A
Vacuum Exhaust System	1 payload interface per rack	1 payload interface per rack	1 payload interface per rack
Nitrogen	1 payload interface per rack	1 payload interface per rack	1 payload interface per rack



International Space Station (ISS) External Research Facilities

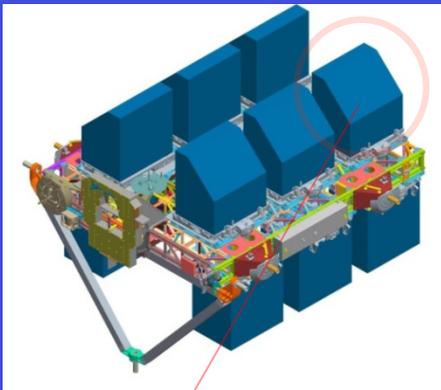
(ULF6, 2011)





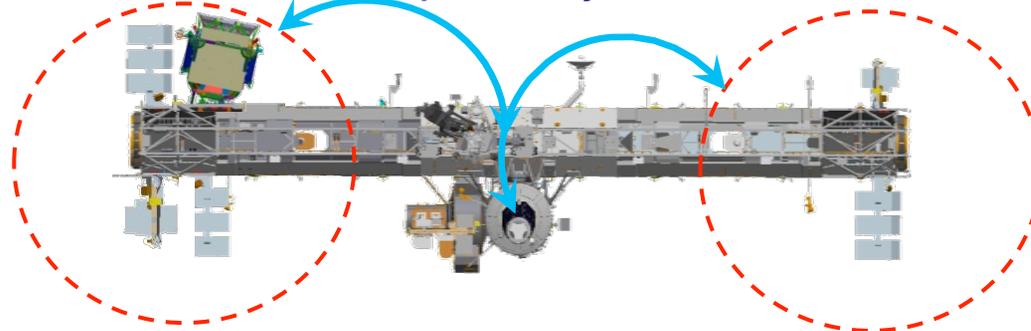
External Research Accommodations

Express Logistic Carrier

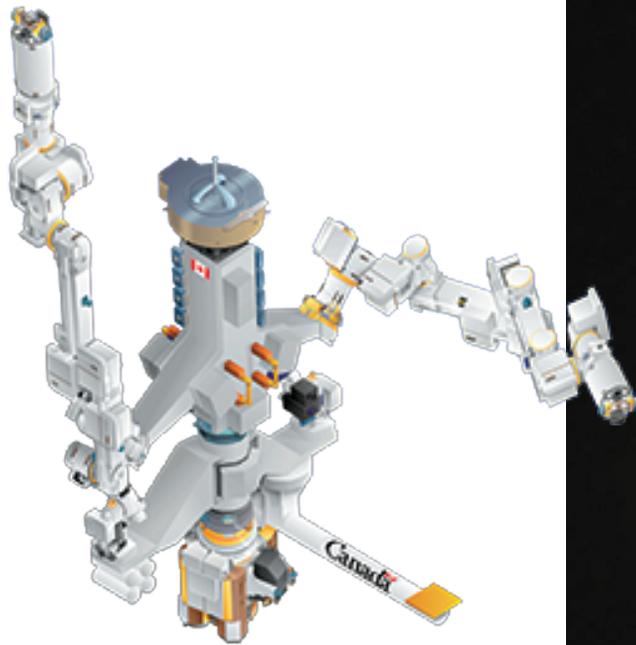
<p>ELC Single Adapter Resources (2)</p> <p>NASA payload sites per ELC)</p> 	Mass capacity	227 kg (500 lb)
	Volume	1 m ³
	Power	750 W, 113 – 126 VDC; 500 W at 28 VDC/adapter
	Thermal	Active heating, passive cooling
	Low-rate data	*1 Mbps (MIL-STD-1553)
	Medium-rate data	*6 Mbps (shared) - Return link (payload to ISS) only
	Sites available per ELC	2 sites
	Total ELC sites available	8 sites

Research Payload ExPA
(see next chart)

Proposed C&DH Enhancement to each Research Payload site
100 Mbps Two Way wireless LAN



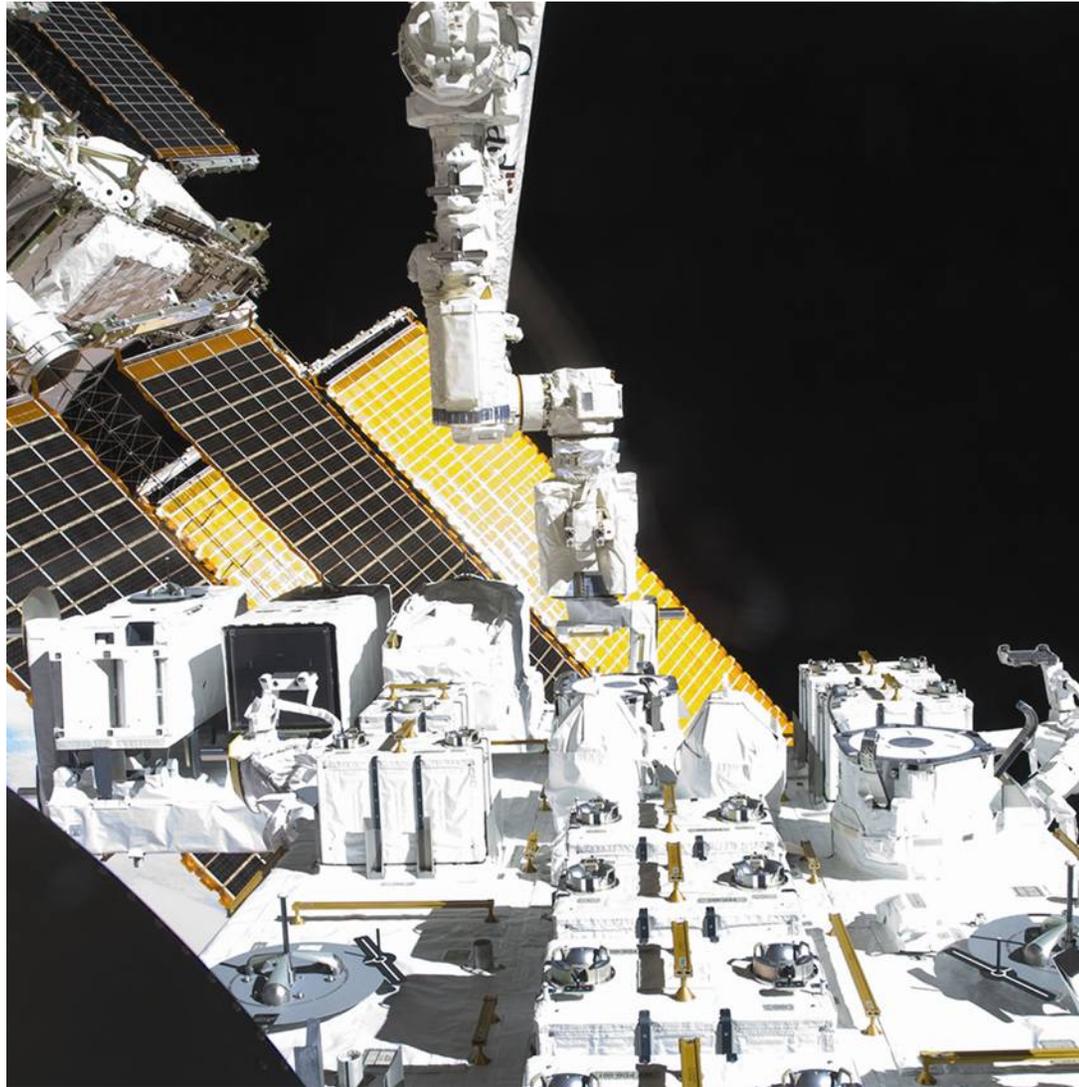
Dexterous End Effector



SSRMS attachment which the ground team or on-orbit crew can use robotically to install, remove and replace payloads and failed components

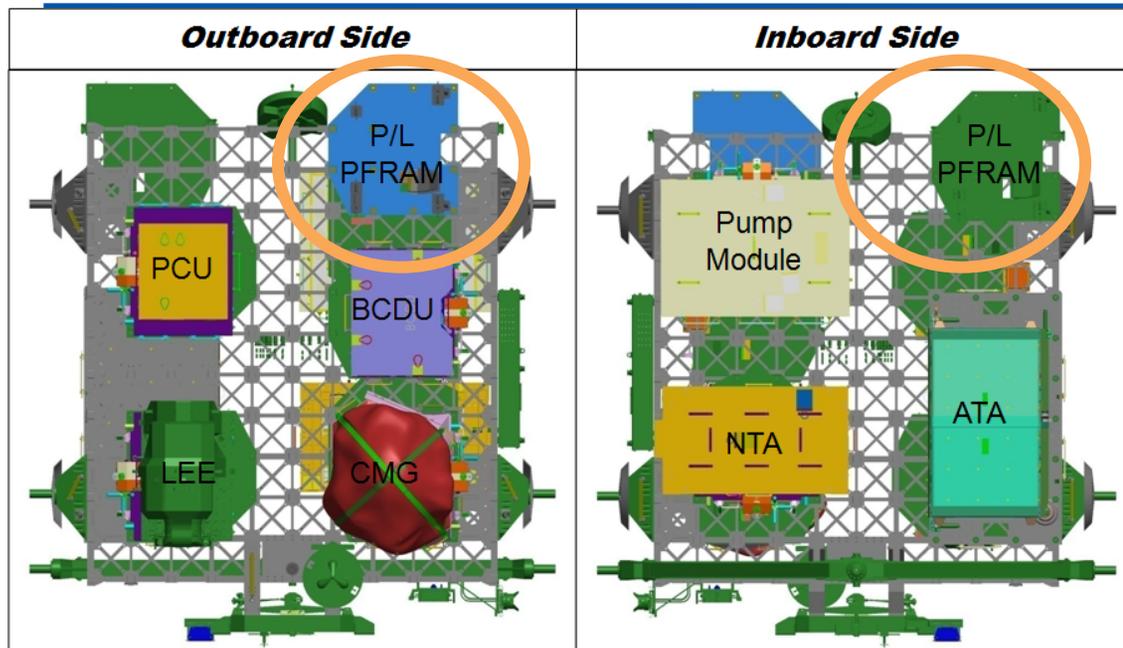


JEM RMS Payload Support



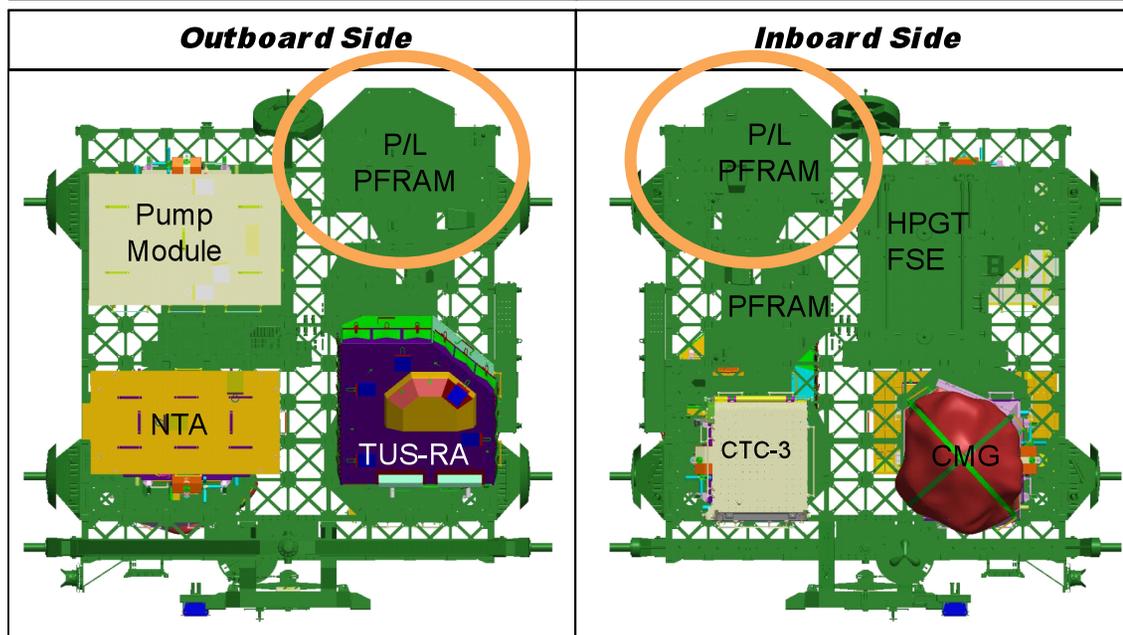


Express Logistics Carriers Overview



Payload Locations Circled

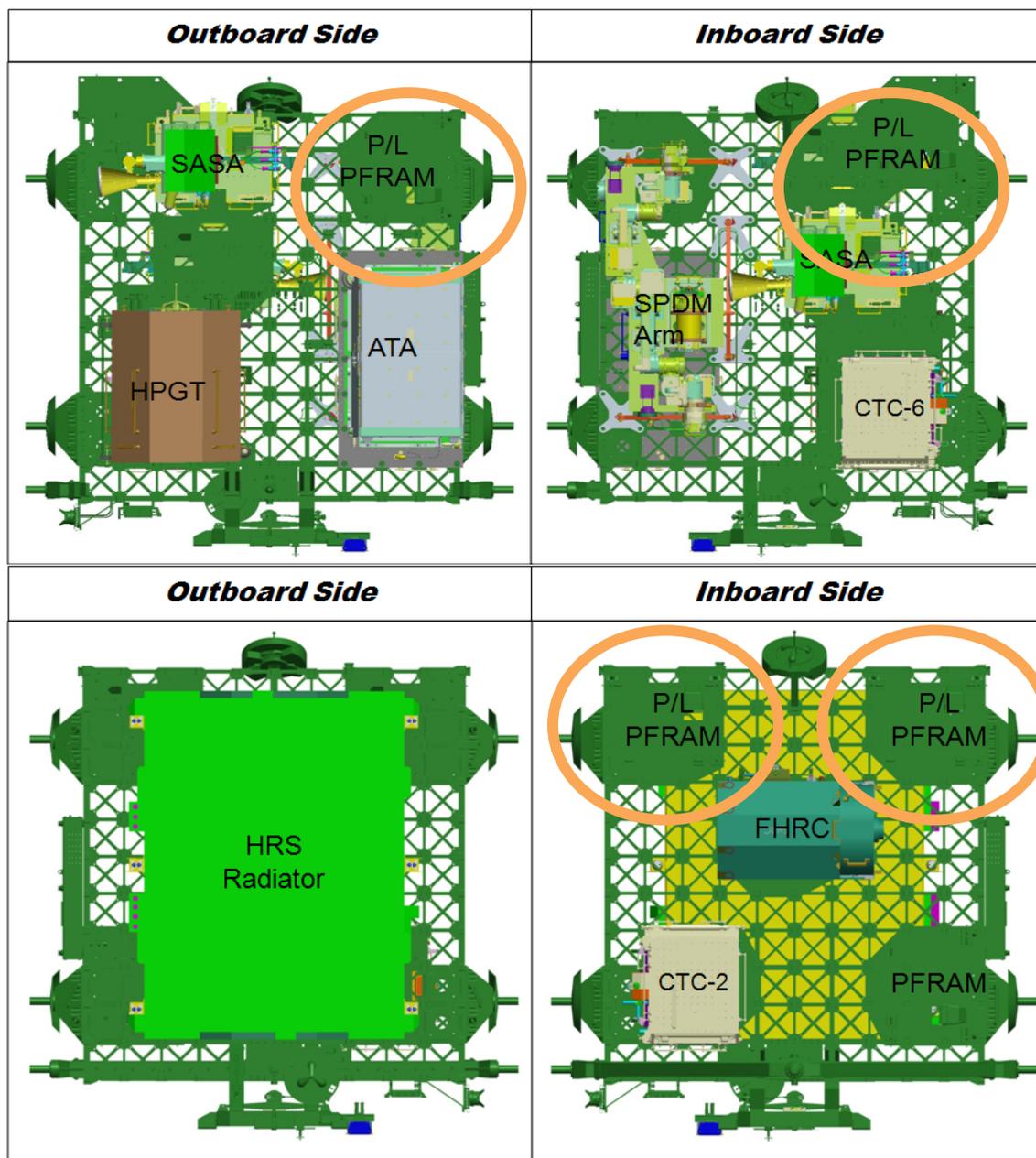
ELC-1
Port lower
2 Nadir payload sites



ELC-2
Starboard upper
2 Zenith payload sites



Express Logistics Carriers Overview



Payload Locations Circled

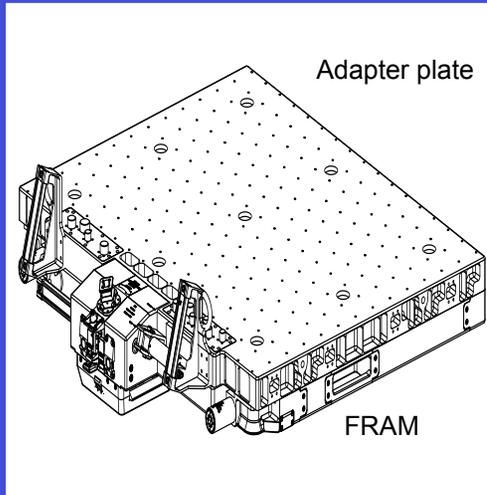
ELC-3
Port upper
2 Zenith payload sites

ELC-4
Starboard lower
2 Nadir payload sites



Express Pallet Adapter (ExPA) Assembly (GFE)

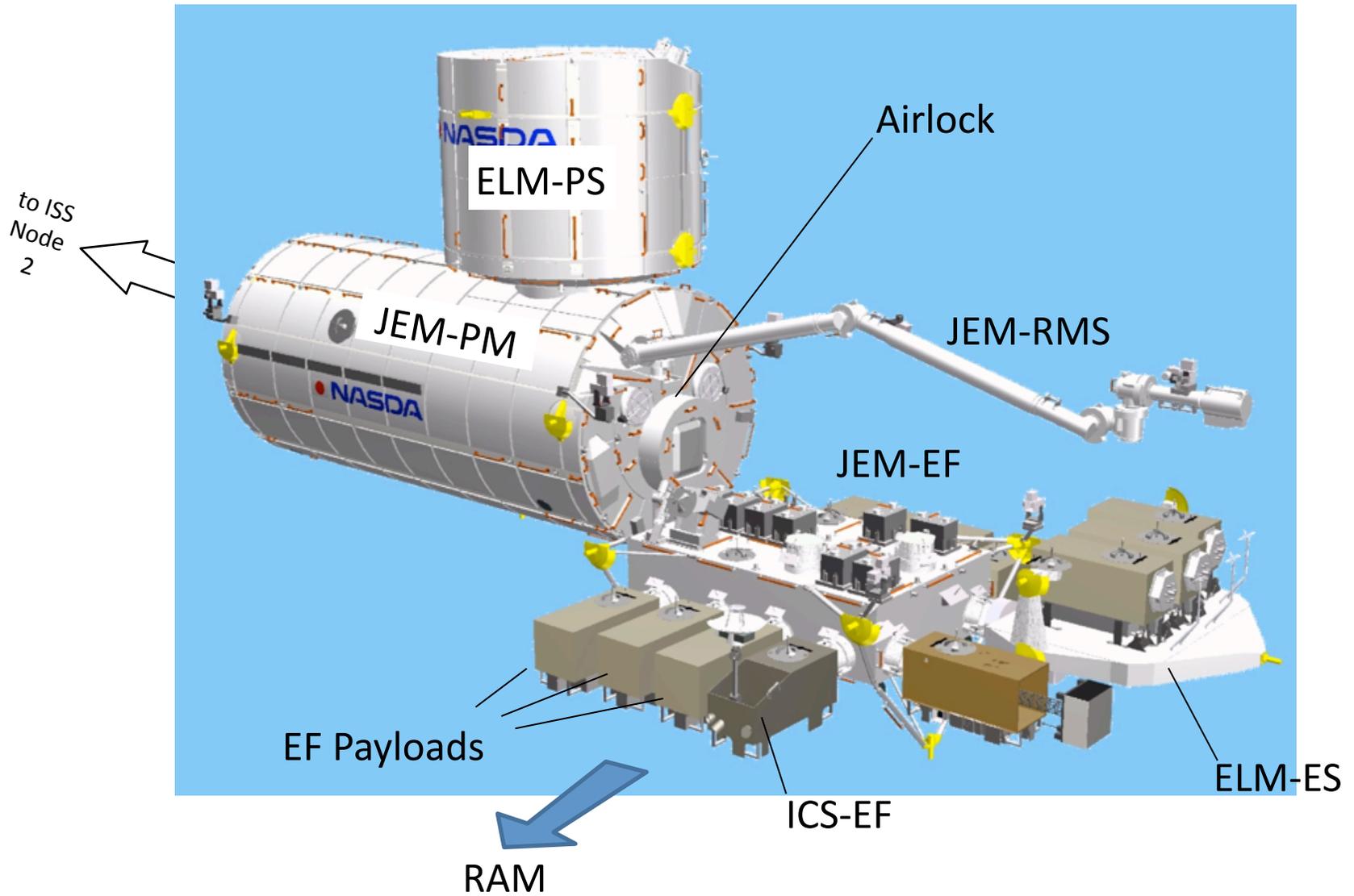
Express Pallet Adapter (ExPA) Assembly



ExPA overall Mass	255 lb
ExPA overall dimension	46.05" x 47" x 13.06" (H)
ExPA payload carrying capability	34" x 46" x 49" (H) and 500 lb"
Payload electrical interface	Power(120VDC & 28VDC): Four NATC connectors Data (1553, Ethernet): Six NATC connectors
Payload thermal interface	Active heating, passive cooling
Payload structural interface	2.756" X 2.756" Grid with 250-28 UNF Locking Inserts and 1.625" diameter Shear Boss Provisions
EVA compatibility	EVA handrail provisions
EVR compatibility	All EVR interfaces on ExPA

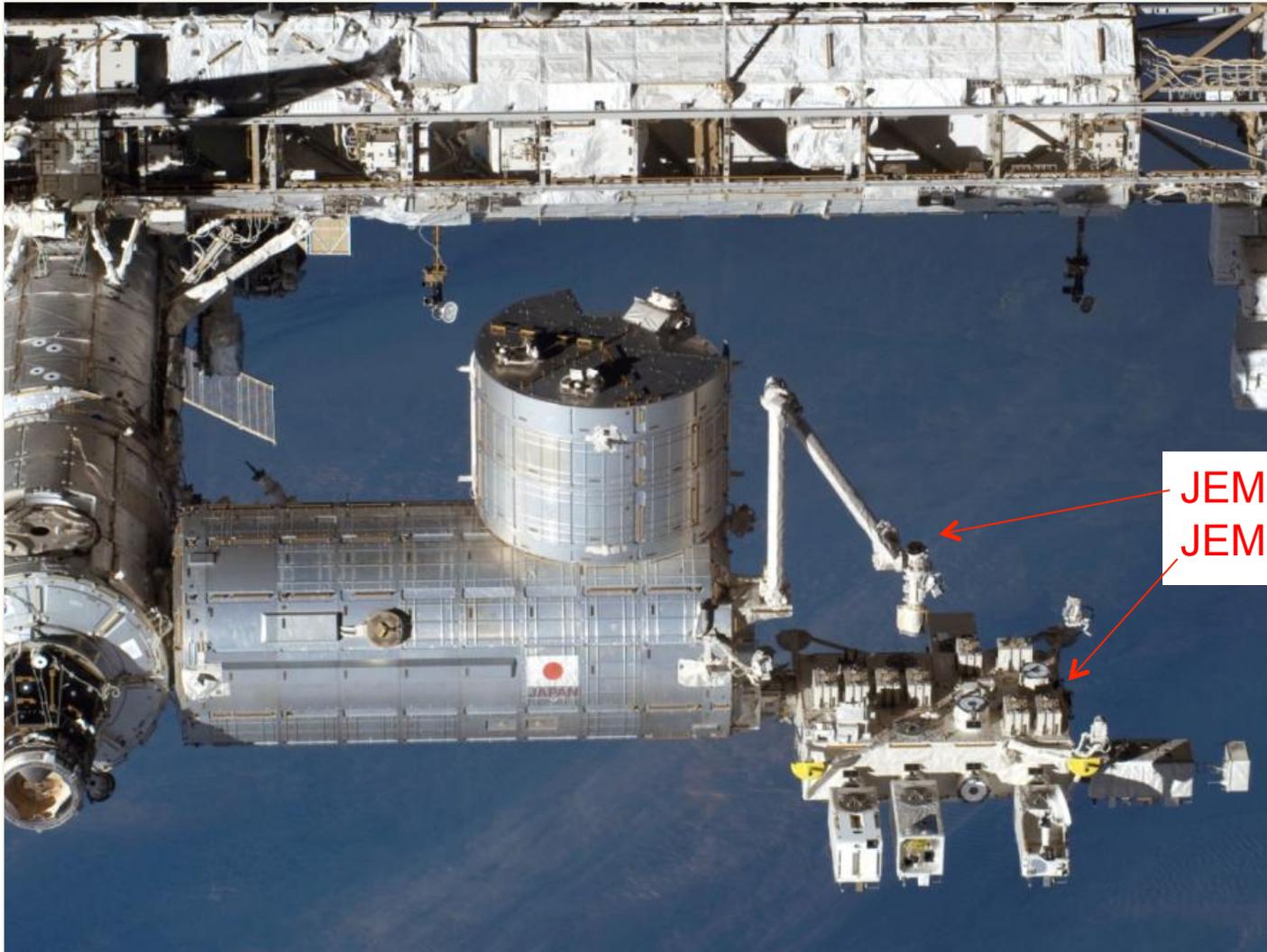


Japanese Experiment Module Exposed Facility (JEM EF) Overview





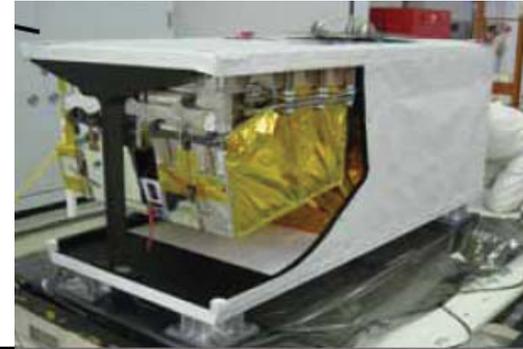
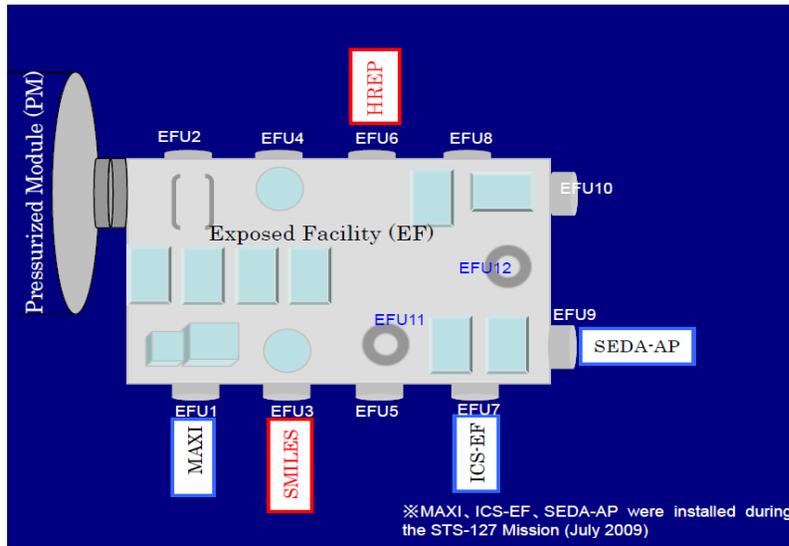
Japanese Experiment Module - *Kibo*



JEM RMS
JEM External Facility

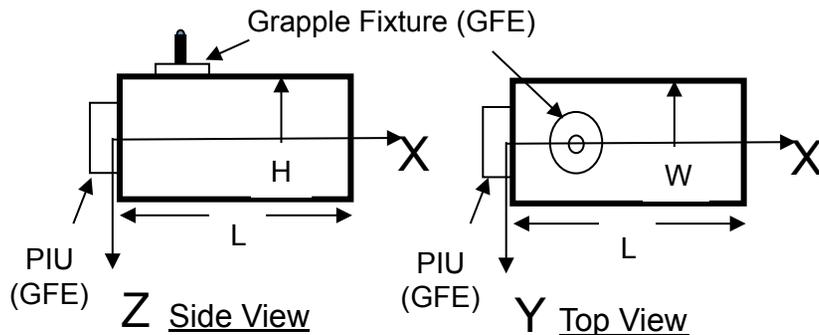


JEM EF External Research Accommodations



NASA/DOD
HREP payload

Mass capacity	550 kg (1,150 lb) at standard site 2,250 kg (5,550 lb) at large site
Volume	1.5 m ³
Power	3-6 kW, 113 – 126 VDC
Thermal	3-6 kW cooling
Low-rate data	1 Mbps (MIL-STD-1553, two way)
Medium-rate data	1EEE-802.3(10BASE-T, two way) *
High-rate data	43 Mbps (shared, one way downlink)
Sites available to NASA	5 sites

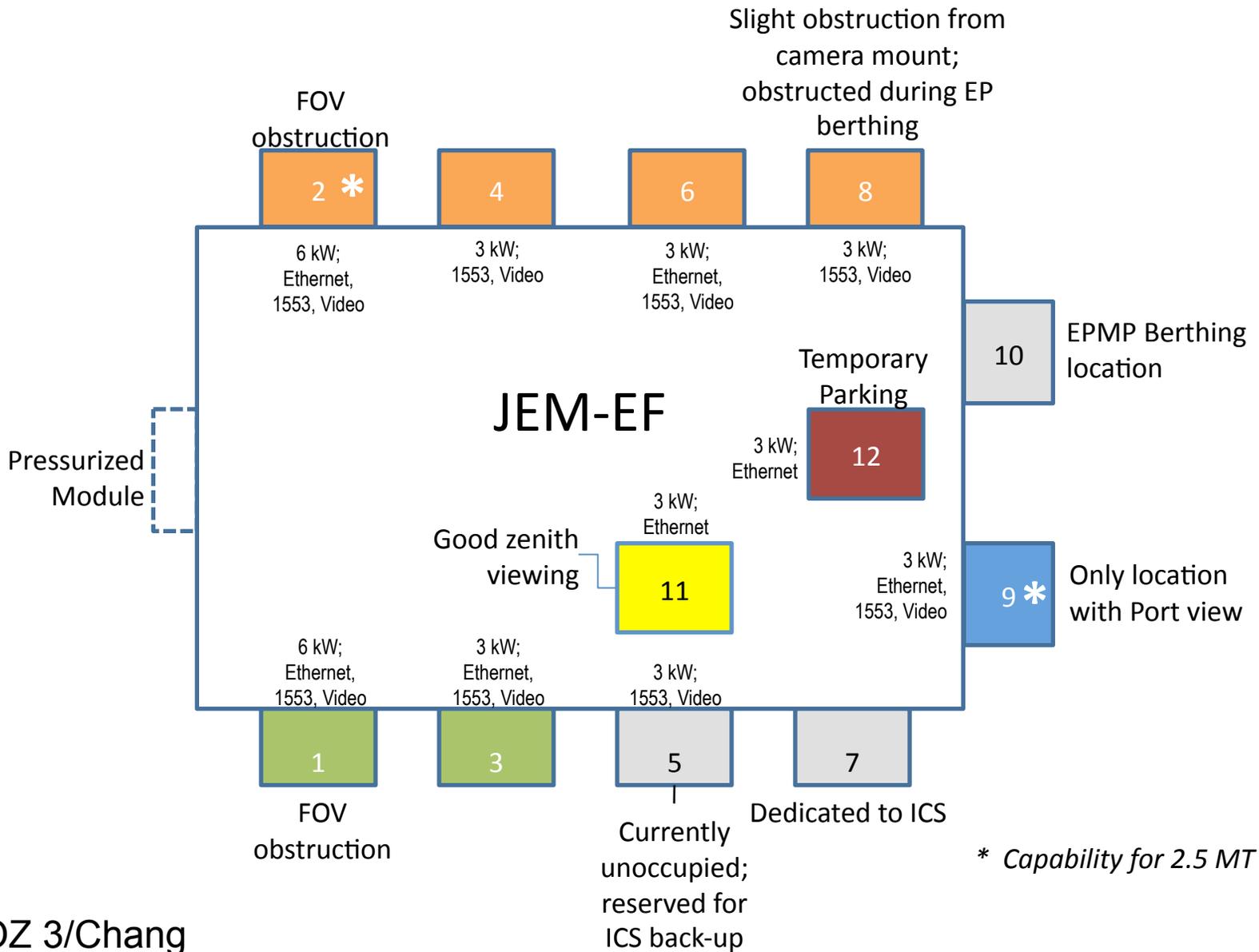


Axis	mm	ft	inch
W	800	2	7.50
H	1000	3	3.37
L	1850	6	0.83

- Ethernet bus is tested to 100BASE-T capacity.
- Upgrade to 100BASE-T is being worked by JAXA



JEM EF EFU Location Overview



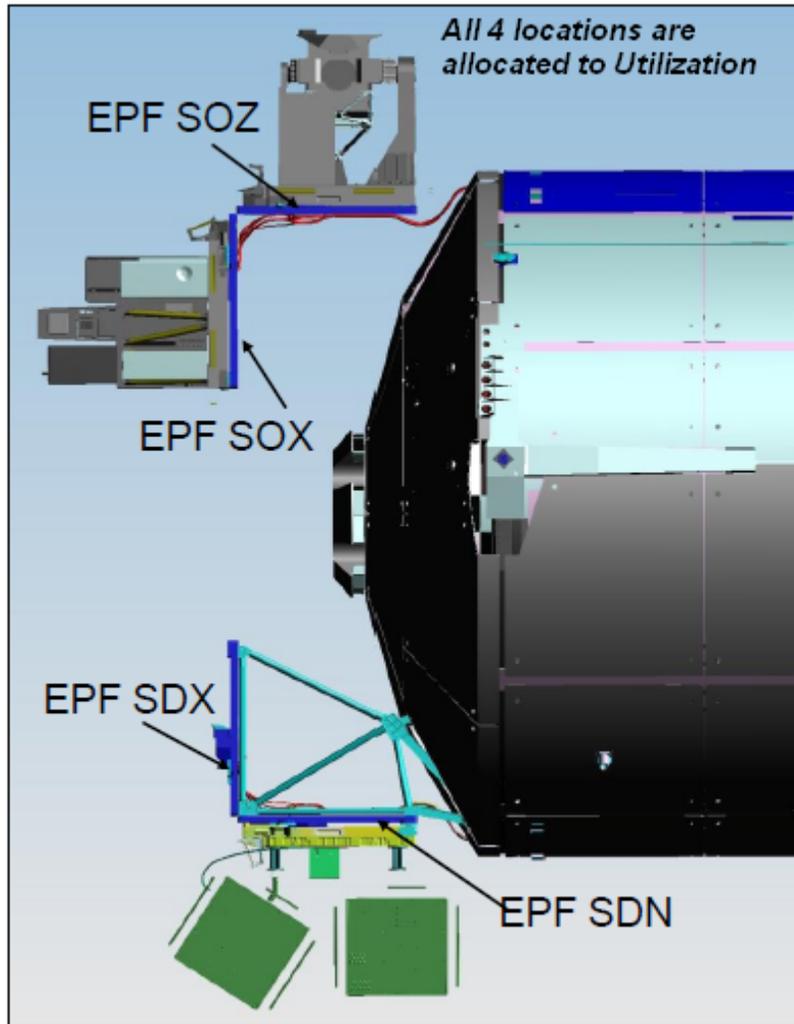


JEM-EF Detailed Accommodations by Site

Location	Viewing	Payload Size	Description / Notes	Power	Data
1	Ram, Nadir, Zenith	500 kg	Ram field of View (FOV) obstruction by JEM module	6 kW	Ethernet, 1553, Video
3	Ram, Nadir, Zenith	500 kg	Clear view	3 kW	Ethernet, 1553, Video
5	Ram, Nadir, Zenith	500 kg	ICS System back-up site (negotiable?)	3 kW	Ethernet, 1553, Video
7	Ram, Nadir, Zenith	500 kg	ICS-dedicated	-	-
9	Port, Zenith, Nadir	2.5 MT	Best volumetrically for large payloads (up to 2.5 MT), but not necessarily the best viewing	3 kW	Ethernet, 1553, Video
2	Wake, Nadir, Zenith	2.5 MT	Can hold large payloads, but has an FOV obstruction by JEM module	6 kW	Ethernet, 1553, Video
4	Wake, Nadir, Zenith	500 kg	Clear view	3 kW	1553, Video
6	Wake, Nadir, Zenith	500 kg	Clear view	3 kW	Ethernet, 1553, Video
8	Wake, Nadir, Zenith	500 kg	Obstruction during EP berthing, slight obstruction from camera mount	3 kW	1553, Video
10	Wake, Nadir, Zenith	500 kg	EPMP berthing site	-	-
11	Zenith only	500 kg	Good Zenith viewing	3 kW	Ethernet
12	Zenith only	500 kg	Temporary stowage location	3 kW	Ethernet



Columbus EF Overview



Location	Viewing	Payload Size	Power	Data
SOZ	Zenith	226 kg + CEPA	1.25 kW at 120 VDC 2.5 kW max	Ethernet, 1553
SOX	Ram			
SDX	Ram			
SDN	Nadir			

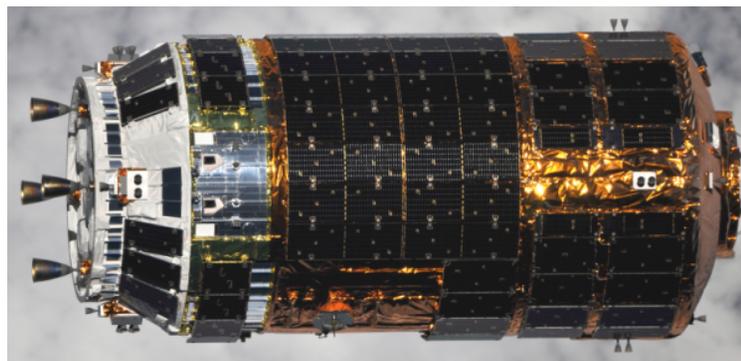
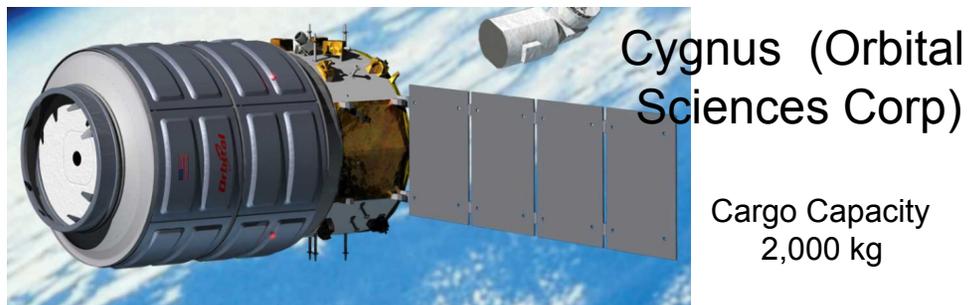


Columbus EF





ISS Cargo Vehicles





Payload Allowable Up-Mass & Volume Summary Table

Attach Payload Location	Allowable Payload Weight (including Flight Support Equipment)	Accommodation Weight (including adapter plate)	Total Weight	Payload Volume (W x H x L)
HTV Exposed Pallet (JEM EF Payload)	979 Lb (445 Kg)	121 Lb (55 Kg)	1100 Lb (500 Kg)	31.5" x 39.4" x 72.8" (800mm x 1000mm x 1850 mm)
HTV Exposed Pallet (ExPA, CEPA Payload)	See ExPA & CEPA payload specification for ELC & CEF	See ExPA & CEPA payload specification for ELC & CEF	*See ExPA & CEPA payload specification for ELC & CEF	*See ExPA & CEPA payload specification for ELC & CEF
ELC (ExPA)	490 Lb (222 Kg)	250 Lb (114 Kg)	740 Lb (336 Kg)	34" x 49" X 46" (863mm x 1244mm x 1168 mm)
Columbus (CEPA)	388 Lb (176Kg)	250 Lb (114 Kg)	638 Lb (290 Kg)	34" x 49" X 46" (863mm x 1244mm x 1168 mm)
JEM-EF	979 Lb (445 Kg)	121 Lb (55 Kg)	1100 Lb (500 Kg)	31.5" x 39.4" x 72.8" (800mm x 1000mm x 1850 mm)

* Location constraint applies in HTV Exposed Pallet



Upgrades In Work

Enhanced Processor and Integrated Communications (EPIC) Project	Phase A will upgrade the three Command and Control (C&C) MDMs and the two Guidance, Navigation, & Control (GN&C) MDMs.
	Phase B will upgrade the two Payload MDMs, and add Ethernet support for the C&C and Payload MDMs.
Air to Ground High Rate Communications System (HRCS) Project	Increase data rates internally and on the RF link 300 Mbps downlink, 7/25 Mbps uplink
	Combine audio and video on orbit
	Provide two way, high quality audio
	Open the door to internet protocol communications
	Open the forward link to multiple users
	Allow for the capability of transmitting & recording HDTV
On Orbit External Wireless High Rate	100 Mbps 2-way Ethernet capability
	1 Mbps 1553 capability
	Up to 4 antennas attached to EVA handrails on US Lab



External Manifesting and ISS Locations ELCs, Columbus-EF (draft through 2015 shown)

FLIGHTS	Calendar Year		2012				2013				2014				2015						
	MONTH	MAR	APR	JUL	OCT	JAN	JUN	JUL	OCT	JAN	JUL	AUG	DEC	APR	JUL	DEC					
	Inc-Year	2012-1	2012-2			2013-1		2013-2		2014-1		2014-2		2015-1	2015-2		2016-1				
	Major Element(s)	Soyuz (Crew)	SpaceX CRS2	SpaceX CRS3	SpaceX CRS4	SpaceX CRS5	HTV	SpaceX CRS6	SpaceX CRS7	SpaceX CRS8	HTV	SpaceX CRS9	SpaceX CRS10	SpaceX CRS11	HTV	SpaceX CRS12					
	Flight	S0S	Sx2	Sx3	Sx4	Sx5	HTV4	Sx6	Sx7	Sx8	HTV5	Sx9	Sx10	Sx11	HTV6	Sx12					
UNPRESSURIZED FACILITY LAUNCH / DISPOSAL PLANS AND ON-ORBIT TOPOLOGY																					
	Carrier	Location	Number	Viewing																	
ELC 1	P3 Lower	3	Outboard / Ram / Nadir												EuTEF-2 ↑ (NASA launch; ESA ops)	EuTEF-2	EuTEF-2	EuTEF-2	EuTEF-2 ↓		
		8	Inboard / Wake / Nadir				OPALS ↑ (NASA)	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS	OPALS ↓	
ELC 4	S3 Lower	2	Inboard / Wake / Nadir												DPP ↓ STP-H4 (from E 3-5)	STP-H4	STP-H4 ↓	Tech, Earth, Exp & Space Sci Platform (ESA) ↑	Tech, Earth, Exp & Space Sci Platform	Tech, Earth, Exp & Space Sci Platform	
		3	Inboard / Ram / Nadir	RRM ^{3,8}	RRM ^{3,8}	RRM ^{3,8}	RRM ^{3,8}	RRM ^{3,8}	RRM ^{3,8} ↓ (NBR-6)		SAGE NVP (NASA) ↑ SAGE III/Hexapod (w/NVP) ↑	SAGE III/Hexapod (w/NVP)	SAGE III/Hexapod (w/NVP)	SAGE III/Hexapod (w/NVP)	SAGE III/Hexapod (w/NVP)	SAGE III/Hexapod (w/NVP)					
ELC 2	S3 Upper	3	Inboard / Ram / Zenith	MISSE 8 ^{2,3}	MISSE 8 ^{2,3}	MISSE 8 ^{2,3}	MISSE 8 ^{2,3}	MISSE 8 ^{2,3} ↓ PRELSET I ¹ (NASA)	PRELSE II ²	PRELSE II ²	PRELSE II ²	PRELSE II ²	PRELSE Resupply (NASA) ↑↓	PRELSE II ²	PRELSE II ²	PRELSE Resupply (NASA) ↑↓	PRELSE II ²	PRELSE II ²	PRELSE II ²	PRELSE II ²	
		7	Outboard / Ram / Zenith					TBD-E2-7 ↑ (NASA)	TBD-E2-7	TBD-E2-7	TBD-E2-7	TBD-E2-7	TBD-E2-7	TBD-E2-7	TBD-E2-7	TBD-E2-7 ↓ TBD E2-7.2 (NASA) ↑	TBD E2-7.2	TBD E2-7.2	TBD E2-7.2	TBD E2-7.2	
ELC 3	P3 Upper	3	Inboard / Ram / Zenith	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed ³	SCAN Testbed-2 ⁴ ↓ SCAN Testbed-2 ⁴ ↑	SCAN Testbed-2 ⁴	SCAN Testbed-2 ⁴				
		5	Outboard / Wake / Zenith	STP-H3 ³	STP-H3 ³	STP-H3 ³ ↓		STP-H4 ↑ (NASA)	STP-H4	STP-H4	STP-H4	STP-H4 (to E4-2)		TBD E 3-5 ↑ (NASA)	TBD E 3-5	TBD E 3-5	TBD E 3-5	TBD E 3-5	TBD E 3-5	TBD E 3-5	TBD E 3-5

All TBDs except TBD-ESA-1 and TBD-ESA-2 (highlighted in gold) represent NASA opportunities that have not yet been named (minimum of 1 Columbus opportunity and 3 ELC opportunities)



External Manifesting and ISS Locations JEM-EF

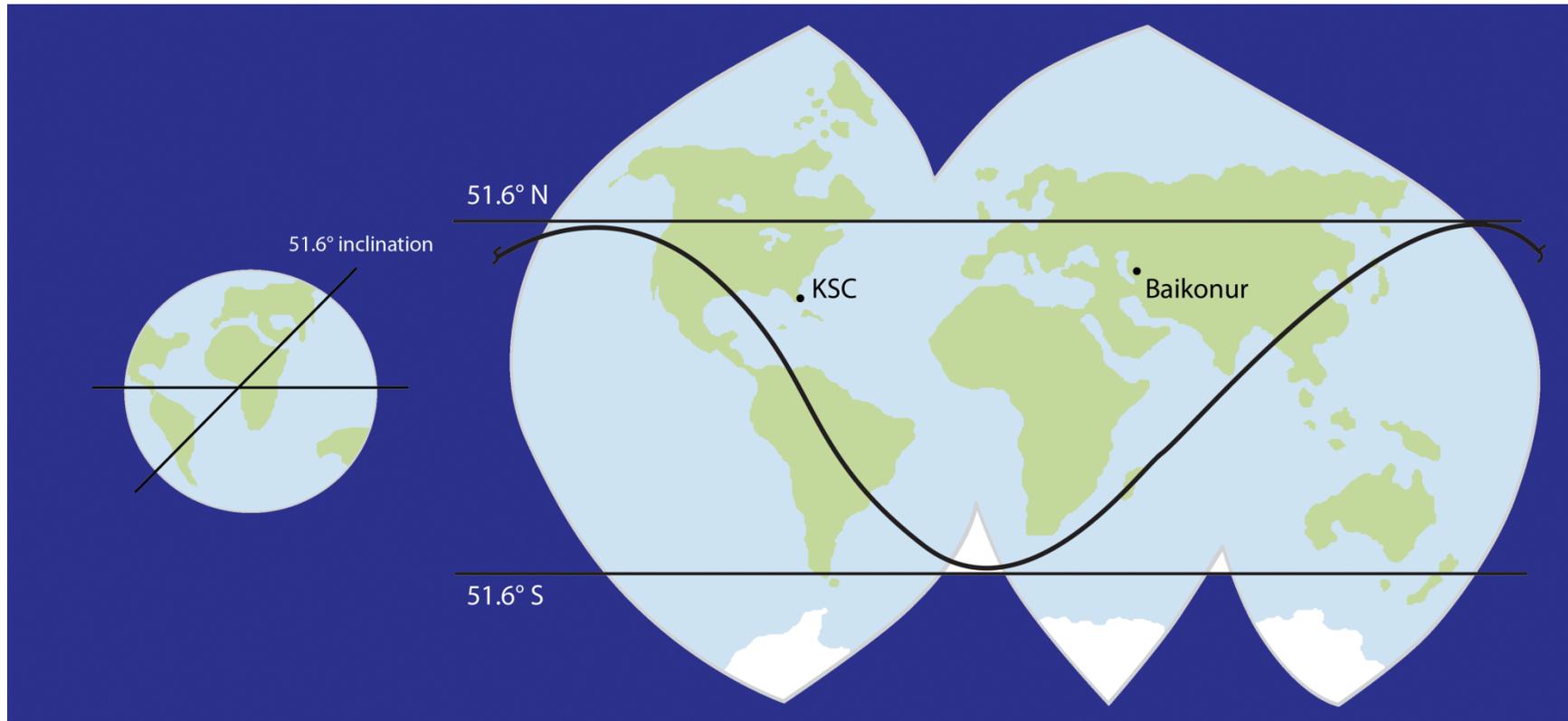
(draft through 2015 shown)

FLIGHTS	Calendar Year		2012				2013				2014				2015		
	MONTH	MAR	APR	JUL	OCT	JAN	JUN	JUL	OCT	JAN	JUL	AUG	DEC	APR	JUL	DEC	
	Inc-Year	2012-1	2012-2		2013-1		2013-2		2014-1		2014-2		2015-1	2015-2		2016-1	
	Major Element(s)	Soyuz (Crew)	SpaceX CRS2	SpaceX CRS3	SpaceX CRS4	SpaceX CRS5	HTV	SpaceX CRS6	SpaceX CRS7	SpaceX CRS8	HTV	SpaceX CRS9	SpaceX CRS10	SpaceX CRS11	HTV	SpaceX CRS12	
	Flight	30S	Sx2	Sx3	Sx4	Sx5	HTV4	Sx6	Sx7	Sx8	HTV5	Sx9	Sx10	Sx11	HTV6	Sx12	
UNPRESSURIZED FACILITY LAUNCH / DISPOSAL PLANS AND ON-ORBIT TOPOLOGY																	
JEM-EF	Carrier Location Number	Viewing															
	1	Ram	MAXI ³	MAXI ³	MAXI ³	MAXI ³	MAXI ³										
	3	Ram	SMILES ³	SMILES ³	SMILES ³	SMILES ³	SMILES ³										
	5	Ram															
	7	Ram															
	<i>Systems (ICS-EF)</i>																
	9 ⁵	Ram	SEDA-AP ³	SEDA-AP ³	SEDA-AP ³	SEDA-AP ³	SEDA-AP ³										
	2	Wake											CALET ³ ↑ (TBR-2) (GOJ)	CALET	CALET	CALET	CALET
	4	Wake											TBD-J-2 ↑ (TBR-2) (NASA)	TBD-J-2	TBD-J-2	TBD-J-2	TBD-J-2
	6	Wake	HREP ³	HREP ³	HREP ³	HREP ³	HREP ³										
	8	Wake	MCE ³	MCE ³	MCE ³	MCE ³	MCE ³										
	10	Wake															
	<i>Systems (HTV/EP)</i>																
11	Zenith											SEDA-AP ³ (from J-9)	SEDA-AP ³	SEDA-AP ³	SEDA-AP ³	SEDA-AP ³	
<i>Temp / Staging⁶</i>																	
S3 Upper Inboard	Zenith	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³	AMS ³		

- All TBDs (highlighted in gold) represent NASA opportunities that have not yet been named (minimum of 3 sites with flight opportunities)
- Location/site numbers are color-coded to match the illustration on page 5



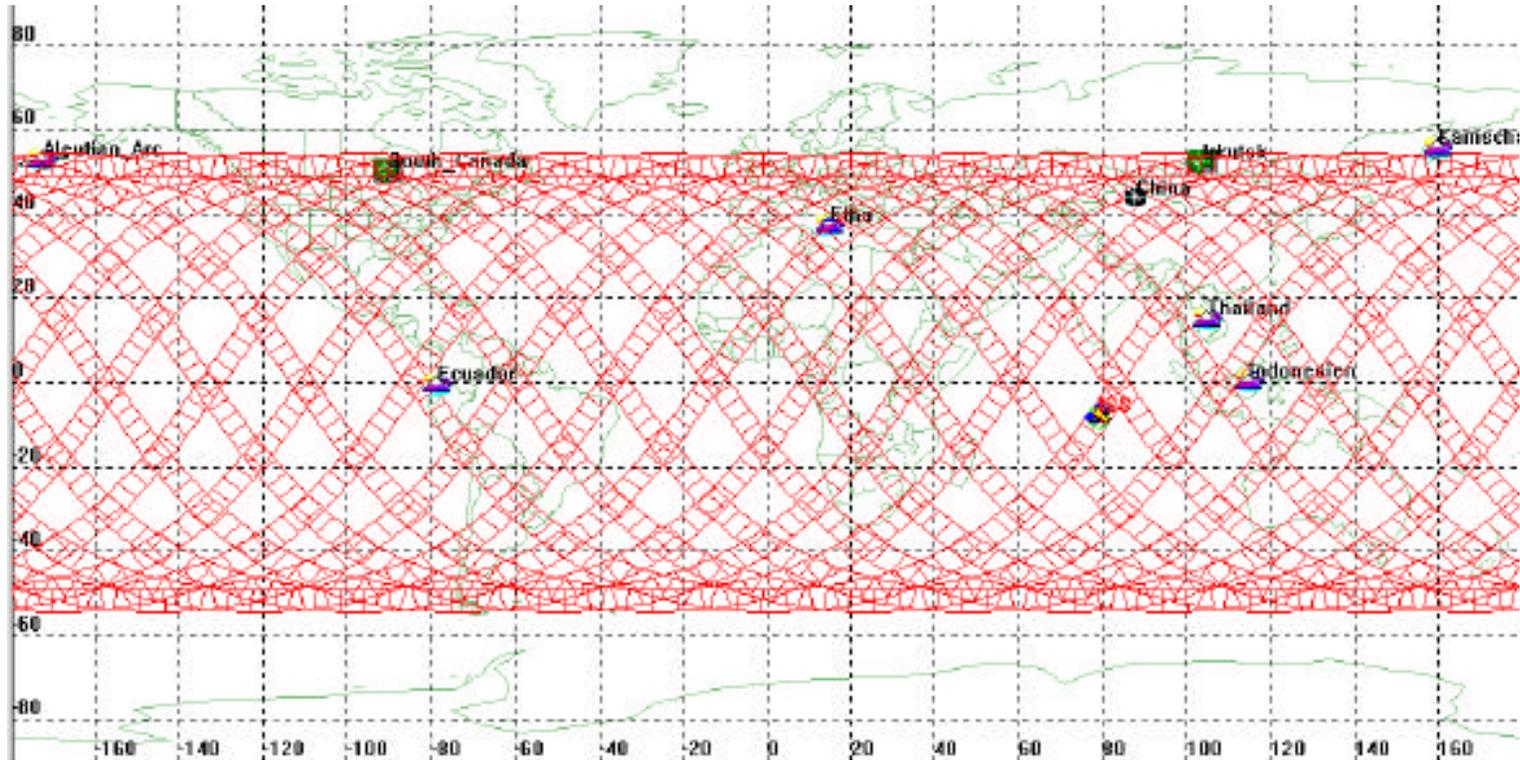
ISS as a Platform for Earth Science



All geographic locations between 51.6 North and South latitude
can be observed NADIR pointing
Provides coverage of 85% of the Earth's surface and 95% of the world's
populated landmass every 1-3 days



ISS as a Platform for Earth Science



ISS coverage in 24 hrs for a 70°-swath optical payload. (Courtesy of ESA)

Processing lighting (changes with subsequent passes)
Well-suited for test bed concepts with hardware change out
and upgrades



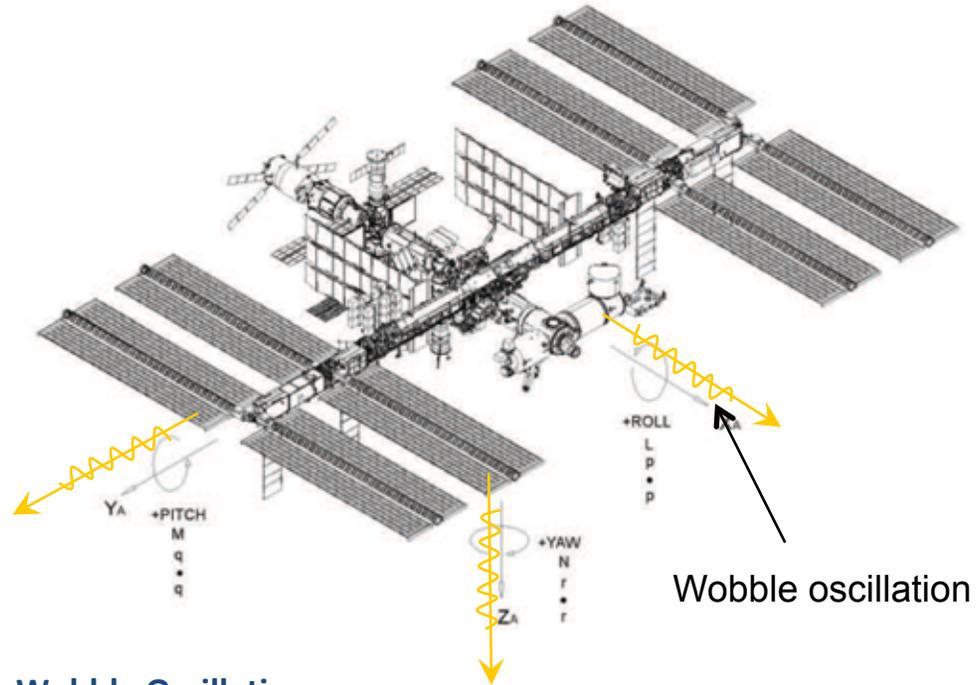
ISS Attitude Torque Equilibrium Attitude (TEA) & Wobble Oscillation Description

For Stage configurations (i.e.; no Orbiter or Orbiter sized vehicle docked on the ISS) in the foreseeable future, the predicted TEA ranges are:

Roll: -1.0 ~ +3.0 deg

Pitch: -7.0 ~ +2.0 deg

Yaw: -15 ~ +15 deg.



Momentum Manager Controller Peak to Peak Attitude Wobble Oscillation

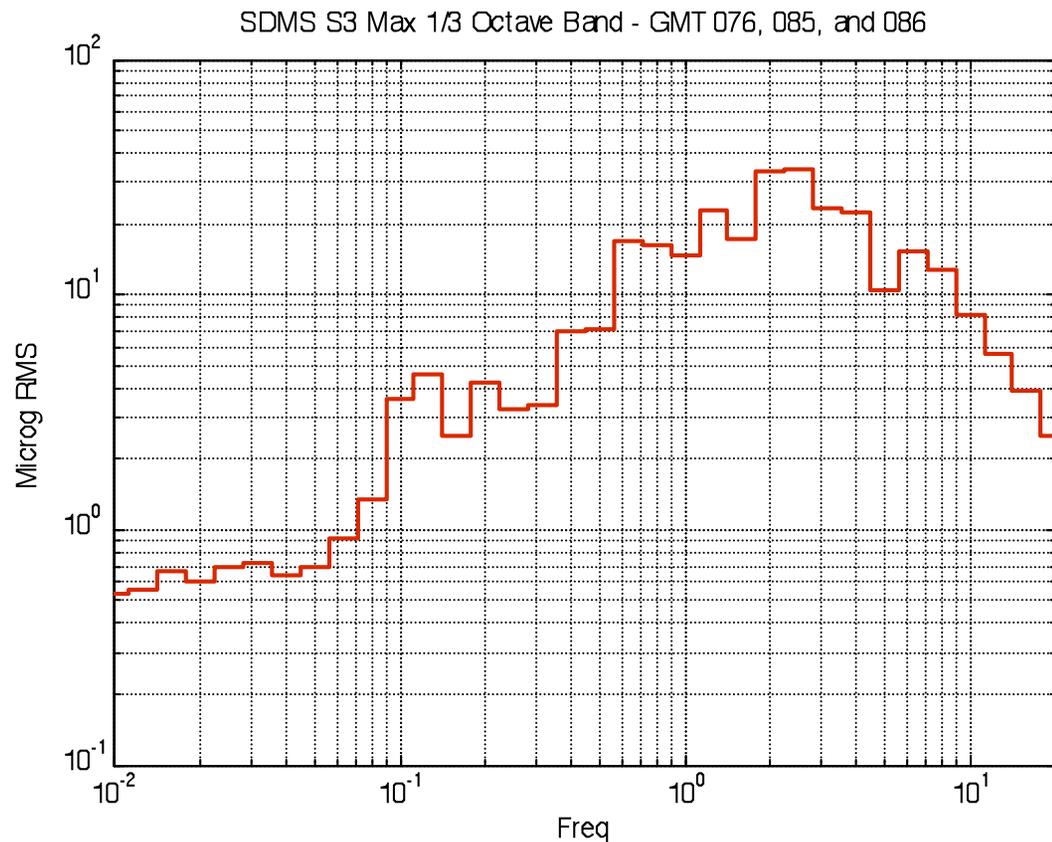
Performance Descriptions	Peak to Peak Attitude Oscillations Per Orbit			Peak Attitude Variation from Steady-State Orbit-Average Attitude		
	Roll (X) (deg)	Pitch (Y) (deg)	Yaw (Z) (deg)	Roll (X) (deg)	Pitch (Y) (deg)	Yaw (Z) (deg)
Non-Micro-Gravity (Assembly Stages) Non-Propulsive (Momentum Manager) Attitude Control Performance Requirement	10.0	10.0	10.0	+/- 5	+/- 5	+/- 5
Micro-Gravity (Assembly Complete) Non-Propulsive (Momentum Manager) Attitude Control Performance Requirement	7.0	7.0	7.0	+/- 3.5	+/- 3.5	+/- 3.5
Typical Steady-State Performance of Minimum CMG momentum oscillation Momentum Manager Controller	1.6	1.6	2.0	+/- 0.8	+/- 0.8	+/- 1
Typical Steady-State Performance of Minimum Attitude oscillation Momentum Manager Controller	1.6	0.4	0.2	+/- 0.8	+/- 0.2	+/- 0.1
Typical Steady-State Performance of Minimum CMG momentum & Attitude oscillation Blended Momentum Manager Controller	1.6	0.7	1.2	+/- 0.8	+/- 0.35	+/- 0.6



ISS Quiescent Mode Truss Vibratory Environment For External Payload Pointing Instrument

Data measured on ISS S3 truss

- ISS quiescent mode = No thruster firings, dockings, EVA, or robotics operations
- Typical response, not worst case
- Maximum per octave band
 - SDMS S3B1N on-orbit accelerometer data.
 - Snapshot of 3 10-minute data takes
 - All data taken on March 16, 26, and 27, Stbd SARJ Rotating, exercise, 3 crew.



ULF-4 analysis concluded peak ELC rotations on the order of 0.03 degrees (quiescent mode)



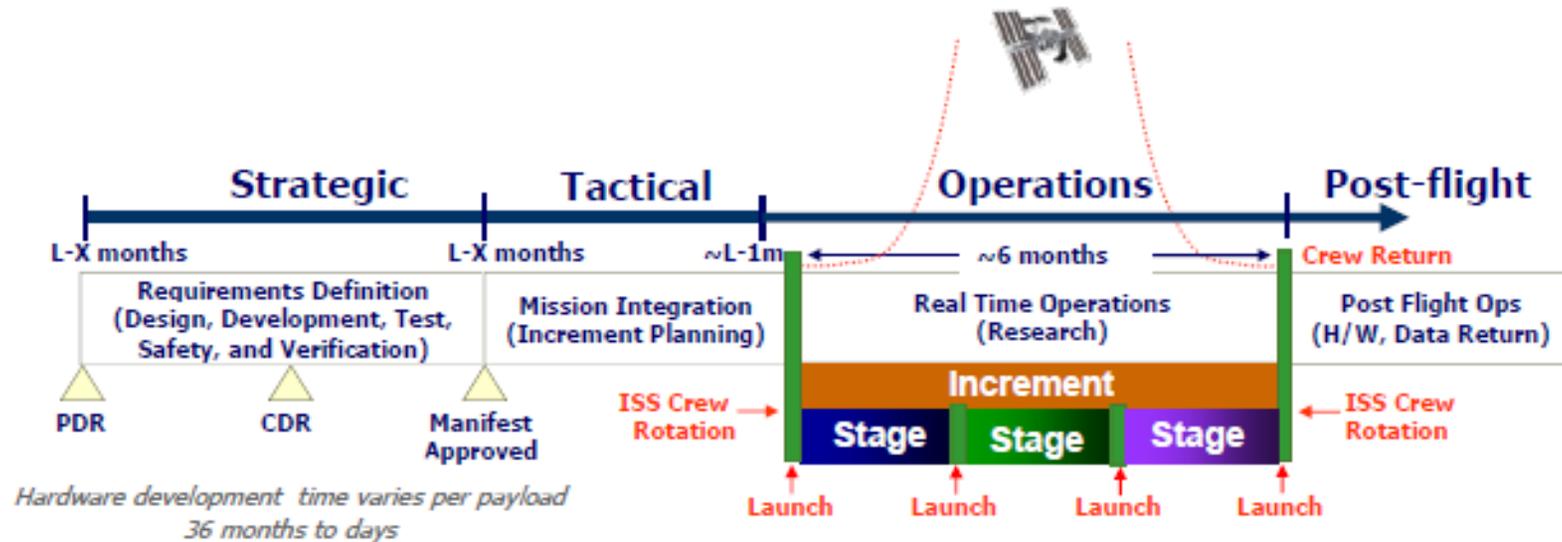
ISS Contamination Environment Description For Truss Attached Payload

- The International Space Station provides an exceptionally clean environment to external payloads and science assets
- External contamination control requirements limit contaminant deposition to 130Å/year on external payloads and ISS sensitive surfaces
 - Specified levels are lower than any previous space station (Mir, Skylab, Salyut) by several orders of magnitude
- Measurements of contaminant deposition on ISS returned hardware have demonstrated that requirements are met at ISS payload sites

Experiment	Side	Requirement (130Å/year)	Measured
MISSE 2	ram	520 Å (4 years)	50 Å
	wake	520 Å (4 years)	500 Å
Node 1 nadir window cover	nadir	390 Å (3 years)	50 Å



ISS Payload Integration Process



ISS provides:

- Launch to ISS
- Installation on ISS at identified site
- On-orbit utilities and operations support, including crew or robotics time (if needed)
- Data handling and delivery
- End-of-life removal and disposal



References

- **ISS Program Scientist Toolbox:** <http://iss-science.jsc.nasa.gov/index.cfm>
- **ISS National Laboratory Office:**
http://www.nasa.gov/mission_pages/station/research/nlab/index.html
- **Advanced Avionics Development Office:**
<http://iss-www.jsc.nasa.gov/nwo/avionics/aado/home/web/>
- **Attached Payload Interface Requirements Document, SSP 57003**
- **FRAM (ELC) Attached Payload Launch Vehicle IRD, SSP 57012**
- **ATV-2 Cargo Summary (24 Sep 2009)**
- **HII Transfer Vehicle Cargo IRD, HTV-CG-001 Rev D**
- **Requirements for International Partner Cargo Transported On Russian Progress and Soyuz Vehicles, П32928-103**
- **Cygnus Fact Sheet (Orbital, 2009)**
- **JEM EF Attached Payload Accommodation Handbook, NASDA-ESPC-2857B_Cargo IRD**
- **Columbus EF Payload Accommodations, COL-RIBRE-SPE-0165-1C_Columbus External Payloads IRD**



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Customer Service Helpline :The International Space Station Payloads Office has both a phone and an email customer service helpline that Payload Developers and others interested in doing research can contact to get assistance. The phone is staffed during regular business hours, or messages may be issued after hours, and a representative will return the call on the next business day. Phone: 281-244-6187, email: jsc-iss-payloads-helpline@mail.nasa.gov.”

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Acronyms

ACES	Atomic Clock Ensemble in Space
AMS	Alpha Magnetic Spectrometer
ASI	Italian Space Agency
ASIM	Atmospheric Space Interactions Monitor
ATA	Ammonia Tank Assembly
BCDU	Battery Charge Discharge Unit
CALET	Calorimetric Electron Telescope
C&DH	Command and Data Handling
CEF	Columbus Exposed Facility
CEPA	Columbus External Payload Adapter
CMG	Control Moment(um) Gyro(scope)
COL-EPF	Columbus Exposed Payload Facility
CSA	Canadian Space Agency
CTC	Cargo Transport Container
DPP	Dextre Pointing Package
ELC	External Logistics Carrier
ELM-ES	Experiment Logistics Module-Exposed Section
ELM-PS	Experiment Logistics Module – Pressurized Section
EF	Exposed Facility
EFU	Exposed Facility Unit
EPF	Exposed Payload Facility
EPMP	Exposed Pallet – Multi-Purpose
ESA	European Space Agency
EuTEF	European Technology Exposure Facility
EVA	Extravehicular Activity
EVR	Extravehicular Robotics
ExPA	EXPRESS Pallet Adapter



Acronyms (*Continued*)

FHRC	Flex Hose Rotary Coupler
FOV	Field of View
FSE	Flight Support Equipment
HPGT	High Pressure Gas Tank
HREP (RAIDS)	Hyperspectral Imager for the Coastal Ocean (HICO)/Remote Atmospheric and Ionospheric Detection System Experiment Payload
HRS	Heat Rejection Subsystem
HTV	H-II Transfer Vehicle (Japanese resupply vehicle)
ICS-EF	Inter-Satellite Communication System – Exposed Facility
ISS	International Space Station
JAXA	Japan Aerospace Exploration Agency
JEM	Japanese Experiment Module
JEM-EF	Japanese Experimental Module-Exposed Facility
JEM-PM	Japanese Experimental Module-Pressurized Module
Kg	kilogram
LAN	Local Area Network
LEE	Latching End EffectorMAXI Monitor All-sky X-ray Image
MCE	Multi-mission Consolidated Equipment
MIM	Multi-Increment Manifest
MiPROM	Multi-Increment Payload Resupply and Outfitting Manifest
MISSE	Materials International Space Station Experiment
NASA	National Aeronautics and Space Administration
NTA	Nitrogen Tank Assembly
ODAR	Obsolescence Driven Avionics Re-Design
OPALS	Optical Planetary Access Link for Space Station
PCU	Plasma Contactor Unit
PFRAM	Passive Flight Releasable Attach Mechanism



Acronyms (*Continued*)

PIU	Power Interface Unit
P/L	Payload
PRELSE	Platform for Retrievable Experiments in a Leo Space Environment
R2D2	Robotic Refueling Dexterous Demonstration using Dextre
RMS	Remote Manipulator System
SAGE III /Hexapod	Stratospheric Aerosol and Gas Experiment III w/ Hexapod
SARJ	Solar Array Rotary Joint
SASA	S-Band Antenna Support Assembly Testbed
SCAN	Space Communication And Navigation Testbed
SDN	Starboard Deck Nadir
SDX	Starboard Deck X-Direction
SEDA	Space Environmental Data Acquisition Equipment
SMILES	Superconducting Sub-Millimeter Wave Limb Emission Sounder
SOLAR	Solar Observatory Grouping
SOX	Starboard Overhead X-Direction
SOZ	Starboard Overhead Zenith
SPDM	Special Purpose Dexterous Manipulator
Stbd	Starboard
Sx	SpaceX (US commercial resupply vehicle)
TBD	To Be Determined
TBR	To Be Resolved
TEA	Torque Equilibrium Attitude
TUS-RA	Trailing Umbilical System-Reel Assembly
ULF	Utilization & Logistics Flight
U.S.	United States
USOS	U.S. Operational Segment