

EXTERNAL PAYLOADS PROPOSER'S GUIDE to the International Space Station



**Goddard Space Flight Center
Greenbelt, Maryland**

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External Payloads Proposer's Guide to the International Space Station

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FOREWORD

The International Space Station (ISS) is a \$100 Billion science platform developed by NASA together with a host of other nations to provide both an earth like laboratory environment as well as a space (vacuum) environment for scientists to perform their science. The ISS resides in a 51.6 degree inclined low earth orbit with an average altitude of 400 km. This unique location provides the ISS with a repetitive good earth and space viewing along with a good microgravity environment for conducting life, physical, Earth and space science research. The ISS, being a science platform, provides all of the resources that a science package (or instrument) would expect from an earth like laboratory such as power, data, cooling, commanding and other accommodation needs.

The ISS Program has a very thorough integration and interface process that an instrument package goes through once it has been selected to be operated on ISS. That integration and interface process is well documented through several ISS specific documents. This ISS Proposers Guide is **not** an integration and interface document and therefore should not be used as such. It is specifically designed to help the scientists in their early stage of proposal writing and submission to NASA Announcement of Opportunity (AO) and other government agencies in the field of Earth Science, Heliophysics and Astrophysics. As such, this Proposers Guide is strictly for ISS external Payloads.

The information contains in this Proposers Guide regarding the ISS resource capability, performance and environment has been reviewed by the ISS Research Integration Office (RIO), formerly known as the "Payloads Office", to ensure that it is accurate and reflect the current state of ISS.

We strongly encourage the science community to make use of the ISS technical interface information contains in this Proposers Guide when developing their ISS instrument proposals, and, of course, the ISS Research Integration Office is always happy to help you if you need additional help using the information herein. Please don't hesitate to contact the RIO for any additional help.

Wishing you all the best with your proposals,



Date 4-27-15

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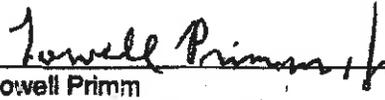
This Guide has been developed by the Systems Engineering Working Group (SEWG) of the Earth Systematic Missions Program (ESMP), the Earth Science Systems Pathfinders Program (ESSP), and the International Space Station (ISS) Research Integration Office (formerly known as "Payloads Office"). Several NASA developers of payloads for the ISS reviewed early drafts of the document and provided inputs. Participation of the ESSP included several members of the Common Instrument Interface team.

01 The intent of this document is to level the playing field in competitions for Earth Science funds, and to improve the quality of NASA Earth science, space science, technology demonstrations, and all other external payloads operating on the ISS. The problem being addressed is that the ISS has recently become much more available for hosting science payloads, but the science community has widely variable levels of experience and expertise to accommodations information about the ISS. This may unduly favor organizations with past experience in the development of ISS payloads. With this document, scientists and principal investigators and their engineering teams can more readily gain the level of understanding of the ISS interfaces and accommodations needed to develop a winning proposal. This improves the chances of the best science being done from the ISS, reduces workload within the proposal development workforce and reduces the expense of developing a proposal, reduces workload on those within the ISS Research Integration Office who work with proposal teams and review proposals, and reduces post-award surprises which might lead to cost over-runs or cancellation.

01 This initial release is under the configuration control of the ESMP and the GSFC Flight Projects Directorate, Earth Science Projects Division. To ensure you have the current version, visit the web site at the bottom of each page.

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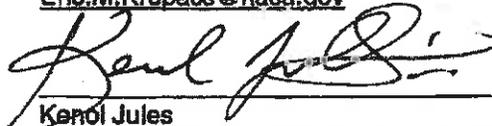
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1.0 Introduction

The Earth Science External Payload Proposers Guide for the International Space Station (ISS) (the Guide) provides proposers that are new to the ISS world an overview of the capabilities, accommodations, and requirements for operating on the ISS. Proposers typically include Principal Investigators, Payload Developers (PD), Systems Engineers, and others who are instrumental in developing a new proposal. The Guide is intended as a one-stop shop, with a supporting documents list, for developing proposals for operating external payloads on the ISS, and provides an overarching view of the ISS.

The ISS program provides an infrastructure capable of providing external payloads valuable short- to long-term access to space. The space station, the on-orbit crew, the launch and return vehicles, and the operation control centers all assist in supporting external payloads and their unique operations. The ISS is the only long-duration platform available in the relevant space environment with an integrated space systems architecture that can be used in this capacity.

External payload accommodations are provided at attach sites on the United States (U.S.)-provided Expedite the Processing of Experiments to the Space Station (EXPRESS) Logistics Carrier (ELC) on the Integrated Truss Assembly (ITA), the Japanese Experiment Module-Exposed Facility (JEM-EF) and the Columbus-External Payload Facility (Columbus-EPF).

The external attachment sites accommodate payload carriers on the multiple viewing sites, both zenith and nadir on the ITA sites, using the ELC. The JEM-EF, a back-porch-like attachment to the JEM pressurized module, accommodates multiple payloads, which may be serviced by the crew via the JEM dedicated robotic arm. The Columbus-EPF is another back-porch-like platform that can accommodate two zenith- and two nadir-looking payloads, using similar payload carrier platforms used on the truss sites. Each of these external accommodations are discussed in detail in subsequent sections of this document.

The Guide outlines the roles and responsibilities of several organizations with whom proposers will interface during the payload planning, development, integration, and operations processes. It provides a section dedicated to the “how to get started” process, having your payloads manifested for flight on the ISS, and whom to interface with within the NASA community. The Guide highlights the many products that proposers will either provide inputs to or develop for their own use and identifies services that are available from several NASA ISS Program Office organizations that proposers will use as part of the overall process for operating on the ISS. Finally, it lists several documents that the proposer may elect to review for further details in specific sections, and it provides a list of references that corresponds with the narrative portion of the Guide.

In addition, the Guide provides overviews of the payload physical and engineering integration processes, operations concepts and processes, an overview of both flight and ground safety requirements, and a document tree that provides names and links to important ISS documents. The Guide highlights hardware that will be required for flying on specific ISS accommodations and hardware that are made available to the PD in simulating ISS interfaces.

This Guide is primarily directed at first-time External Payload proposers. The Guide is NOT a design-to document. The Guide provides a level of detail that should be sufficient in developing a proposal. However, numerous ISS design-to documents exist within the ISS system and this document provides web links to some of these documents that eventually will be required for flying on the ISS.

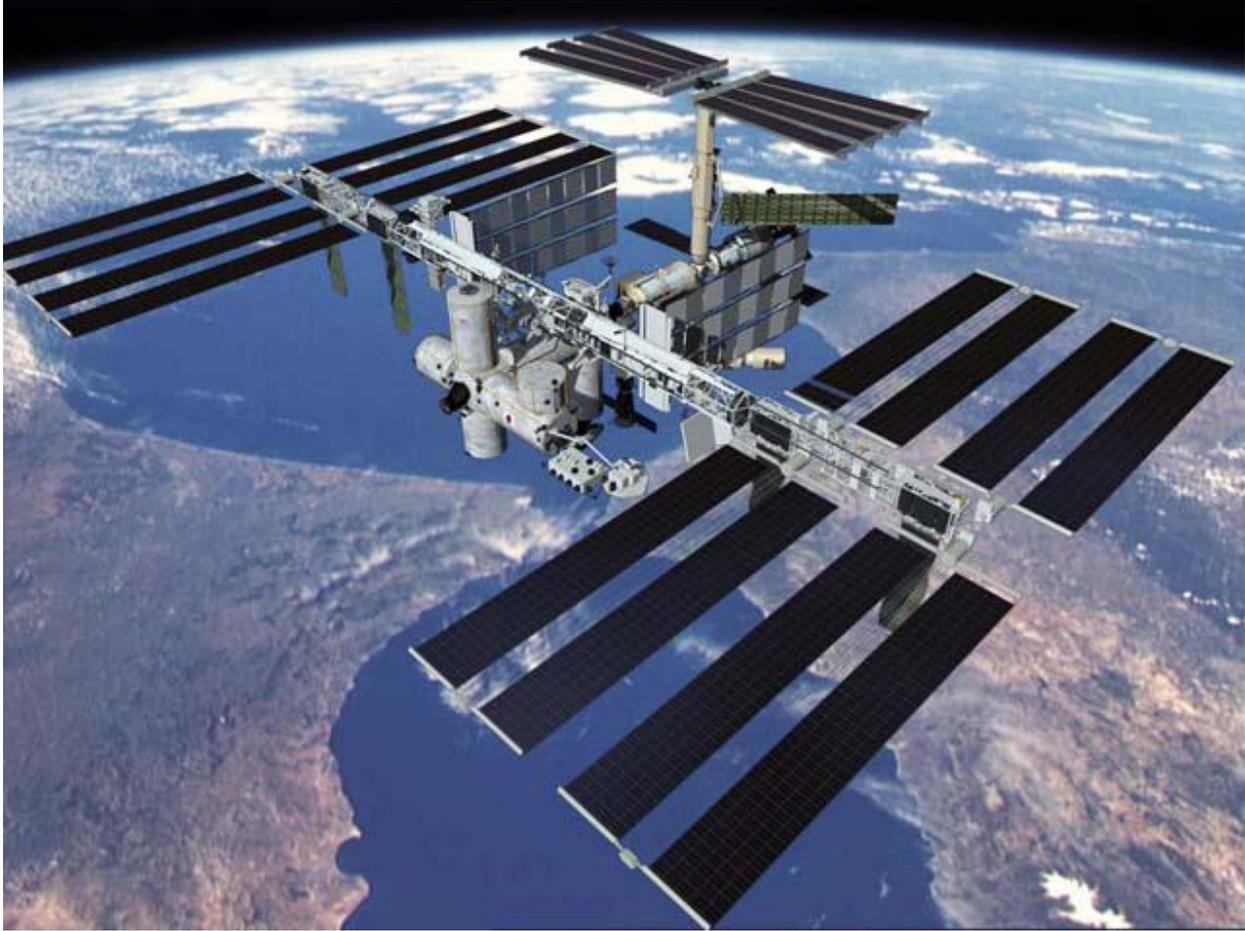


Figure 1.0-1. Welcome to the world of the International Space Station!

2.0 General Information for Operating on ISS

2.1 How to Get Started

The intent of the Proposer's Guide is to provide sufficient detail about the ISS accommodations and interfaces for you to formulate proposals properly, without having to delve deeply into the vast number of ISS payload documents. Examples of the Guide's technical content include; mass and volume allowances, power availability and limitations, data rates, platform stabilization rates, thermal conditions, electromagnetic compatibility (EMC)/electromagnetic interference (EMI) compatibility, observational geometry requirements, launch vibration constraints, etc. It also includes sections on integration, operations, safety, and a list of documents included as references.

The Guide is not a design-to document or a substitute for the vast amount of ISS payload accommodation and interface information that you will need to be familiar with to carry out detailed design, integration, test, and reviews if your proposal is selected.

During the proposal process, you will be required to provide an overview briefing to the ISS Research Integration Office (RIO), located at the Johnson Space Center (JSC), Houston, TX, containing your proposed payload concept. This briefing is important, as the ISS Payloads Office will then conduct a feasibility assessment based on your inputs. The assessment ultimately determines the feasibility of safely conducting your proposed mission on the ISS, and the results of the assessment play a critical part of the proposal approval process. In addition, the RIO will provide you specific information required for the feasibility assessment, the completion/submittal timetable, and any meeting or telecom attendance requirements. Subsequently, the ISS Payloads Office will issue you an assessment letter following internal review of your information. The assessment letter will address your specific accommodation needs, including any possible exceedances (e.g., on-orbit volume), and will state allowances. Finally, after the feasibility assessment is approved, your proposal must address management, risks, and costs in addition to meeting ISS technical and interface requirements.

The ISS RIO, and more specifically, the ISS NASA Research Office within the ISS RIO, will be your primary point of contact during both the proposal process, and following selection of your payload. For information and to schedule a payload concept briefing, please contact the Manager of that Office. If you are responding to an Announcement of Opportunity (AO), the AO will identify your ISS Program Office Point of Contact.

If selected, an ISS Payload Integration Manager (PIM) will be assigned to your payload by the ISS Program Office or the ISS RIO to address the specifics of what you need following proposal selection. The ISS has many location-specific requirements to review for more detailed design interface information. A document outline is provided in Section 9.0 for guidance on getting started. Two links available for ISS documentation are:

- http://www.nasa.gov/mission_pages/station/research/ops/research_information.html#.U6sC9RCiV2B
- <https://iss-www.jsc.nasa.gov/nwo/apps/edms/web/> (account application required)

It is assumed that most users of this Guide are responding to a NASA AO. AOs generally follow the Stand Alone Missions of Opportunity Notice (SALMON) format; see <http://nspires.nasaprs.com/external/>. This guide is only a formal part of an AO if it is specified to be part of the library of that particular AO.

Be aware that some of the ISS documents may be undergoing revisions in the form of Preliminary Intermediate Revision Notices (PIRN). Contact the ISS RIO (and later, during design activities, your PIM) to ensure any PIRNs do not adversely affect ISS documents in a way that could lead to difficulties and inconsistencies during your design and verification process.

2.2 What You Should Know

The three ISS external payloads sites are outlined in Section 3.0 of the Guide, and include accommodations, capabilities, and limitations for each. The three sites are: ITA using the ELC; JEM-EF; and the Columbus External Payload Facility (EPF).

The ISS interface hardware for the ELC and Columbus are very similar. Each site has a robotic interface and arm for installation/removal of payloads. The JEM-EF site has an active cooling loop for payload thermal control, whereas the ELC and Columbus sites are strictly thermally passive.

The ISS program supplies much of the necessary hardware to attach payloads at these locations at no cost to you. A more detailed list of ISS-provided hardware, and PD-supplied hardware, is contained in Section 7.0 and Section 8.0.

Your design considerations must meet ISS requirements for interfaces and must fit into specific mass and volume allocations defined for each ISS external payload site. These three sites also provide power and data connector access. In addition, JEM-EF provides thermal control.

On any of ISS external payload sites, your payload will have neighboring payloads. These neighboring payloads, and the overall ISS performance, may have operational constraints of their own that affect your payload location choice, installation/removal of payload, and/or science operation. Operational constraints will be considered by the ISS RIO in their feasibility study for your proposed payload, and throughout the payload manifesting process. This is a "good neighbor policy." This policy could impact your science expectations by, for example, limiting field of view (FOV), increasing contamination design sensitivity, decreasing on-orbit data downlink rates and power draw, increasing radiating/EMI/EMC sensitivities, or complicating thermal design. Ultimately, your design will need to conform to the parameters of flight, ground, and launch vehicle/visiting vehicle safety.

At this stage in the proposal process, you must be aware of a few limitations. First, your payload will require use of ISS robotics. For planning purposes, the ISS ITA and Columbus sites have no ISS-provided power for 6 hours while undergoing robotic manipulation. For a JEM-EF payload, there is no ISS-provided power for 7 hours. This power limitation is discussed in detail in Section 3.0. Second, carefully examine the ISS accommodations, outlined in Section 3.0, for each external site. If your payload requirement(s) fall substantially outside the accommodations listed in Section 3.0, contact the Manager, ISS NASA Research Office. If the requirement is such that it cannot be waived, this might be a determining factor on whether to continue with the proposal process.

The level of effort required for supporting ISS payload analytical and physical integration, and the ISS safety data process, is a substantial effort that requires heavy documentation. For example, if selected, your support of, the following will be required: design drawings and schematics of your payload and interface to the ISS; a payload Interface Control Document (ICD) and related ISS ICD's interface traceability; requirement documentation for all verification steps; computer aided design (CAD) and finite element models (FEM); materials lists; equipment lists; coupled loads analyses interaction and review of results; stress and thermal analyses for on orbit, robotics, and launch loads; payload to visiting vehicle hardware and

launch vehicle/visiting vehicle ICD information; thermal models; functional and physical configuration audits; safety, physical, and engineering documentation; and Kennedy Space Center (KSC) ground processing. The integration process is outlined in Section 4.1 of this Guide, and Section 6.0 provides details for the ISS safety process.

The ISS operations process requires your submittal of data sets and participation in operations discussions and meetings to ensure operations requirements are met. Section 4.2 outlines the requirements for ISS payload operations.

Consultation with the ISS RIO is recommended to assist you in producing accurate and complete estimates for your proposed labor, schedule, and cost. A complete understanding of the breadth of the interaction and deliverables with the ISS program will minimize surprises down the road. Again, Section 4.1 and Section 6.0 will assist you in determining the approximate level of effort required.

If it is necessary to waive a requirement in your proposal, which is determined to be vital to continuing with your proposal effort, please contact the Manager of ISS NASA Research Office or the point of contact named in the AO.

For selection of your launch vehicle/visiting vehicle and corresponding integration and safety processes, please refer to Section 5.0 of the Guide.

2.3 Roles/Responsibilities of ISS Organizations

ISS Payloads Office. The ISS RIO, located within the ISS Program Office, is located at JSC. The ISS Payloads Office, and more specifically, the Manager, ISS NASA Research Office, will serve as your interface during proposal formulation. After selection, a PIM will be assigned to serve as your primary point of contact for interfacing with the ISS.

The ISS RIO, through the PIM, also functions as the program/technical interface for the launch/visiting vehicle during the early stages of your design process and will provide direction with regard to any launch vehicle/visiting vehicle interface requirements that require your consideration in ground processing launch environments and on-orbit installation and removal operations. The ISS RIO is responsible for confirming the accuracy of the Guide and its endorsement.

Marshall Space Flight Center (MSFC) (Located in Huntsville, AL): The ISS operations function for ISS payloads has been designated to MSFC. Your interface with the ISS/MSFC team will commence during later stages of design/development as outlined in Section 4.2.

Kennedy Space Center (KSC, FL): KSC functions as the primary ISS facility interface for payload processing and final verification, and launch processing in coordination with the launch vehicle/visiting vehicle provider. Ground processing will involve coordination with NASA, NASA's launch site payload processing personnel, NASA's contracted launch services provider, and the launch range safety office.

Earth Systematic Mission Program Office (Goddard Space Flight Center (GSFC)) (Located in Greenbelt, MD): For questions about the contents of this Guide, please contact the GSFC, Earth Systematic Mission Program Office.

ESSP (Langley Research Center (LaRC)) (Located in Hampton, VA): ESSP participated in development of the Guide and is the primary intended user of this guide as a companion document to EV AOs.

Although the sponsoring organization for the JEM-EF is the Japan Aerospace Exploration Agency (JAXA) and the sponsoring organization for Columbus is the European Space Agency (ESA), your primary point of contact will continue to be the ISS Payloads Office (JSC).

2.4 Roles /Responsibilities of Payload Providers

Your responsibility is, simply, to provide the required technical, management, risk, safety, and cost information in your payload proposal to meet the ISS interface and safety requirements. You should strive to meet each ISS-required milestone, data submittal, hardware/software delivery, and timely flight to and integration on the ISS. For any questions on ISS accommodations or interfaces, contact the appropriate organizations as quickly as possible. The ISS program is fully prepared to help during the proposal process; however, it is your responsibility to develop the necessary knowledge of the ISS to formulate your proposal.

2.5 Manifesting Process

The goal of the ISS is to fly a payload as soon possible and to enable the flexibility for investigators to adapt their research plan based on new and unexpected findings. In addition, the ISS strives to make the integration and operation of payloads on ISS as simple as possible. During your payload's life cycle, you will provide inputs to ISS program forums that will enable the ISS program to find the appropriate ISS increment that supports your payload, in conjunction with your payload readiness date. At the completion of the process, the ISS program will add the payload to the Multi-Increment Resupply and Outfitting Manifest, which is the official ISS flight manifest.

3.0 ISS Accommodations

Introduction

This section contains very important information, and is the crux of the technical accommodations information that you will utilize for your proposal. External payload accommodations are provided on the U.S.-provided ELC, the JEM-EF and the Columbus-EPF.

The introductory section is presented as a comparative summary in tabular format. In the subsequent sections, each of the three ISS sites are presented in a more detailed fashion. Section 3.1 (ITA and ELC, specifically), Section 3.2 (JEM-EF), and Section 3.3 (Columbus-EPF) contain information needed for your proposal effort, along with additional references, as may be needed. Section 3.4 contains several factors common to all three ISS external sites, e.g. Micrometeoroid/Orbital Debris, Magnetic Field and Gravitational Field, Electro-Magnetic Radiation, Ionizing Radiation, Contamination, Atomic Oxygen, Contingency extravehicular activity (EVA) and End of Mission.

ISS Overview

The ISS consists of pressurized modules, external trusses, solar arrays and many other components. The ISS serves as a microgravity and space environment research laboratory in which crew members conduct experiments in numerous scientific disciplines: astronomy, biology, physics, and numerous microgravity research disciplines. It may also be utilized for the testing of spacecraft systems and equipment required for missions to the Moon and Mars.

Since November 2000, the ISS has been continuously occupied, which is the longest continuous human presence in space. The ISS a joint project among five participating space agencies: NASA, Russian Federal Space Agency (Roscosmos), JAXA, ESA, and the Canadian Space Agency (CSA). The ownership and use of the ISS is established by intergovernmental treaties and agreements. The ISS maintains an Earth-orbit with an altitude of between 330 km (205 mi) and 425 km (270 mi) by means of re-boost maneuvers, and it completes 15.51 orbits per day.

The Integrated Truss Segment (ITS) provides the backbone structure for the ISS. It attaches the solar and thermal control radiators to the rest of the complex; houses cable distribution trays, EVA support equipment such as hand-holds and lighting; and provides for extravehicular robotic (EVR) accommodations using the Mobile Servicing System (MSS). The ITS also provides logistics and maintenance as well as payload attachment sites. The attachment sites accommodate logistics and maintenance and payloads carriers, both zenith and nadir.

The ELC, which is attached to the ITS, can accommodate multiple payloads, and currently, there are four separate ELC sites on the ISS truss. The JEM-EF, a back-porch-like attachment to the JEM Pressurized Module (JEM-PM), accommodates external payloads, which can be serviced by the crew via the JEM-PM airlock and dedicated robotic arm. The Columbus-EPF is another back-porch-like platform that can accommodate two zenith- and two nadir-looking payloads.

There are ten JEM-EF external payload mounting locations for hosting external payloads. Five of the ten locations are allocated for NASA external payloads. None of the allocations are location specific. The Columbus-EPF can accommodate four external payloads, two of which are provided to NASA and are also not specified. Figure 3.0-1 shows the locations of the Columbus-EPF and the JEM-EF.

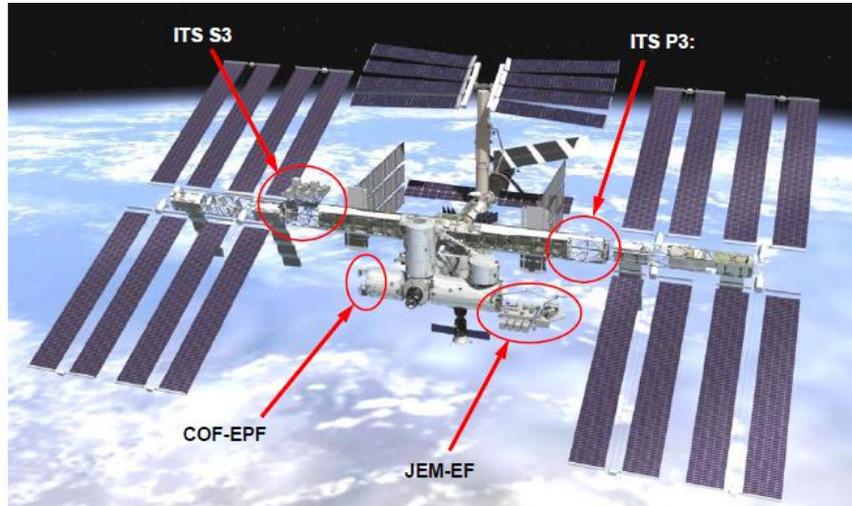


Figure 3.0-1. External payload attachment locations.

Although the ISS crew provides hands-on tasks for pressurized payloads, this capability is not generally offered to external payloads. The vast majority of external payloads are remotely operated from ground stations and from the PD's science facility.

As shown in Figure 3.0-2, the ISS maintains a 51.6° inclination, with the corresponding ground coverage area.

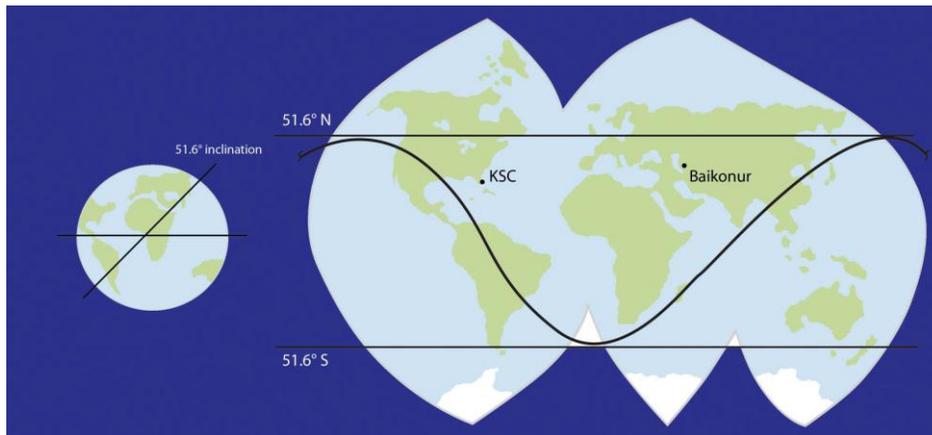


Figure 3.0-2. ISS orbit.

The ISS provides a myriad of capabilities and accommodations, some of which are outlined in Table 3.0-1.

Table 3.0-1. On-orbit payload resources.

Power	30 kw (average)
Internal Payload Racks	13 NASA Lab
	11 ESA Lab
	10 JAXA Lab

CHECK THE ESP DIVISION WEBSITE AT
<http://espd.gsfc.nasa.gov/isseppg/>
 TO VERIFY THAT THIS IS THE CORRECT VERSION PRIOR TO USE.

External Sites	8 NASA Truss ELC Platform Sites
	10 JAXA Platform Sites
	4 ESA Platform Sites
Crew Time	Exceeding 35 hours per week (average)

ISS C&DH Overview

The ISS Command and Data Handling (C&DH) system consists of hardware and software that provide services for command, control, and data distribution for all ISS systems, subsystems, and payloads. The top level (system-level) C&DH architecture contains redundant command and control (C&C) multiplexer/demultiplexers (MDMs) and MIL-STD-1553B control buses. The external payload services includes the payload MDM Low Rate Data Link (LRDL) (1553B local bus) data and command distribution, a High Rate Data Link (HRDL) for payload-to-payload communication and data downlink, and both a wired and wireless Ethernet Medium Rate Data Link (MRDL). LRDL (other than payload safety-related) data are downlinked via the HRDL to the ground. Safety-related data are routed via the C&C MDM to the S-band data services for downlink. The Portable Computer System (PCS) is used by the on-board crew for command and display interfaces. Payload commands can be uplinked from a ground site, issued from the PCS, or issued by a payload MDM automated procedure.

Enhanced features for payload communications include direct two-way Ku-band links between a user and their ISS instrument via Ethernet using standard internet protocols. These capabilities use the additional network services provided by the onboard Joint Station LAN (JSL), of which the payload Ethernet MRDL is a part, and the ground Payload Operations Integration Center (POIC). Refer to POIC Capabilities Document, SSP 50304.¹

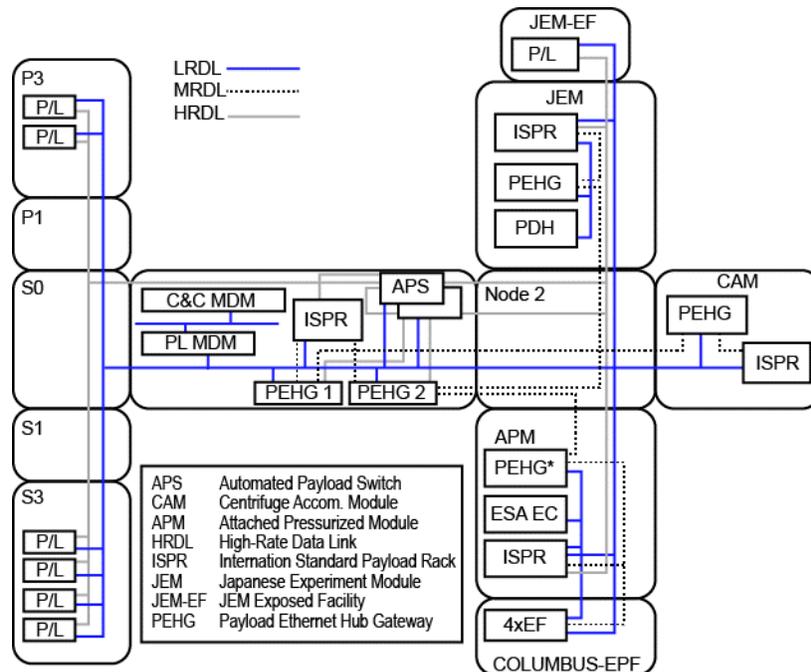


Figure 3.0-3. ISS C&DH Architecture Diagram.

Capabilities, such as power, microgravity environments, structural attachment interfaces, C&DH capabilities, etc. for each site, are discussed in the subsequent sections. Table 3.0-2, contains a brief comparison of each of the sites, but is summary in nature. You may use this table for top-level discriminations between the three ISS external sites, which might help to determine the

best site for your payload. More details for each external site are included in Sections 3.1, 3.2, and 3.3. In addition, the *Attached Payloads Accommodations Handbook*, SSP 57021 Rev. A², provides a design-to-level of information for your subsequent activities.

ISS Robotics Overview

The following provides a general overview to the ISS robotics systems, which each external payload will utilize. The proposer will need to have a basic understanding of these systems in order to provide a sound, technically accurate proposal.

The Special Purpose Dexterous Manipulator (SPDM), or Dextre, in conjunction with the Space Station Remote Manipulator System (SSRMS), is used to manipulate payloads or payload components. Dextre and the SSRMS are two external components of the MSS of the ISS. The other external components of the MSS are the Mobile Transporter and the Mobile Base System (MBS) which provide re-locatable accommodations for SSRMS and Dextre along the ITA. Internally, the Robotics Workstations are utilized by the onboard crew to operate the external components of the MSS. Direct robotic interactions with payloads are predominantly through Dextre, although for payloads with a mass greater than 600 kg (1320 lbs) the SSRMS will be primary manipulator, and all of the robotic payload operations are controlled by ground-based operators in Mission Control Center – Houston (MCC-H).

Dextre's construction consists of a Latching End Effector (LEE), the Enhanced ORU (Orbital Replacement Unit) Temporary Platform (EOTP), a Body Roll Joint, two arms, a tool holster, a Power and Data Grapple Fixture (PDGF), and four cameras.

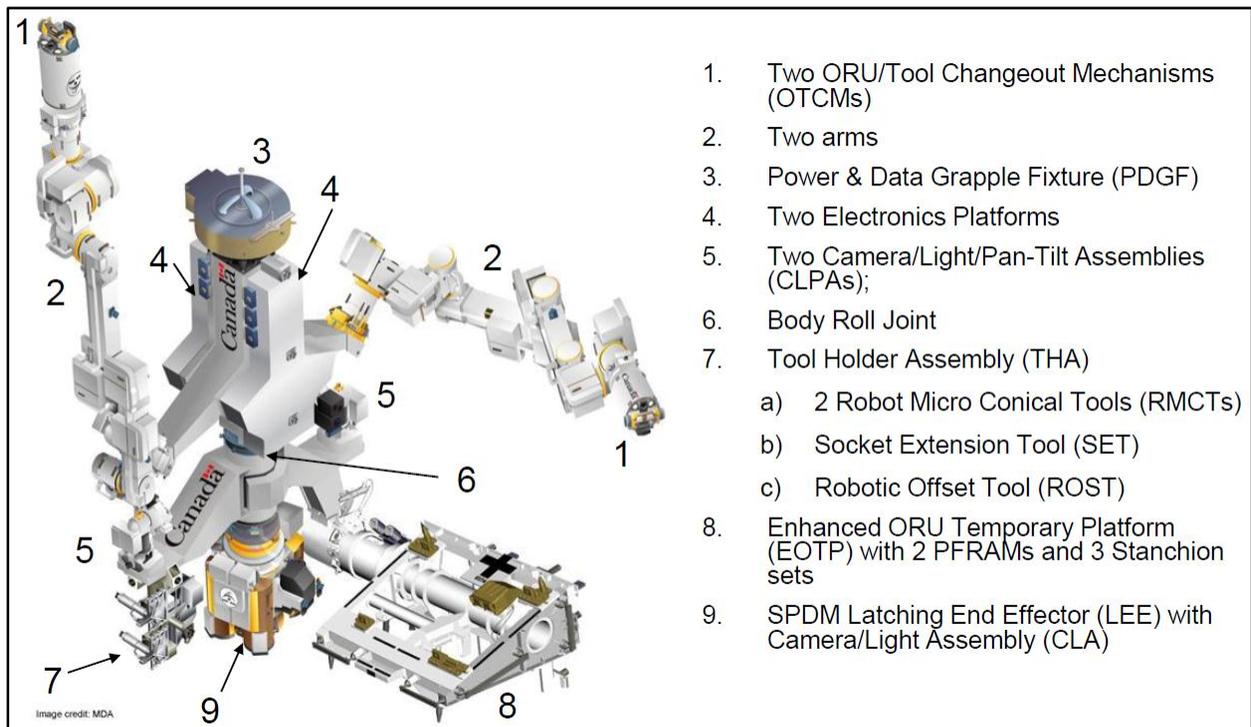


Figure 3.0-4. ISS Dextre.

Dextre's manipulation control is provided by two seven-jointed arms, which are both terminated with ORU/Tool Change-out Mechanisms (OTCMs). The OTCMs attach to specially designed fixtures on a payload and feature a retractable motorized socket wrench used to torque bolts, a retractable umbilical connector used to provide electrical, data, and video connections to

payloads, and a camera and lights used for close-up viewing and to align the OTCM with the fixtures using targets. Figure 3.0-5 illustrates the OTCM.

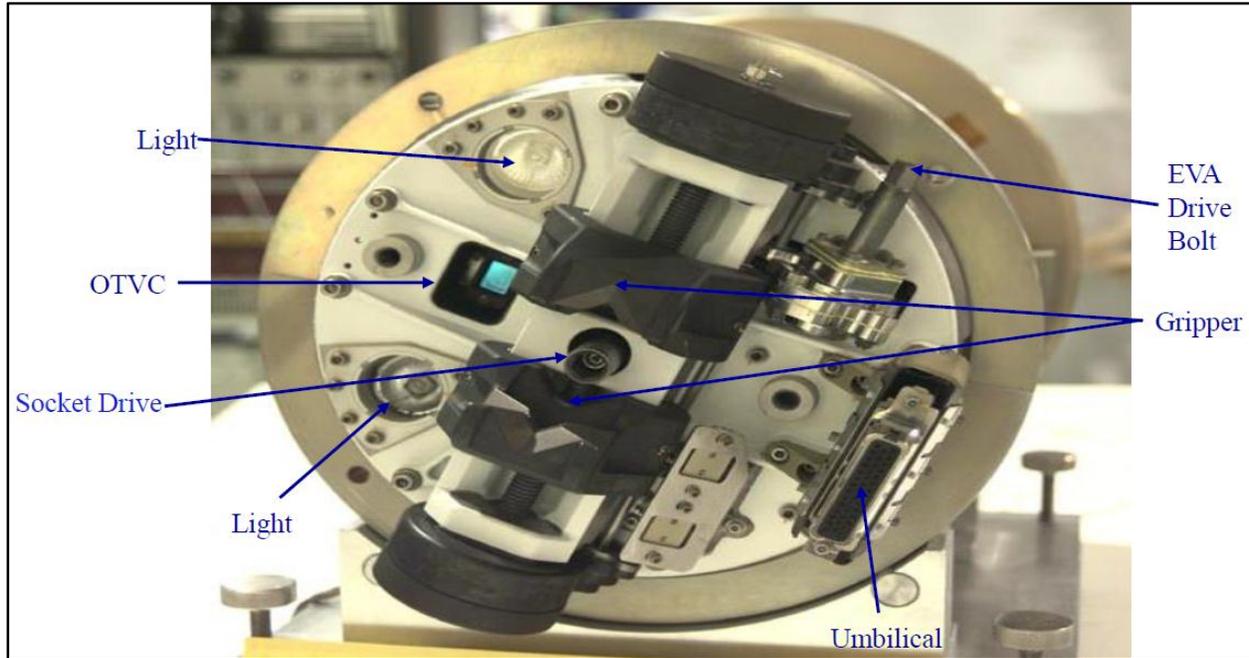


Figure 3.0-5. Dextre OTCM.

There are a number of interfaces available to allow interaction of Dextre with a payload. The various interfaces are shown in Figure 3.0-6. The standard interface is a Micro-fixture, which is also known as a Micro-square, which allows for a direct grasp by the OTCM. Having a Micro-fixture allows for the use of a collocated bolt as well as an umbilical connector for access to power, data, and video through the OTCM. In order to access the Micro-fixture, a payload must leave enough space around the fixture to accommodate the OTCM clearance envelope as shown in Figure 3.0-7. Like a Micro-fixture, an H-fixture also allows for a direct grasp with the OTCM as well as allowing for the use of a collocated bolt and umbilical connector. However, H-fixtures are usually only utilized on payloads for high-load or high-mass cases. For payloads that require Dextre interaction, but cannot accommodate the OTCM clearance envelope, a Micro-Conical Fitting (MCF) is available. The MCF requires the use of the Robot Micro-Conical Tool (RMCT) which has a much smaller clearance envelope as shown in Figure 3.0-8. The MCF allows for the use of a collocated bolt, but does not allow for the use of an umbilical connector. Acquiring the RMCT also has operational overhead which must be accounted for when planning payload operations.

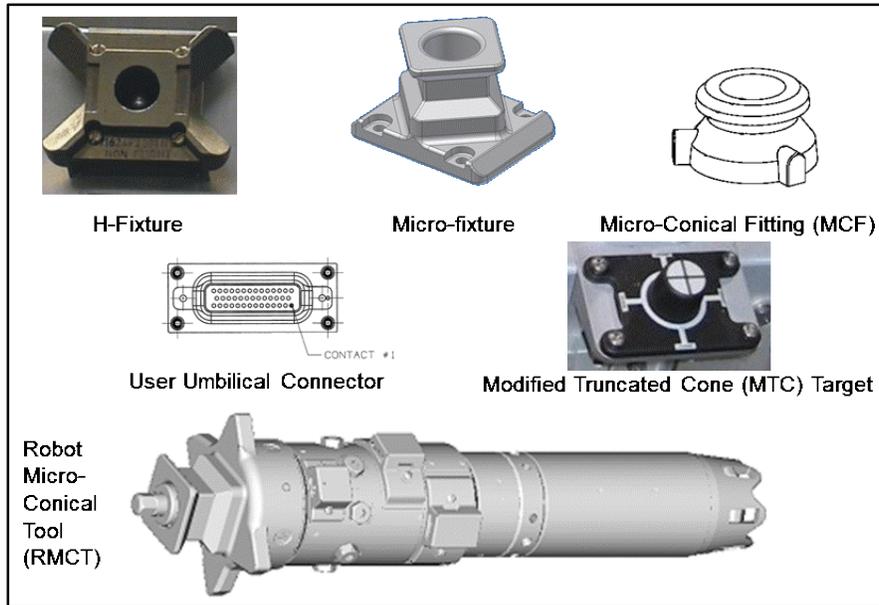
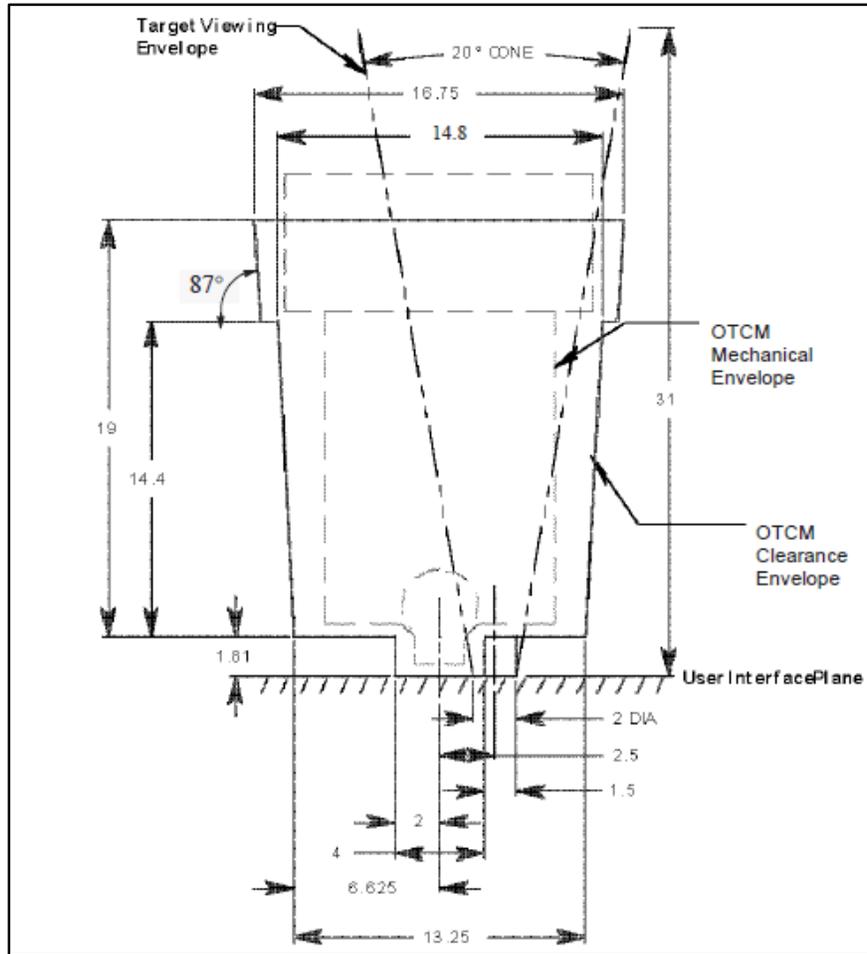


Figure 3.0-6. Dextre interfaces.



NOTE: Not to scale. All Linear dimensions in inches.

Figure 3.0-7. OTCM clearance envelope.

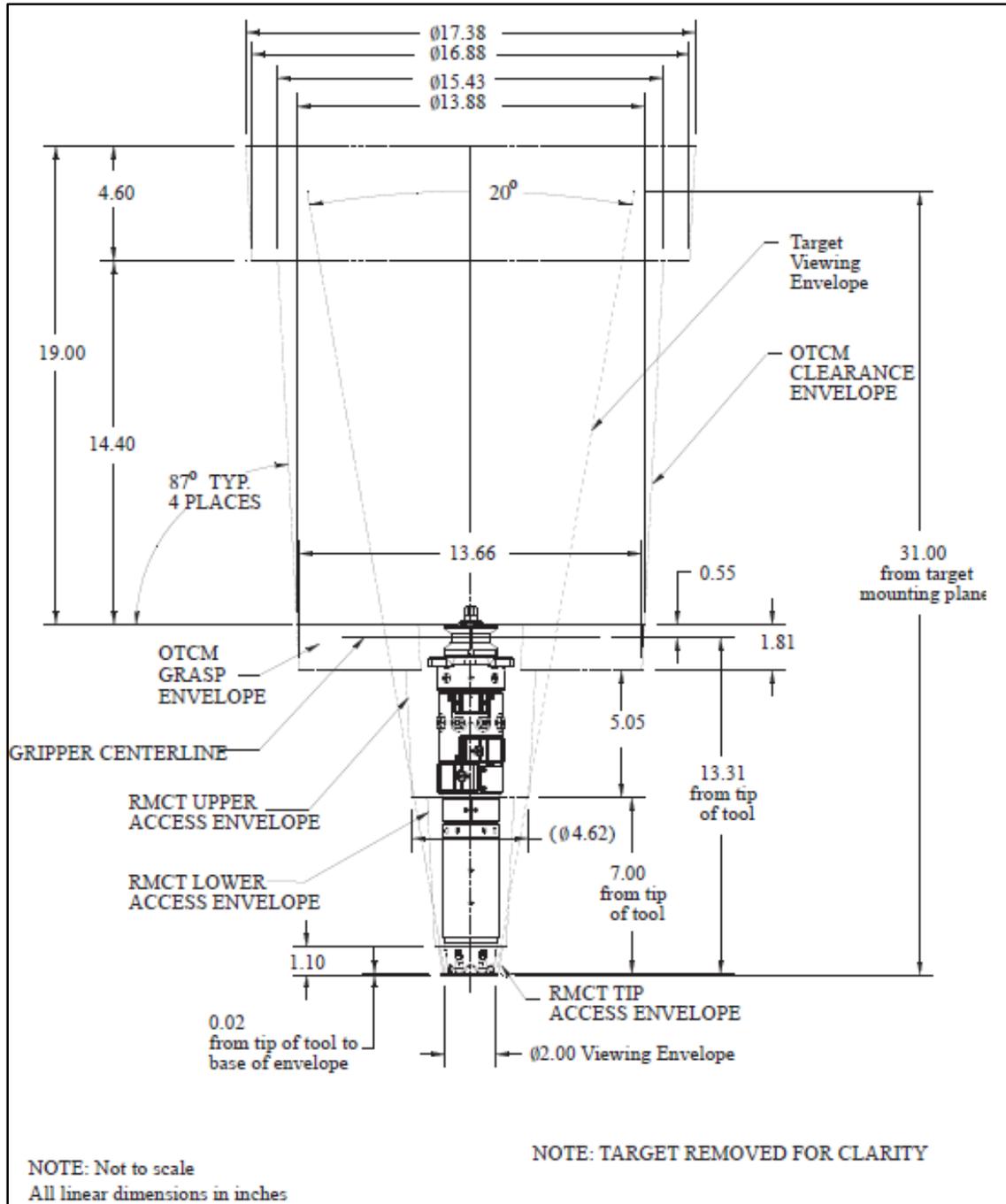


Figure 3.0-8. RMCT clearance envelope.

As previously mentioned, for payloads with a mass above 600 kg (1320 lbs), the SSRMS is required for manipulation of the payload. The SSRMS is a single seven-joint arm terminated on either end with two LEEs. There are four color cameras. Figure 3.0-9 illustrates the SSRMS. The SSRMS is symmetrical which allows it to “walk off” between seven base locations across the ISS (one on Node 2, one on the US Laboratory (USL), four on the MBS, and one of the Functional Cargo Block (FCB)).

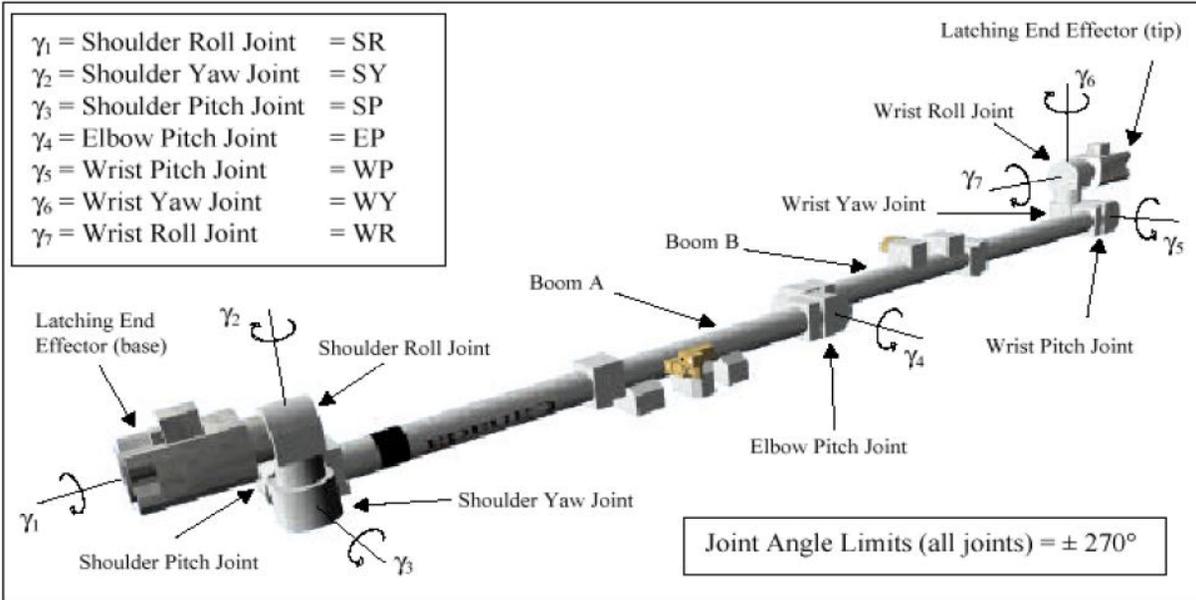


Figure 3.0-9. ISS SSRMS.

Interfacing with the SSRMS requires the use of one of the two types of grapple fixtures as shown in Figure 3.0-10. The standard interface is a Flight Releasable Grapple Fixture (FRGF). This type of grapple fixture only allows for a simple mechanical attachment with the SSRMS LEE. The other available fixture for interfacing with the SSRMS is the Power and Video Grapple Fixture (PVGF). A PVGF not only provides the simple mechanical interface of the FRGF, but also has connectors which provide access to the power, video, and data services through the SSRMS. The SSRMS is also latched to PVGF when utilizing the connectors which increase the rigidity between the SSRMS and the payload.

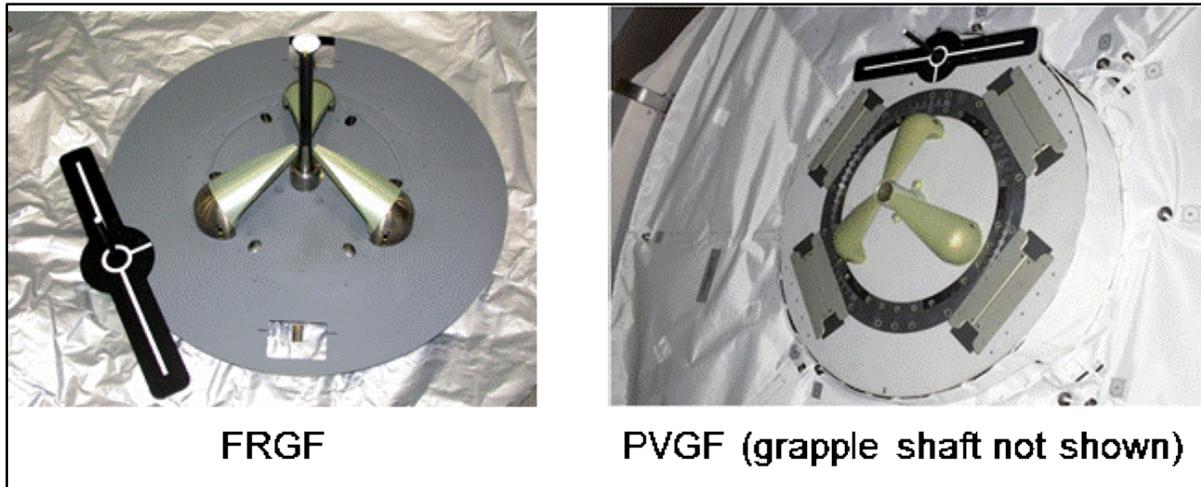


Figure 3.0-10. SSRMS grapple fixtures.

There are number of things that a PD needs to be cognizant of when being involved with the SSRMS and the SPDM:

- Unpowered survival – Unless a payload is equipped with an umbilical connector to access power through an arm, the payload will be unpowered during transport phases.

The required unpowered survival time is 6 hours for Flight Releasable Attach Mechanism (FRAM)-based payloads and 7 hours for EF-type payloads. The expectation is that a payload will “fail-safe” in the event power cannot be applied within the payload’s analyzed survival clock.

- Impact energy – While being manipulated by one of the robots, a payload can be at risk for impact when clearances between the payload and some other structure are within certain ranges (3 to 6 inches when SPDM is in motion, 12 inches when the SSRMS is in motion). A payload that interfaces with the SPDM is expected to be able to take a 1-Joule impact in any areas that are within the clearance ranges for each of the robots depending on which robot is in motion at any one time. For payloads interfacing with the SSRMS, there is an equation that determines the impact energy that could be imparted based on the payload’s mass. PDs are encouraged to evaluate their entire payload for impact to provide the maximum operational flexibility throughout the life of the payload, but specific areas of potential contact will be determined with the help of the robotics community. Areas at risk for impact are expected to “fail safe” and not result in a catastrophic failure.
- EOTP – All FRAM-based payloads are expected to protect for the use of the EOTP for their installation phase and their disposal phase. The main issue is electrical compatibility. Fuses need to be in place on the payload side to protect the EOTP’s 4A fuses which are only replaceable via EVA. Also, payload wiring needs to be such that, when installed on the EOTP, motors or any other type of reactive load are not being powered.
- Government-furnished equipment (GFE) – Hardware to interface with the robots is provided to the PD by NASA except for cases where the PD intends to modify the hardware. In that case, the hardware specifications are provided to the PD for use in building their modified hardware. Analysis, testing, and drawings of the modified hardware are expected as part of the verification and acceptance of the hardware for use with the robots.

Finally, other factors common to all three external sites are included in Section 3.4 of the Guide. These factors include: ISS Attitude and Pointing, Pointing Accuracy, Micrometeoroid and Orbital Debris, Magnetic Field and Gravitational Field, Electro-Magnetic Radiation, Contamination, Atomic Oxygen, Contingency EVA, and End of Mission.

Table 3.0-2. ISS external capabilities comparison.

Service/Location	ELC/ExPRESS Payload Adapter (ExPA)	JEM-EF	Columbus-EPF
Payload Mass lbs [kg]	500 [226.8]	1100 [500] @ standard payload locations 5500 [2500] @ heavy payload locations	500 [230]
Payload Volume	34x46x49 (H, in.) or 1 m ³	72.83x31.5x39.37 (H, in.) or 1.5 m ³	34x46x49 (H, in.) or 1 m ³
Thermal	Passive cooling, active heating (using PD-provided heating elements)	Active Cooling 3 kW @ standard-power locations, 6 kW @ high-power locations	Passive cooling, active heating (using PD-provided heating elements)
Power			
Operational Power watts	750 @ 113-126 Vdc 500 @ 28 Vdc per ExPA (adapter)	3-6 kW, 113-126 Vdc	2.5 kW total (shared)
Survival Power watts	Primary/Secondary of 300 @ 106.5 to 126.5 Vdc each	120 @ 110.5 to 126 Vdc	1.2 kW @ 120±7 Vdc shared between the payload compliment
Command & Data Handling			
Ethernet	One-way Ethernet (10 BASE T) link for downlinking ELC cargo science data.	Refer to MRDL	Refer to MRDL
Low-Rate Data Link	20 kbps (typical) telemetry downlink using MIL-STD-1553	20 kbps (typical) telemetry downlink using MIL-STD-1553	20 kbps (typical) telemetry downlink using MIL-STD-1553
High-Rate Data Link	N/A	User data rate determined by parsing the 100 Mbps encoded signaling rate with Sync symbols. One-way downlink only via optical interface.	32 Mbps maximum shared data rate in 32 Kbps increments. Data rate determined by parsing the 100-Mbps encoded signaling rate with Sync symbols. One-way downlink only via optical interface.
Medium-Rate Data Link (Wired)	N/A	Wired Ethernet (10/100 BASE T) for downlink and two-way Ku-band communication	Wired Ethernet (10/100 BASE T) for downlink and two-way Ku-band communication
Medium-Rate Data Link (Wireless)	Wireless Ethernet (3 Mbps typical) using radio frequency (RF) per IEEE 802.11n*	Wireless Ethernet (3 Mbps typical) using RF per IEEE 802.11n	Wireless Ethernet (3 Mbps typical) using RF per IEEE 802.11n
Sites for External Payloads	Four ELC providing eight sites for payloads (2 sites per ELC)	Five available to NASA	Two available to NASA

* IEEE 802.11 is a set of media access control (MAC) and physical layer (PHY) specifications for implementing wireless local area network (WLAN) computer communication.

3.1 Integrated Truss Assembly/ExPRESS Logistics Carrier

Introduction

The ISS ITA provides unique opportunities for external payloads. The six sites are very valuable. One site is devoted to the Alpha Magnetic Spectrometer (AMS), since 2011, and one site is devoted to External Stowage Platform (ESP)-3. The four remaining sites are devoted to ELCs. The ELC was developed by NASA as a carrier that can be utilized by several payloads at each of the four truss locations and is designed to carry a variety of ORUs, or spares, first-time outfitting cargo, and external payloads (science experiments). The ELCs are located on the ISS ITA as depicted in Figure 3.1-1. Fully integrated with mounted cargo/payloads, the ELC was delivered to the ISS via space shuttle and is a NASA-funded payload facility. External ELC payloads, replacing the payloads already on orbit, or utilizing unused ELC sites will now be delivered to the ISS on visiting vehicles, such as the Dragon spacecraft, which is launched currently by Falcon 9 v1.1.

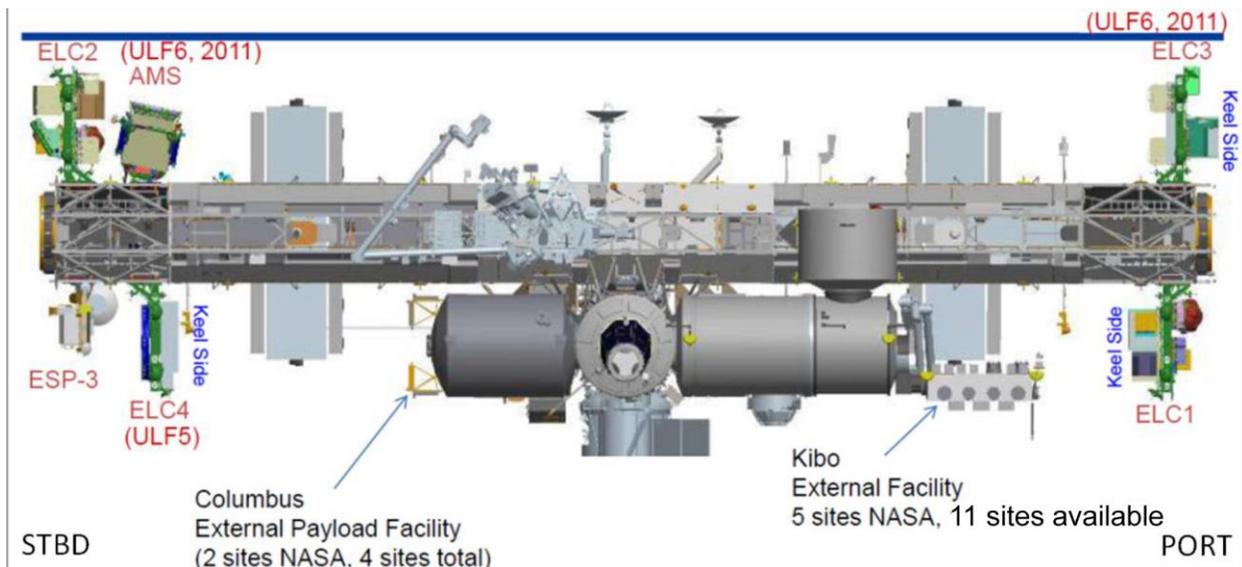


Figure 3.1-1. ELC locations on ISS.

The ELCs are located on both the port and starboard truss segments on the ISS and utilize the Mobile Remote Servicer Base System. ELC2 is starboard zenith, ELC4 is starboard nadir, ELC3 is port zenith, and ELC1 is port nadir. The ELC is one of the primary means of providing external accommodations to the science community. External payload proposers should evaluate the ELC as a potential accommodation site for their payload. For detailed-design information, please refer to the *Attached Payload Interface Requirements Document (IRD)-ELC*, SSP 57003-ELC³, and the *Attached Payload Interface Requirements Document*, SSP 57003⁴.

3.1.1 Physical Accommodations

There are four ELCs (ELC-1 through ELC-4) for use by PDs. Each ELC is equipped to carry two ExPRESS Payload Adapters (ExpPAs), which serve as the physical and electrical (inclusive of data connectivity) interface to external payloads. The ExPA Plate is a single piece of mounting hardware integral to the Active Flight Releasable Attach Mechanism (AFRAM) and the

payload mounting provisions. The ExPA Plate provides the mechanical and structural basis for the AFRAM assembly. The AFRAM serves as the attachment mechanism for connecting the ExPA (and your payload) to the Passive FRAM (PFRAM) that is already located on the ELC.

For further details regarding the ExPA, please refer to *ExPRESS Payload Adapter (ExPA) Interface Definition Document (IDD), Revision E, D683-97497-01 Revision E⁵*.

Details of the ELC locations are provided in Figures 3.1.1-1 and 3.1.1-2. The ELCs provide a total of eight external payload hosting locations. Your external payload will physically mount on an ExPA, which in turn, will be mounted on an ELC.

Your payload is limited to the envelope defined in Figure 3.1.1-3. This is both the launch and on-orbit envelope.

Your payload volume is centered according to Figure 3.1.1-4. This is both the launch and on-orbit envelope.

An illustration of the entire AFRAM and APFRAM adapter plate system is contained in Figure 3.1.1-5 in an expanded view format. This arrangement is also referred to as the ExPA.

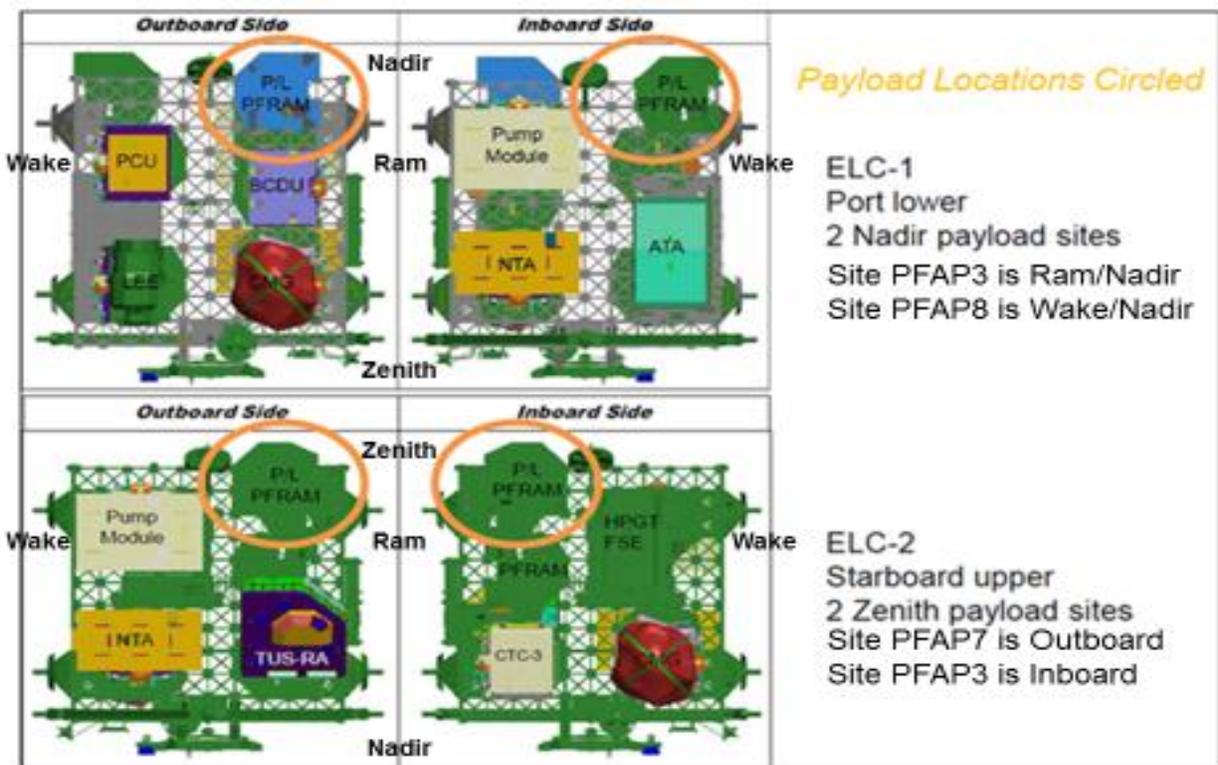


Figure 3.1.1-1. ELC location on ISS.

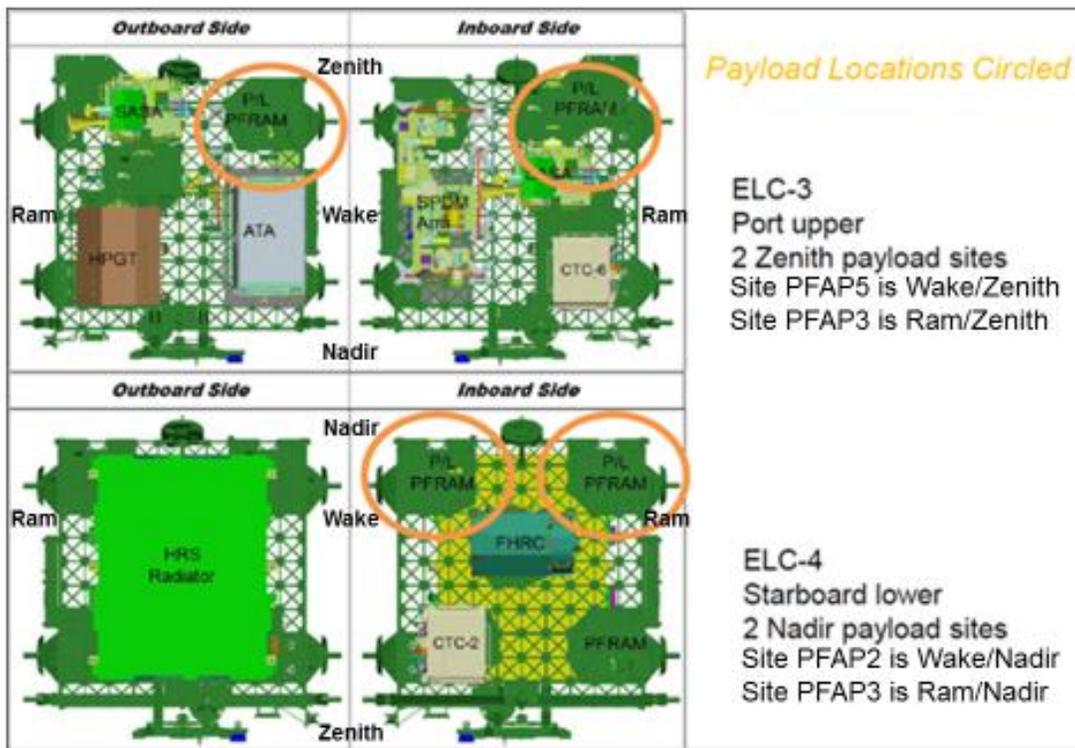


Figure 3.1.1-2. ELC location on ISS.

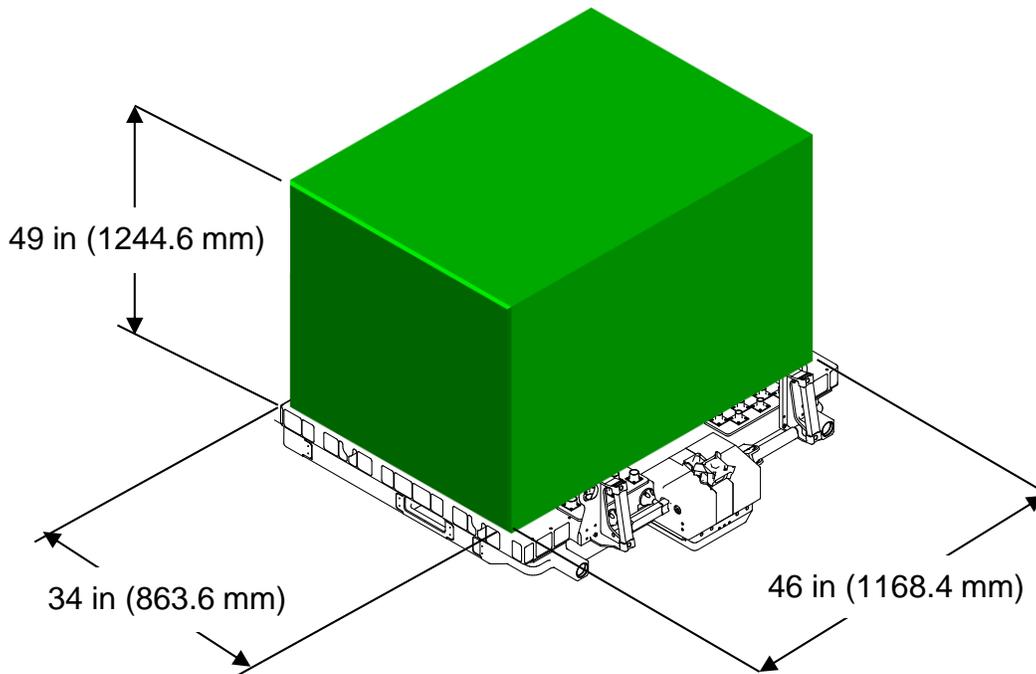


Figure 3.1.1-3. ExPA payload envelope.

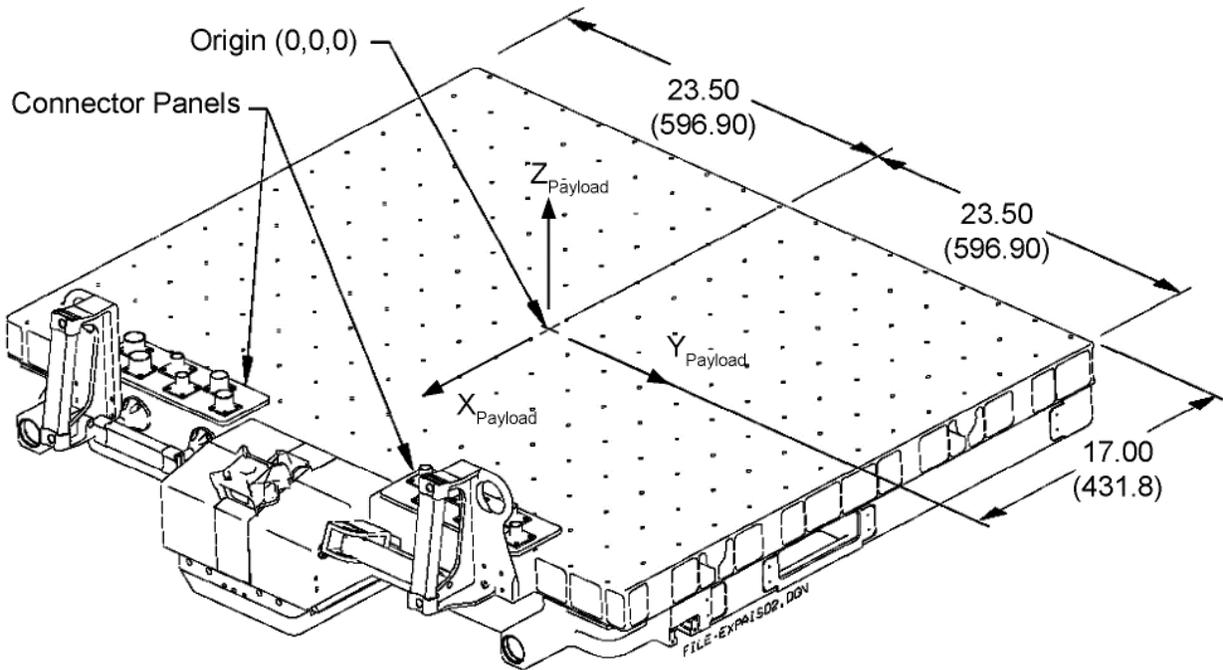


Figure 3.1.1-4. Generic ExPA payload coordinate system.

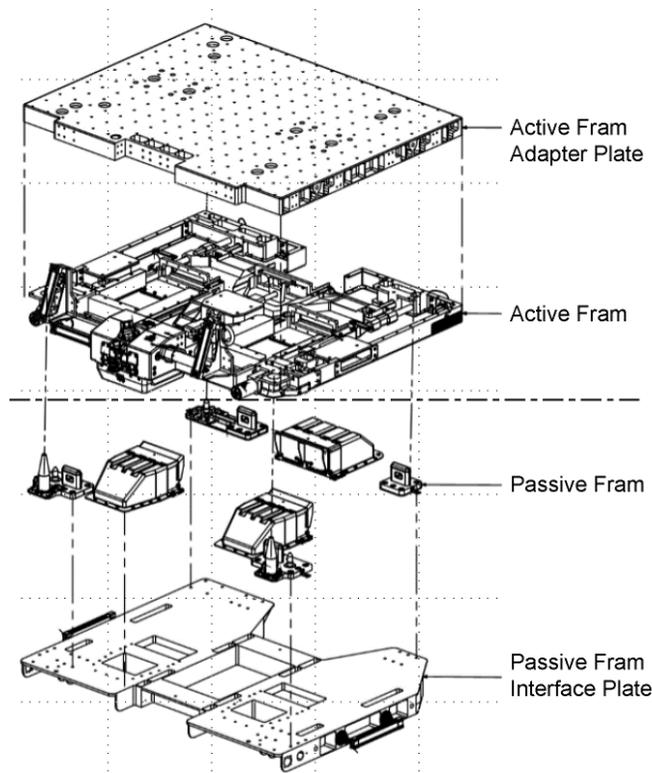


Figure 3.1.1-5. AFRAM and PFRAM expanded view (NOTE: AFRAM adapter plate and the AFRAM is considered a single unit).

The Payload Mass capabilities for each ExPA, excluding support equipment (e.g., ExPA/AFRAM, Flight Releasable Grapple Fixture, electrical connector, handrails, etc.) are specified in Table 3.1.1-1 based upon the payload center of gravity (C.G.) location.

Table 3.1.1-1. Allowable Mass and C.G. Location of ExPA Payloads.

Payload Mass lb [kg]	Maximum Deviation From Geometric Center In The X_{Payload} , Y_{Payload} Plane in [mm]	Maximum Height (Z_{Payload}) of Generic CEP CG Above The FRAM Plate Mounting Plane in [mm]
401 – 500 [181.9 – 226.8]	7.5, 7.5 [190, 190]	19.5 [495]
301 – 400 [136.5 – 181.4]	9, 10 [229, 254]	24.0 [610]
201 – 300 [91.2 – 136.1]	10.5, 12 [267, 305]	28.0 [711]
<200 [<90.7]	12, 14 [305, 356]	30.0 [762]

3.1.2 Robotics

Please review the robotics section contained in Section 3.0, prior to reviewing the specific robotic requirements/accommodations outlined below.

The SSRMS, also referred to as Canada Arm 2, is used in conjunction with the SPDM, or Dextre, to remove a payload from the docked visiting vehicle, transfer the payload to the ELC location, and perform the installation of the payload onto the appropriate external payload location.

Dextre can interface directly with an ExPA, by using the attachments contained on the ExPA. This interface should not impact payloads, if the PD stays within the defined mass and volume allocated on the ExPA. The primary purpose of the Dextre is to perform tasks that require fine, minute control. In the past, such tasks have been performed by EVA crewmembers, so being able to perform these tasks with Dextre removes the risk to EVA crewmembers. Dextre's construction consists of a Latching End Effector (LEE), the EOTP, a Body Roll Joint, two arms, a tool holster, a PDGF, and four cameras. Figure 3.1.2-1 illustrates the Dextre.

The Dextre's manipulation control is provided by two seven-jointed arms, which are both terminated with OOTCMs. The OTCMs attach to specially designed fixtures on the ExPA, and feature a retractable motorized socket wrench used to torque bolts, a retractable umbilical connector used to provide electrical, data and video connections to payloads, and a camera and lights used for close-up viewing and to align the OTCMs with the fixtures. Figure 3.1.2-2 illustrates the OTCM.

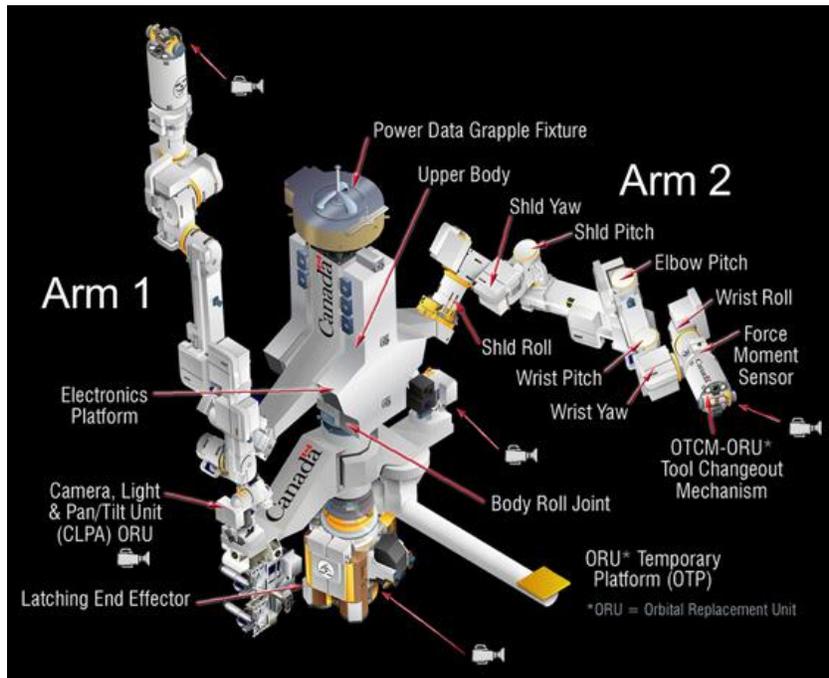


Figure 3.1.2-1. ISS Dextre.

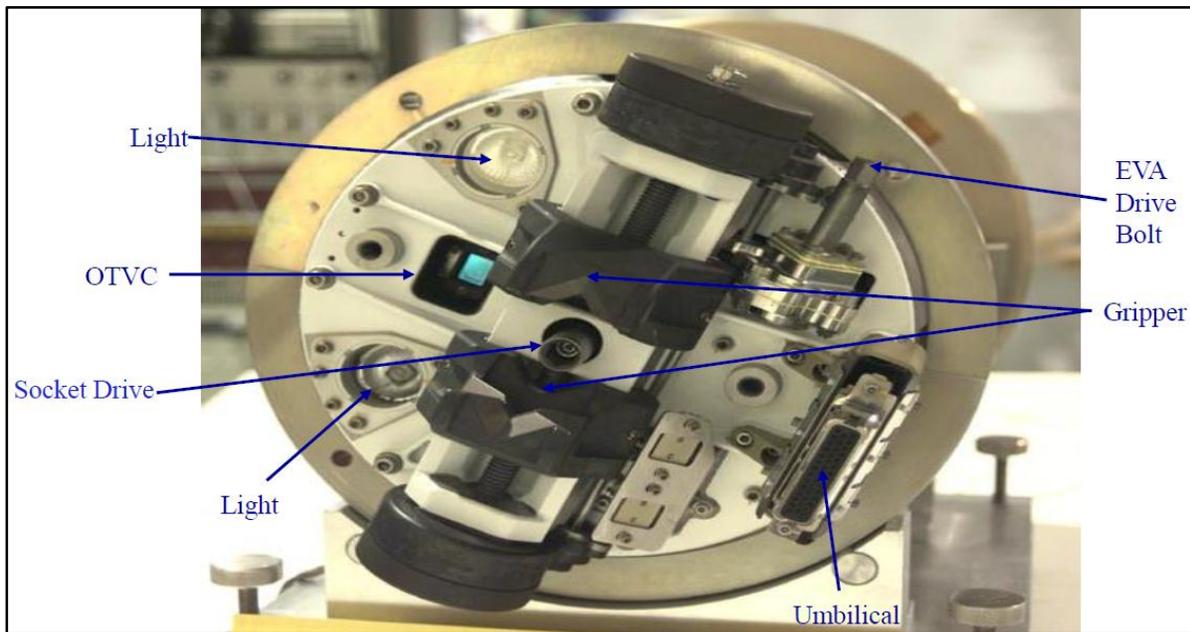


Figure 3.1.2-2. Dextre OTCM.

A SpaceX Dragon carrying an ExPA payload in the Dragon Trunk will be attached at the designated ISS node location. The SSRMS will pick up the Dextre robot, and bring it to near the Dragon. Dextre will reach into the Dragon trunk and an OTCM will engage the fixture on the ExPA (active FRAM portion) of the integrated attached payload. The payload will be released

from the Dragon by OTCM driving a bolt with the torque wrench to release the connector and the four clamps, and then Dextre will remove the payload. After bringing the external payload to the ELC site, it will be installed onto the appropriate external location. After release of the Dragon connector, the payload will have no survival heater power and must be able to survive a minimum of six (6) hours without power.

3.1.3 Thermal

You should note that the ELC payload sites do not provide active thermal control interfaces. Thus, your external attached payload will need to be compliant with the following key thermal parameters for interface with the ISS, you may refer to References 3 & 4. The external attached payload will need to be designed to rely solely on payload-based thermal control mechanisms, such as optical coating selection, insulating blankets, heater circuits, heat-pipe radiators, etc.

Unpowered Survival

All external payloads shall be able to survive six (6) hours without power in the translation configuration and during on-orbit power-downs (planned loss of power) (SSP 57003-ELC, Revision D, Section 3.4.4.1.1).³

External Payload Temperatures

The AFRAM and PFRAM thermal extremes shall have minimum and maximum temperature limits of $-93\text{ }^{\circ}\text{C}$ ($-135\text{ }^{\circ}\text{F}$) to $127\text{ }^{\circ}\text{C}$ ($260\text{ }^{\circ}\text{F}$) (SSP 57003-ELC, Revision D, Section 3.4.4.3.1).³

External Payload Temperature Constraints

The PD should plan for thermally conditioning the integrated external payload such that the maximum temperature differential between the AFRAM and PFRAM shall be no more than $102\text{ }^{\circ}\text{C}$ ($215\text{ }^{\circ}\text{F}$) (SSP 57003-ELC, Revision D, Section 3.4.4.3.1).³

Thermal Conduction

The external payload must not use the ELC as a heat sink and must not employ thermal control methods that reject heat to neighboring payloads (SSP 57003-ELC, Revision D, Section 3.4.4.3).³

Thermal Radiation

All external payload heat rejection requirements shall be met through radiation to the environment with all thermal radiators being confined to the FRAM payload envelope dimensions previously defined (SSP 57003-ELC, Revision D, Section 3.4.4.3).³

Payload Radiator View Factor

External payload active radiation surfaces (surfaces designed to reject heat generated by the payload) shall be oriented so that they have a cumulative view factor no greater than 0.1 to any ISS radiators or any surface of the generic envelope, as defined in Section 3.1, placed on any other S3 or P3 attachment site (SSP 57003-ELC, Revision D, Section 3.4.4.2).³

Payload Surface Specular Reflectivity

External payload surfaces with a view to any ISS or other external payload surface shall have a specular reflectivity of 10% or less (SSP 57003, Revision H, Section 3.4.3).⁴

Pressure

The on-orbit minimum pressure environment that the integrated external payload will experience is 1.0×10^{-7} Torr (1.333×10^{-5} Pa) (SSP 57003-ELC, Revision D, Section 3.5.2.2.1).³

Space Sink Temperature

The integrated external payload will be exposed to, and must be compatible with, a space sink temperature of 3 K (5.4 R).

Thermal Models

Key thermal considerations that must be addressed by the external PD are the development and delivery of a thermal model in required format and generation of thermal analyses to demonstrate that the payload will remain safe during Launch-To-Activation (LTA), as well as applicable on-orbit stowage sites on the ISS. These models will be provided to support required thermal assessment by the Launch Vehicle integrator and the ISS vehicle integrator.

The external PD will be required to develop and deliver a thermal model compatible with Thermal Desktop[®] and TRASYS/SINDA. The model is expected to be comprised of no more than 500 geometric and 500 mathematical nodes that represent all geometric and material properties required for determining critical hardware temperatures. These models will be utilized by the PD to assess the external payload in its on-orbit configuration, including the time periods while stowed in the Launch Vehicle and on the ISS ELC. These models will be provided to the ISS Program in order to support required thermal assessments by the Launch Vehicle integrator and the ISS vehicle integrator.

Thermal Analysis

The external attached payload must remain safe during the LTA phase while stowed in the Launch Vehicle, as well as, during the subsequent phase while it is stowed on an ISS ELC when it is exposed to a space sink temperature of 3 K, the thermal environment parameters defined in Table 3.1.3-1, the ISS flight attitudes and solar beta angle ranges defined in Table 3.1.3-2 (similar attitude requirements will be defined for the applicable Launch Vehicle) and the thermal interactions with all other on-orbit hardware. Verification of safe operation will require PD generated thermal analysis using ISS Program provided Launch Vehicle and ISS thermal models integrated with a PD-developed thermal model of the proposed payload.

Table 3.1.3-1. Thermal Environment Parameters

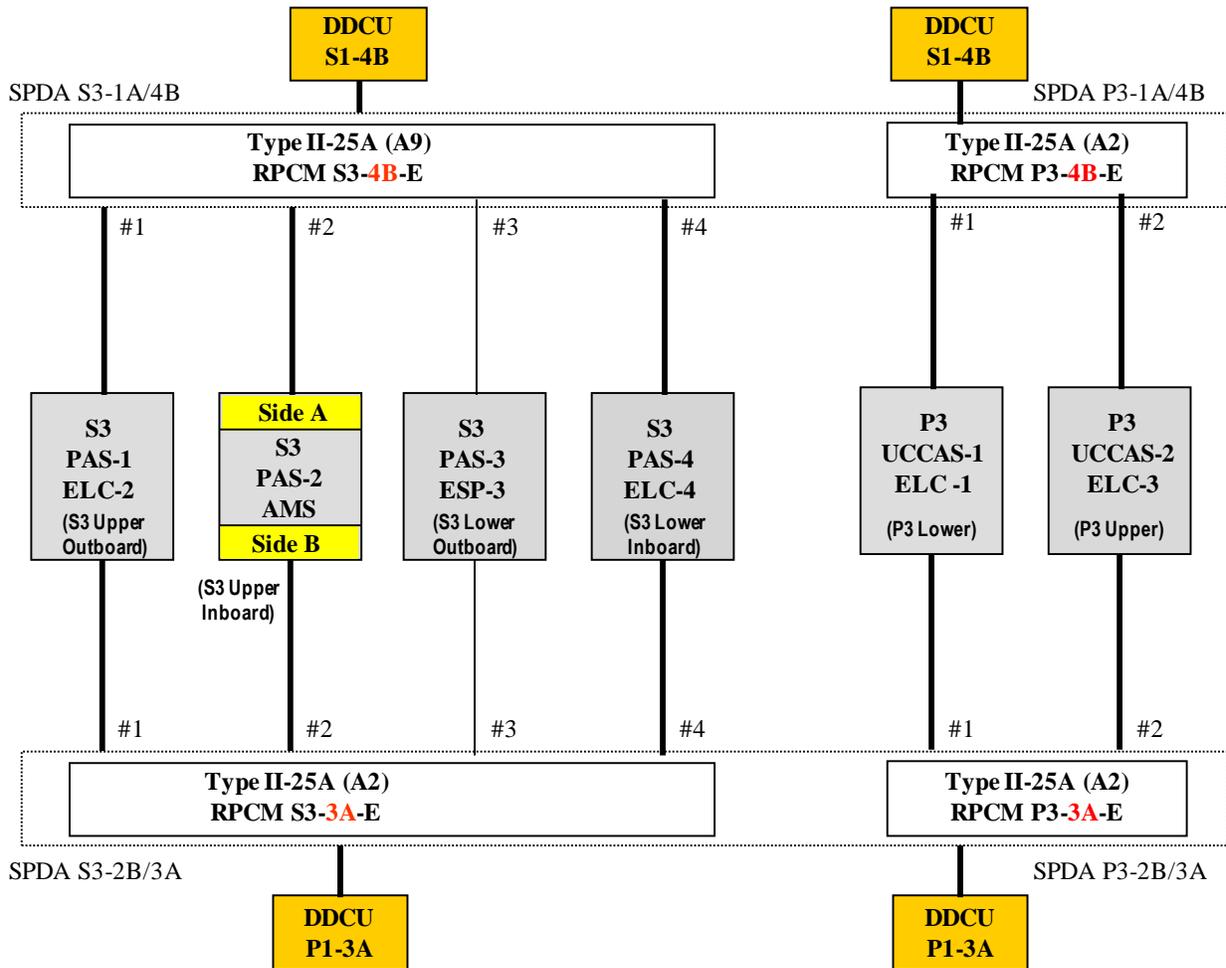
Case	Solar Constant (W/m ²)	Earth Albedo	Earth OLR (W/m ²)	Altitude (km)
Cold	1321	0.2	206	500
Hot	1423	0.4	286	278

Table 3.1.3-2. ISS Flight Attitudes

ISS Attitude Name	Attitude Reference Frame	Solar Beta Range (β)	Yaw	Pitch	Roll	Time in Attitude
+XVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	-15° to $+15^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
-XVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	$+165^\circ$ to $+195^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
+YVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +10^\circ$	-110° to -80°	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
-YVV +Z Nadir	LVLH	$-10^\circ \leq \beta \leq +75^\circ$	$+75^\circ$ to $+105^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
+ZVV -X Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	-15° to $+15^\circ$	$+75^\circ$ to $+105^\circ$	-15° to $+15^\circ$	3 Hours
-ZVV -X Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	$+165^\circ$ to $+195^\circ$	$+75^\circ$ to $+105^\circ$	-15° to $+15^\circ$	3 Hours

3.1.4 Power

The ISS provides two 3-kW feeds at each S3/P3 attach site, which provide power to the ELC interfaces. Each site must use one of the feeds as main (primary) power for operational use, and the other as auxiliary (keep-alive) power. External payloads may use both power feeds simultaneously provided the power bus isolation requirements are met. External payloads delivered to the ISS should nominally expect to be unpowered during transfer to the ISS for a minimum of 6 hours, as described in Section 3.1.2. Payloads that cannot withstand a lack of power for this duration must prearrange with the ISS for special accommodations or provide their own power source. Figure 3.1.4-1 illustrates the interfaces for the ITA sites, available for external payloads.



Note: The items that appear with a red marking are not to be utilized by external payloads.

Figure 3.1.4-1. ISS ITA interfaces.

Each ExPA payload will be provided with two +120 Vdc heater power buses, one +120 Vdc operational power bus and one operational +28 Vdc power bus.

The ExPA also contains the power and data connectors, which connect with the corresponding connectors already mounted on the ELC. The PD simply needs to provide the cabling, etc., required to mate with the ExPA connector. Figure 3.1.4-2 provides an electrical block diagram that represents accommodations between the ISS and ExPA.

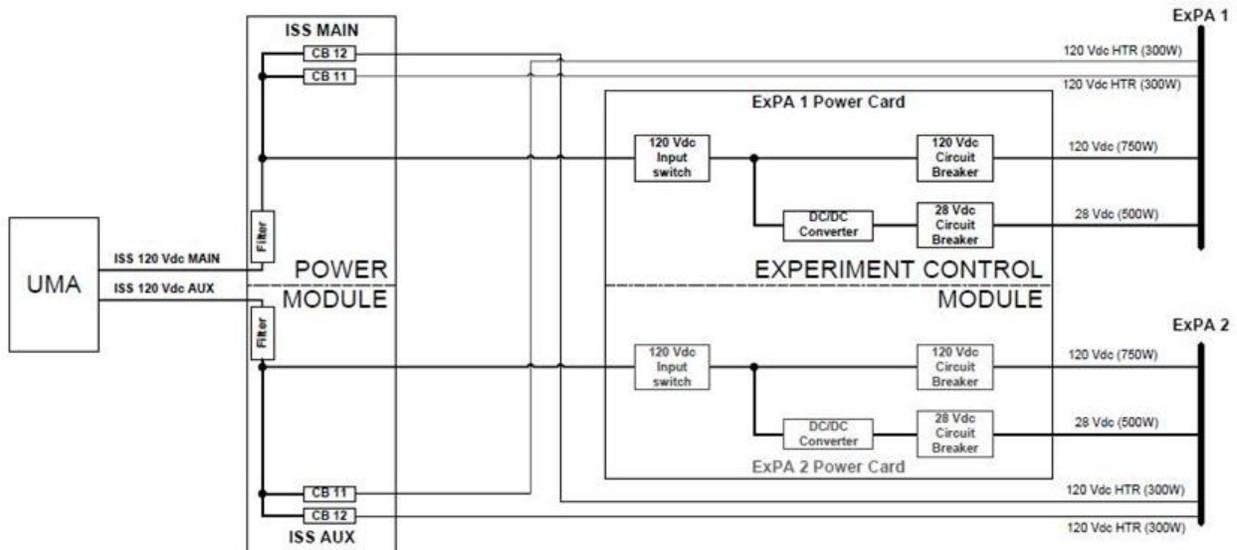


Figure 3.1.4-2. ISS to ExPA power block diagram.

+120 Vdc Heater Power

There are two +120 Vdc heater power services provided to each ExPA payload and these are only to be used for heaters systems. One of these services is considered to be primary and the other service is considered to be auxiliary and intended to be utilized as a backup heater power source in case of loss of primary 120 Vdc heater power. External payloads may use both power feeds at the same time for heaters systems. The heater power service provides the maximum 300 Watts power consumption. For more detailed power characteristics, such as ripple voltage and noise, non-normal voltage transients, surge current, load impedance, reverse current and wire de-rating, you may refer to Reference 3.

Note: External payload heaters systems which use bimetallic thermostats and resistive heaters to control temperatures do not require power characteristics verification as defined by Section 3.2.3.2.1 of SSP 57003-ELC.³ Refer to approved EPS-TIA-076 for details.

+120 Vdc Operational Power

There is a single +120 Vdc operational power service provided to each ExPA payload with the maximum 750 W power consumption. The steady state voltage varies from +106.5 Vdc to +126.5Vdc. For more detailed power characteristics, such as ripple voltage and noise, non-normal voltage transients, surge current, load impedance, large signal stability, reverse current and wire de-rating, you may refer to Reference 3.

+28 Vdc Operational Power

There is a single +28 Vdc operational power service provided to each ExPA payload with the maximum 500 W power consumption. The steady-state voltage of this power bus varies from +27 Vdc to +30.8. For more detailed power characteristics, such as turn-on voltage profile, voltage ripple, normal and abnormal transient voltages, transient current, current rise time,

current rate of change, reverse EMF limits and input impedance, ground isolation and compatibility with soft start, you may refer to Reference 3.

ExPA Power Connectors

There are four circular connectors provided to the payload for power interfaces. These are described in Table 3.1.4-2 by connector designator, total number of contacts, contact gauge, function, and mating connector part number. Note that J4 is not connected on-orbit and request for use of this connector for launch must be coordinated with NASA Visiting Vehicle Office at JSC. Refer to Reference 3 for pin assignments on each of the connectors.

Table 3.1.4-2. ExPA power connectors.

Connector Designator	Total # of Contacts	Contact Gauge	Function	Mating Connector P/N (or equivalent)
J1	4	12	+120 Vdc Operational Power	NATC06G15N4PN
J3	6	20	+120 Vdc Primary/Auxiliary Heater Power	NATC06G11N98PN
J4	9	12	+28 Vdc Heater Power (launch)	NATC06G19N96PN
J5	4	12	+28 Vdc Operational Power	NATC06G15N4PA

3.1.5 Command and Data Handling (C&DH)

There are various data interfaces available for FRAM-based cargo utilizing an ExPA. Figure 3.1.5-1 illustrates the ExPA experiment-to-ELC C&DH interfaces. For each ExPA payload, there are various data interfaces, namely, 1553 data bus, Ethernet, analog and digital (discrete) signals, and wireless MRDL for the PD to utilize. The specifics of these interfaces are defined in the following paragraphs. Note that there are no audio or video interfaces available for the ELC/ExPA experiment.

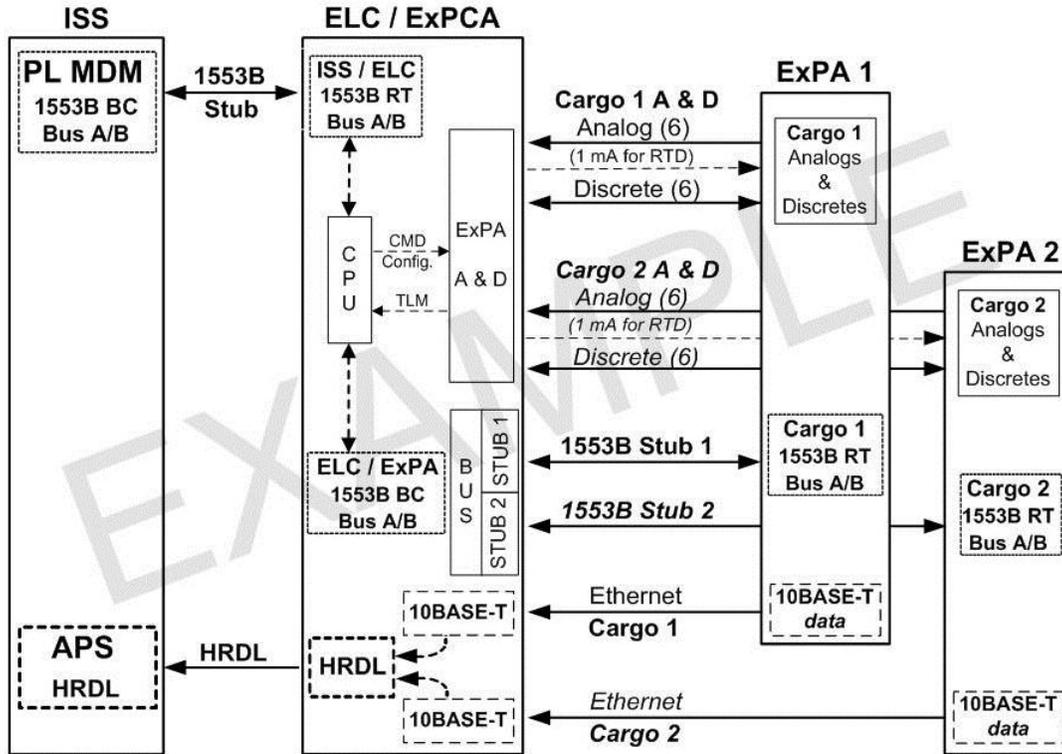


Figure 3.1.5-1. ISS to ExPA C&DH block diagram.

Low Rate Data Link

The EXPRESS Payload Control Assembly (ExPCA) provides a MIL-STD-1553B serial data bus for Low Rate Data Link (20 kbps typical telemetry throughput) for up to two ExPAs. This serial data bus will be used to exchange commands, broadcast time, request responses, Broadcast Ancillary Data (BAD), perform file transfers, and transmit payload health and status data, payload requests, and payload science data (when selected) for downlink. The ExPCA will operate as the 1553 Bus Controller (BC) and the payload will operate as the 1553 Remote Terminal (RT). The ExPA payload shall develop standard messages for the ELC ExPA/FRAM-based MIL-STD-1553 bus in accordance with International Standard Payload Rack to International Space Station, Software Interface Control Document Part 1, SSP 52050, Rev. K, Paragraph 3.2.3.3.⁶ Refer to Reference 3 for detailed characteristics such as RT address and parity inputs. Examples of BAD are available in Section 4.2 of this document.

Discrete

There are six discrete (interfaces) channels provided via the ELC ExPCA. The discrete channels are bi-directional and can be individually programmed, via 1553B commanding, to be input (to the ExPCA) or output (from the ExPCA) as single-ended signals. The sampling rate to/from ExPCA is 1 Hz. Discrete input interfaces operate as steady state while discrete output interfaces can operate as either steady state or pulsed.

Discrete steady-state outputs and inputs operate using +5 and +28 logic-level signals. Discrete pulsed outputs operate using +5 and +28 V from the ExPCA. A switch closure detection mode is also available via an ExPA provided excitation voltage of +2.4 Vdc to +6 Vdc at a maximum current of 40 mA \pm 5 mA. Refer to Reference 3 for detailed discrete interface characteristics.

Analog

Each payload will have six analog interfaces to provide temperature or preconditioned signal inputs to the ExPCA. The preconditioned signal inputs are configurable to either a current, or voltage, mode with an opportunity for the payload to multiplex signals in voltage mode. For temperature monitoring, the ExPCA will provide a 1-mA constant current to drive and monitor 1 K Ω , 2-wire Resistance Temperature Detectors (RTDs). Each analog channel (interface) will be sampled at a rate of 10 Hz and then combined into a composite output will be transmitted to the ISS at a 1-Hz rate. Refer to Paragraph 3.3.3.3 of Reference 3 for additional information regarding the analog interface capabilities, signal characteristics and the variable gain and offset programmability provided by the ExPCA. On-orbit changes to the configuration of the analog channels are possible through a ground-commanded upload of the ExPCA electrically erasable programmable read only memory (EEPROM).

Ethernet

Ethernet provides a one-way 10 Base T downlink for science data. The data stream is converted to HRDL by the ExPCA and sent to the ISS data system. Each ExPA can send data at up to 10 Mbps. The Ethernet frame content conforms with the Payload Ethernet Hub Gateway (PEHG) HRDL gateway Layer 2 format per SSP 52050 Rev. K, Section 3.3.5.1.⁶

Medium Rate Data Link (Wireless)

A MRDL path using a wireless Ethernet data link is available for ELC payloads. It consists of a two-way high data rate communications link using radio frequencies per IEEE 802.11n. This External Wireless Communication (EWC) system provides two-way data transfer between the ELC payload sites and the PEHG in the USL. Data rates available to the payload are dependent upon the specific ELC they attach to as described in Table 3.1.5-2. Average payload user data rates of 3Mbps or less typically are supportable. Higher rates are possible, but require coordination with other payloads.

Table 3.1.5-2. Wireless data link rates.

ELC	Total Received Power (dBm)	Link Margin (dB)	
Transmit (ELC to USL)			
ELC1	-68.24	-3.2	
ELC2	-66.46	-2.5	

ELC	Total Received Power (dBm)	Link Margin (dB)	
ELC3	-65.55	-2.5	
ELC4	-61.64	-1.3	
Receive (USL to ELC)			
ELC1	-62.24	-0.1	
ELC2	-60.46	0.1	
ELC3	-59.55	0.1	
ELC4	-55.64	0.7	

- This table is based on a 15-dBi payload antenna, net of cable losses, and 54-Mbps link rates. EWC is a shared resource and payload will negotiate operationally for a fraction of the channel capacity.

The NASA-provided EWC payload hardware GFE consists of the following items:

1. Payload Antenna Coax Cable Assembly, P/N 684-015637-0001 – Connects MOXA Wireless Access Point, EWC network interface card (NIC) to payload antenna.
2. EWC Payload Antenna, P/N 684-015636-001 – Transmits payload data to USL part of EWC system, a linearly polarized 17dBi 16° beamwidth antenna, 9” x 9” x 3”, must be pointed toward EWC system on USL.
3. EWC Network Interface Card, P/N 684-15644-001 – Universal Serial Bus (USB) form factor WiFi client, 2x2 Multiple-Input and Multiple-Output (MIMO), collects payload data for transmission by the payload antenna.
4. MOXA AWK-4131-M12-CT-T Wireless Access Point – Wireless Application Protocol (WAP) form factor WiFi client, 2x2 MIMO, collects payload data for transmission by the payload antenna.

The PD must identify if they are using the NASA provided EWC GFE or if they are developing their own Commercial Off The Shelf (COTS) solution to connect to the external wireless Ethernet. If the PD is using the EWC GFE they must define if they choose to interface the wireless Ethernet through either the NIC or the MOXA Wireless Access Point (items 3 and 4 listed above).

Finally, it should be noted that all RF operation around the ISS requires advance approval of the transmitting equipment through the ISS RF authorization process. The payload developer will submit the RF system characteristics and planned operating location information via the JSC Radio Frequency Spectrum Management Homepage (<http://ea.jsc.nasa.gov/webapp/fmdb/login.asp>). The initiation of this review process is recommended after System Requirements Review (SRR) and well ahead of Preliminary Design Review (PDR). The payload developer is also responsible for obtaining the required regulatory license grant and submit to the JSC Spectrum Management Office to support the completion of the ISS compatibility review process.

ExPA C&DH Connectors

There are six circular connectors provided to the payload for communications interfaces. These are described in Table 3.1.5-3 by connector designator, total number of contacts, contact gauge, function, and mating connector part number. Note that no signals are available thru the J8 connector on the FRAMs on the ELCs. Refer to Reference 3 document for pin assignments on each of the connectors.

Table 3.1.5-3. ExPA C&DH connectors.

Connector Designator	Total # of Contacts	Contact Gauge	Function	Mating Connector P/N (or equivalent)
J7	22	22	1553 Bus 1 and Ethernet	NATC06G13N35SA
J8	22	-	1553 Bus 2	-
J9	13	22	Analog Monitor Channel 1 - 3	NATC06G11N35SA
J10	13	22	Analog Monitor Channel 4 - 6	NATC06G11N35SB
J11	13	22	Discrete Monitor Channel 1 - 3	NATC06G11N35SC
J12	13	22	Discrete Monitor Channel 4 - 6	NATC06G11N35SN

3.1.6 Field Of View (FOV)

The following provides the proposer a series of fish-eye images of the FOV from those ELC ExPA/FRAM locations available to hosted NASA external payloads. The viewpoint location for these images (Figures 3.1.6-1 through 3.1.6-8) was defined with respect to the ExPA origin and is 22 inches in toward the ExPA center and 16 inches up. These images were generated by the MAGIK (Manipulator Analysis Graphics and Interactive Kinematics) Team software tools, at ISS/JSC. The solid objects in the FOV remain there at all times while the pink shaded areas in the FOV represent the areas swept by the ISS solar (photovoltaic) arrays as they track the sun. A summary of these fish-eye FOV images is provided in Table 3.1.6-1.

Table 3.1.6-1. Index of ELC fish-eye FOV images.

Carrier	Location	Site	Optimum Viewing Directions	ISS Aft FOV	ISS Forward FOV	ISS Nadir FOV	ISS Zenith FOV
ELC-1	P3 Lower	ExPA 3	Ram/Nadir	-	☐	☐	-
		ExPA 8	Wake/Nadir	☐	-	☐	-
ELC-2	S3 Upper	ExPA 7	Ram/Zenith	-	☐	-	☐
		ExPA 3	Ram/Zenith	-	☐	-	☐
ELC-3	P3 Upper	ExPA 5	Wake/Zenith	☐	-	-	☐
		ExPA 3	Ram/Zenith	-	☐	-	☐
ELC-4	S3 Lower	ExPA 3	Ram/Nadir	-	☐	☐	-
		ExPA 2	Wake/Nadir	☐	-	☐	-

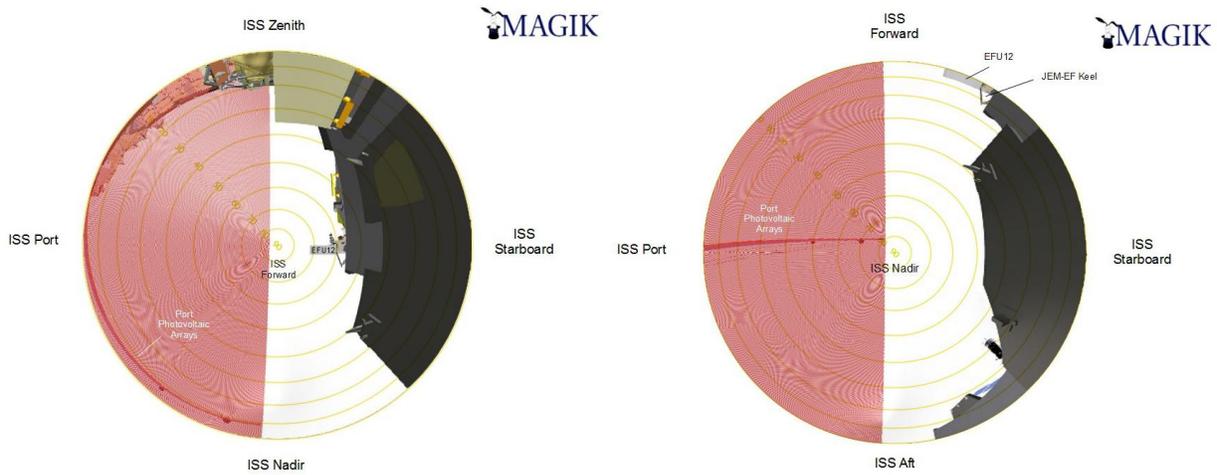


Figure 3.1.6-1. ELC-1 outboard ExPA payload accommodation forward (ram) and nadir.

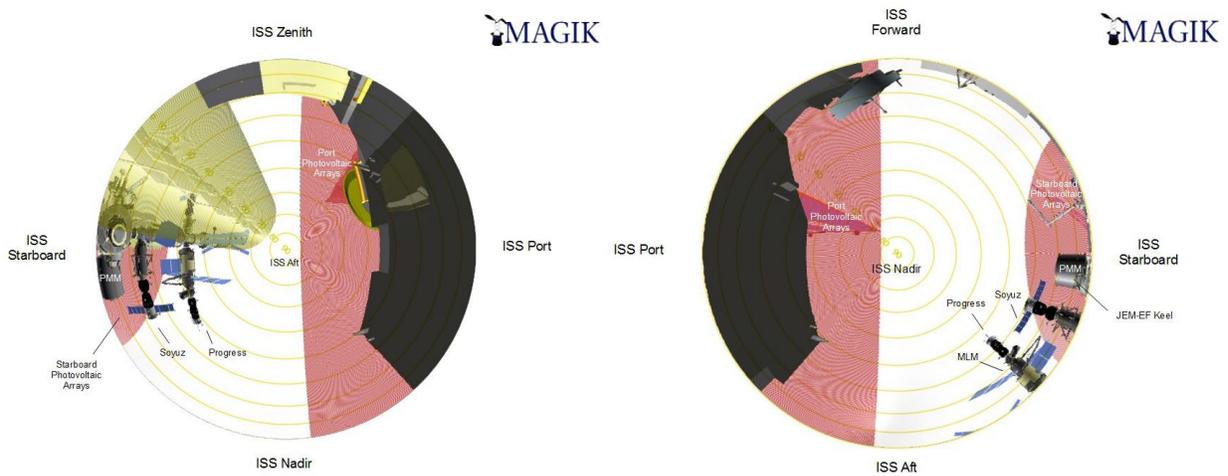


Figure 3.1.6-2. ELC-1 inboard ExPA payload accommodation aft (wake) and nadir.

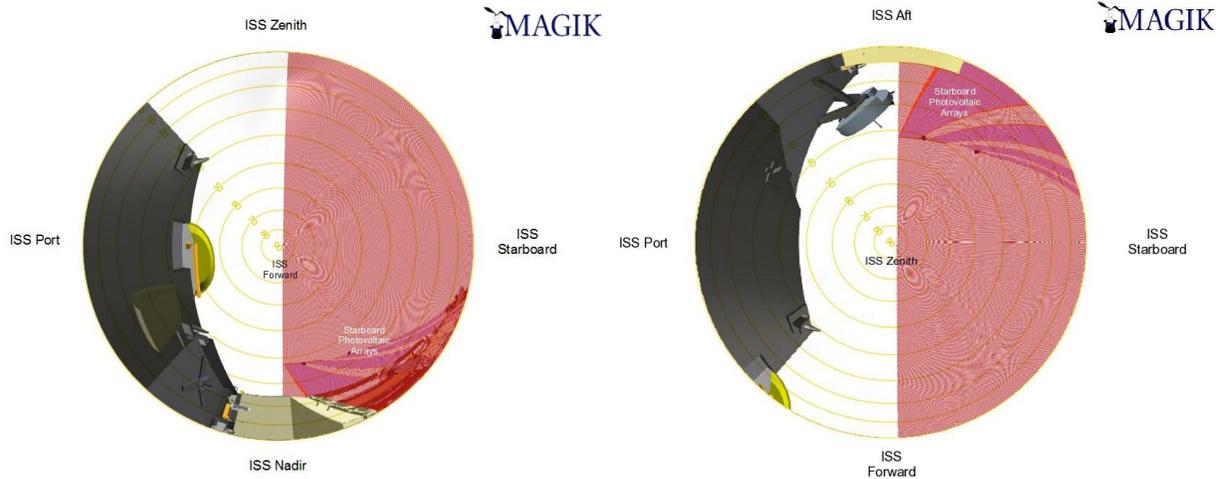


Figure 3.1.6-3. ELC-2 outboard ExPA payload accommodation forward (ram) and zenith.

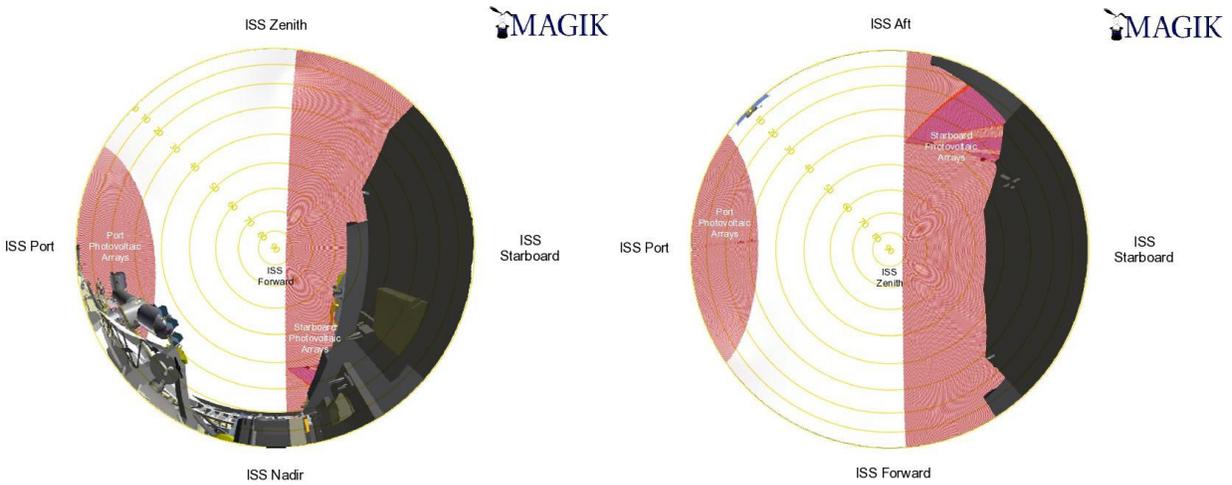


Figure 3.1.6-4. ELC-2 inboard ExPA payload accommodation forward (ram) and zenith.

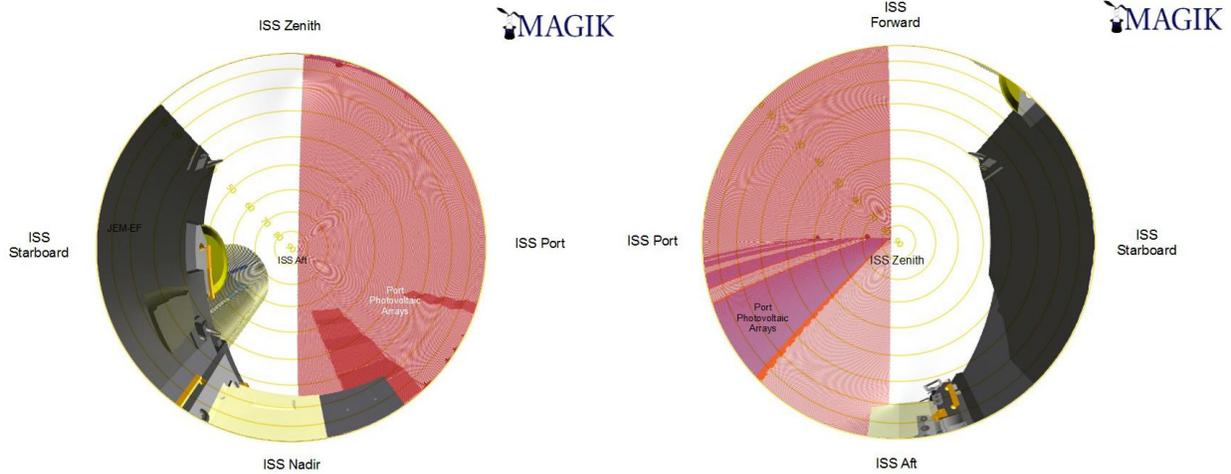


Figure 3.1.6-5. ELC-3 outboard ExPA payload accommodation aft (wake) and zenith.

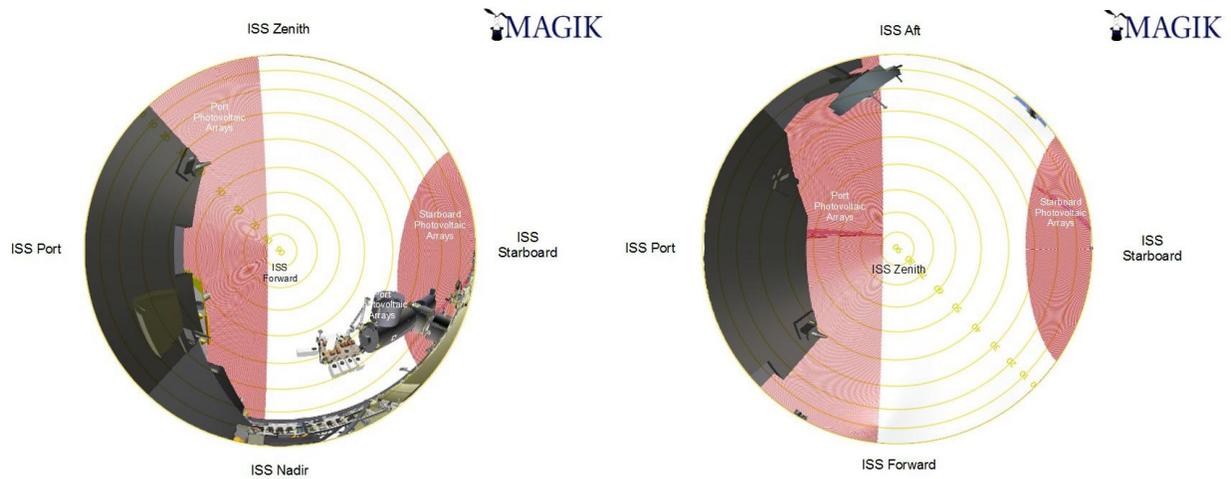


Figure 3.1.6-6. ELC-3 inboard ExPA payload accommodation forward (ram) and zenith FOVs.

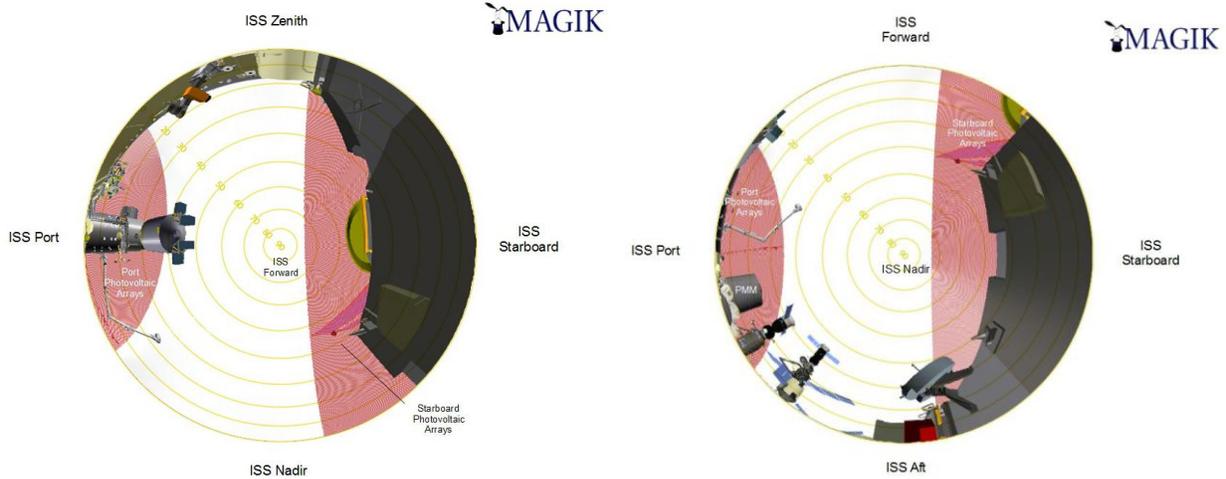


Figure 3.1.6-7. ELC-4 inboard ExPA payload accommodation forward (ram) and nadir FOVs.

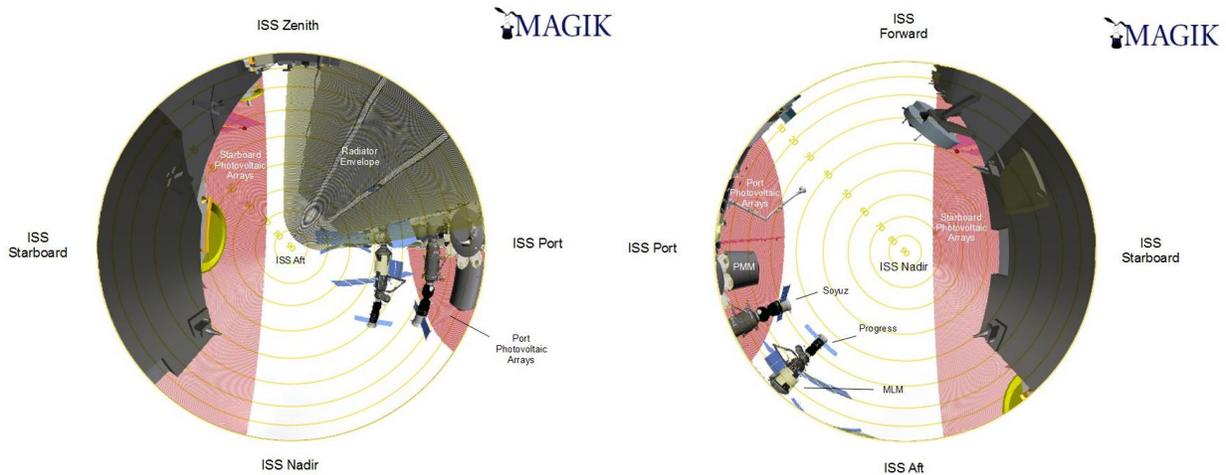


Figure 3.1.6-8. ELC-4 inboard ExPA payload accommodation aft (wake) and nadir FOVs.

3.1.7 Microgravity Environment

The mechanical vibrations generated from all of the ISS systems are transmitted throughout the structure, and may affect experiments sensitive to acceleration. Attached payloads are required to restrict their mechanical vibrations so as not to disturb sensitive experiments while the ISS is in microgravity mode. To assure that ELC payloads are able to survive these mechanical vibrations, payloads shall be designed to withstand the on-orbit vibration environment in Table 3.1.7-1. Please note that during nominal operating conditions, additional safety critical loads requirements are levied through ELC ICD.

Table 3.1.7-1. On-orbit random vibration environment.

Frequency (Hz)	Level
10–50	0.0005 g ² /Hz
50–100	+3 dB/oct
100–1000	0.001 g ² /Hz
1000–2000	–3 dB/oct
2000	0.0005 g ² /Hz
Composite	1.3 grms
Duration	10 hr/Year

Note: Accelerations must be evaluated as acting one axis at a time in each of three orthogonal axes.

Microgravity Disturbances

Microgravity disturbances induced by external payloads are separated into the three categories; quasi-steady for frequencies below 0.01 Hz; vibratory for frequencies between 0.01 Hz; and 300 Hz and transients. These requirements are defined to limit the disturbances to the microgravity environment of other payloads during microgravity mode periods and are not applicable during servicing, crew activities, maintenance, and reconfiguration. The microgravity disturbances induced by the external payload are not applicable during servicing, crew activities, maintenance, and reconfiguration. The microgravity environment induced by the external payload must not exceed the following levels.

Quasi–Steady Requirement

For frequencies below 0.01 Hz, payloads shall limit unbalanced translational average impulse to generate <10 lbs (44.5 N–s) within any 10- to 500-second period, along any ISS coordinate system vector.

Vibratory Requirement

For frequencies between 0.01 and 300 Hz, payloads shall limit vibration as defined in Paragraph 3.1.3.2.6.2.1 of Reference 3.

Transient Requirements

1. Payloads shall limit force applied to the ISS over any 10-second period to an impulse ≤10 lbs (44.5 N–s).
2. Payloads shall limit their peak force applied to the ISS to <1000 lbs (4448 N) for any duration.

Meeting the above transient requirements does not obviate the need to also meet the 100-second vibration requirement for vibration included in and following the transient disturbance.

Microgravity Environment Profile

Figures 3.1.7-1 and 3.1.7-2 depict typical microgravity environment profiles for the ELC/ITA location on ISS.

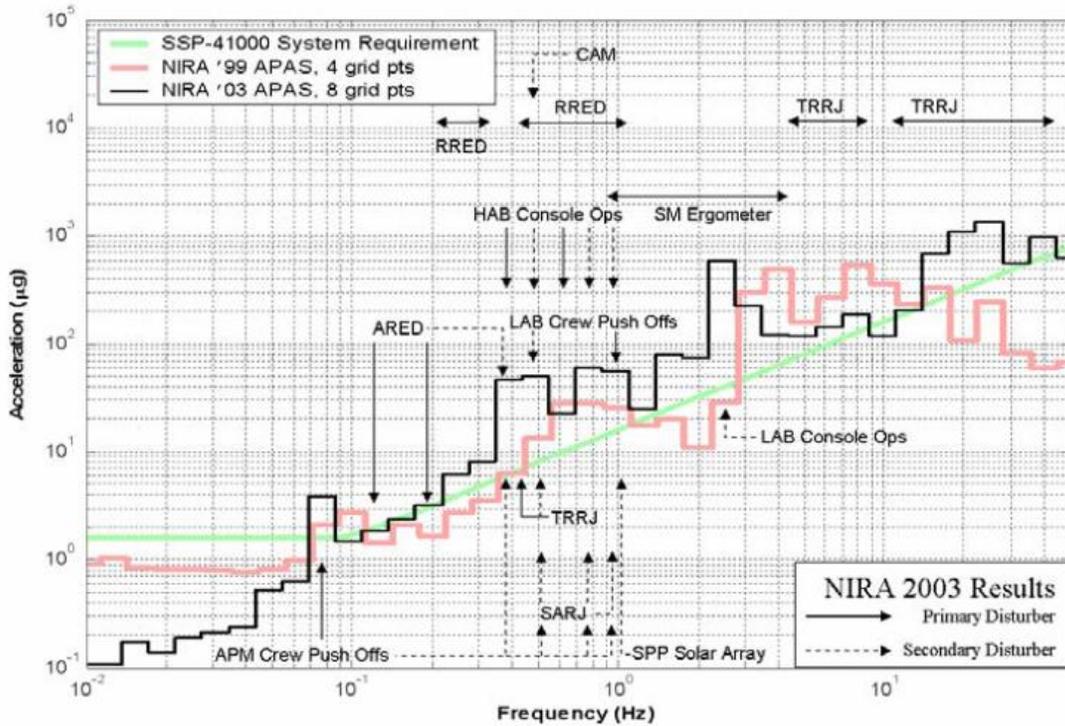


Figure 3.1.7-1. Typical ELC microgravity environment.

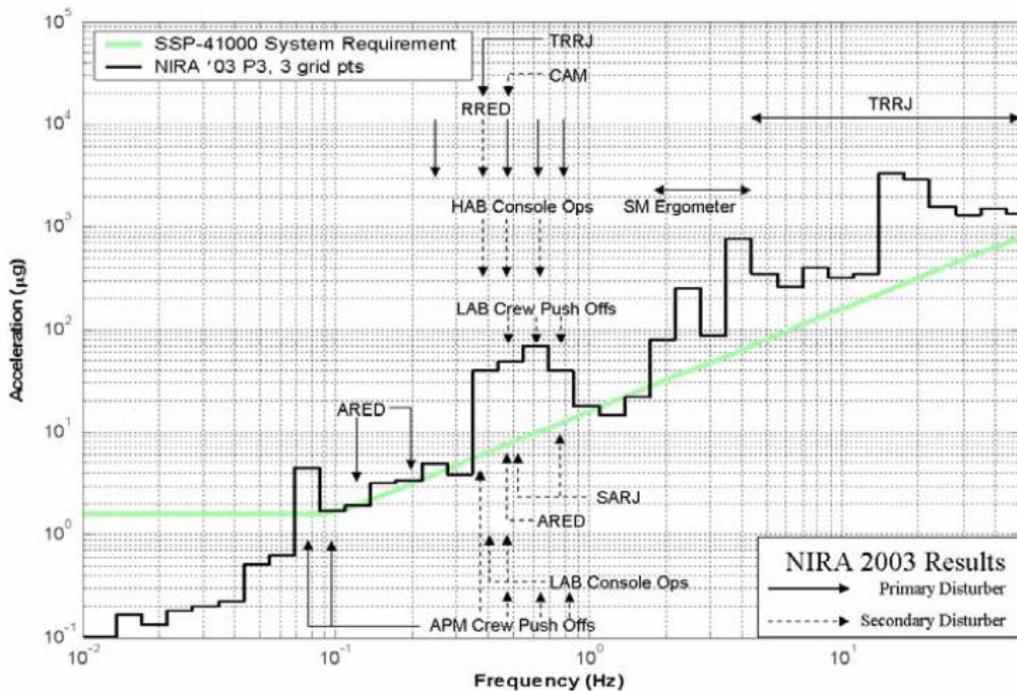


Figure 3.1.7-2. Typical ELC microgravity environment.

3.1.8 Electromagnetic Compatibility/Electromagnetic Interference (EMC/EMI)

EMI/EMC requirements are contained in Space Station Electromagnetic Emission and Susceptibility Requirements, SSP 30237,⁷ electrical grounding (Space Station Grounding Requirements, SSP 30240)⁸, cable and wired design (Space Station Cable/Wire Design and Control Requirements for Electromagnetic Compatibility, SSP 30242),⁹ corona and electrostatic discharge (ESD) immunity (Space Station Requirements for Electromagnetic Compatibility, SSP 30243),¹⁰ and electrical bonding (Space Station Electrical Bonding Requirements, SSP-30245)¹¹. The PD should, at a minimum, review the following paragraphs:

SSP 30237:

CONDUCTED EMISSIONS: Paragraph 3.2.1 and all subordinate paragraphs
CONDUCTED SUSCEPTIBILITY: Paragraph 3.2.2 and all subordinate paragraphs
RADIATED EMISSIONS: Paragraph 3.2.3 and all subordinate paragraphs
RADIATED SUSCEPTIBILITY: Paragraph 3.2.4 and all subordinate paragraphs

SSP 30240:

GROUNDING: Paragraph 3.2 and all subordinate paragraphs
CABLE/WIRE DESIGN for EMC: Paragraph 3.1
ESD: Paragraph 3.2.9
CORONA: Paragraph 3.2.13
ELECTRICAL BONDING: Paragraph 3.1

Alternately, Attached Payloads may choose to accept a minimal increase of EMI risk with a somewhat less stringent electrical field radiated susceptibility (RS03) requirement on equipment considered non-safety critical to the vehicle and crew as described in Reference 3, Paragraph 3.2.2.4.4.

Attached Payload circuits that exhibit a failure or anomalous behavior due to EMI and can cause or propagate a hazard due to such behavior is deemed safety-critical and shall meet the margins defined in SSP 30243, paragraph 3.2.3.¹⁰

Questions concerning requirements applicability or purpose can be addressed to the ISS Electromagnetic Effects Panel (EMEP) co-chairs.

For more information concerning designing to meet EMC requirements, see NASA Reference Publication 1368, Marshall Space Flight Center Electromagnetic Compatibility Design and Interference Control (MEDIC) Handbook, available on the NASA Technical Reports Server at <http://ntrs.nasa.gov>.¹²

Additionally, ISS Management Policy Directive (MD-113) applies to all U.S.-funded intentional transmitting and receiving systems for operation on the ISS. This policy provides the authority and requirement to conduct technical reviews in ensuring RF compatibility of the proposed RF system/usage with existing ISS RF operation.

Regarding International Partner (IP)-managed RF payloads for the ISS, the process for coordination of frequency selection and compatibility assurance will be worked through the JSC Spectrum Manager (designated as the ISS Spectrum Manager) as documented in SSP 50423.¹³ Additionally, regulatory filing and approval for space operation of an IP-managed RF payload is the responsibility of the IP, consistent with the applicable domestic regulation and process of the IP. The International filing documentation, at the minimum of Advanced Publication Information

(API) stage, is to be provided to the ISS Spectrum Manager for final validation of ISS onboard compatibility certification. Upon this validation, the payload will be issued an ISS RF Authorization document for Program ICD and IRD requirements closure.

3.2 Japanese Experiment Module-External Facility (JEM-EF)

While there are numerous similarities between the ELC and Columbus-EPF capabilities, the JEM-EF stands apart. The JEM-EF offers active cooling, for example, and its physical accommodations and layout are strikingly different from the other two ISS external payload sites. The JEM-EF, in its entirety, is 6 m x 5 m x 4 m (20 ft x 16.7 ft x 13.3 ft) and weighs approximately 4000 kg (8890 lb), and is represented in Figures 3.2.0-1 and 3.2.0-2. The JEM-EF can accommodate 11 different payloads, five of which are NASA allocated. The JEM-EF consists of the Pressurized Module, the Exposed Facility (EF), the Experiment Logistics Module (ELM), and the JEM Remote Manipulator System (RMS). This Section of the Guide provides an overview suitable for ISS proposers.

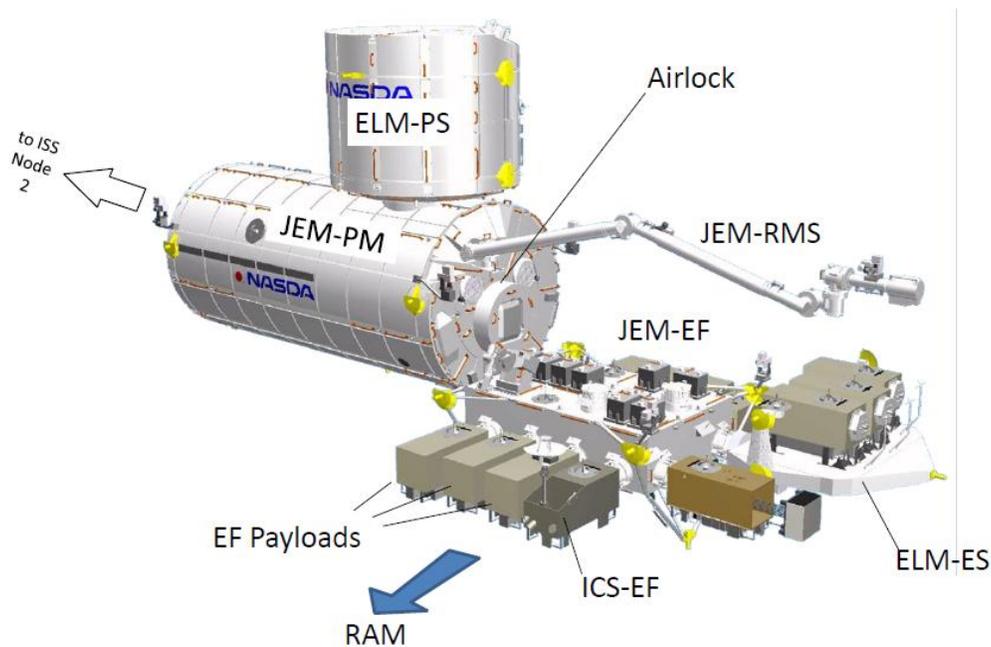


Figure 3.2.0-1. Illustrated JEM-EF overview.

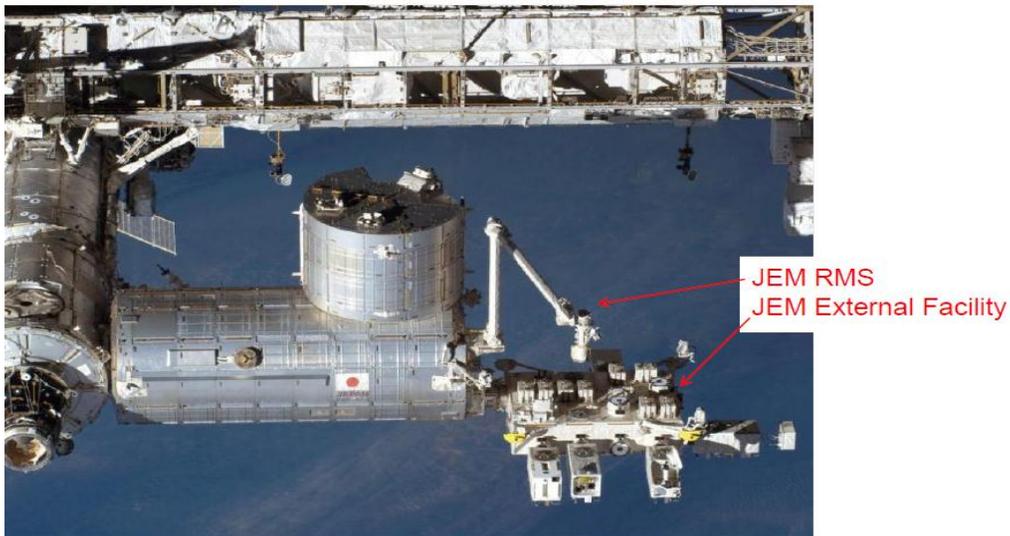


Figure 3.2.0-2. Image of JEM-EF from space.

3.2.1 Physical Accommodations

For interfacing with the JEM-EF, the locations are referred to as a standard site and a large site. The standard site has a 550-kg (1,150-lb) mass limit, and the large site has a 2,250-kg (5,500-lb) mass limit. However, these mass limits must include the Payload Interface Unit (PIU), EVA supporting hardware, Grapple Fixture (GF) supporting robotics interfaces, and other supporting hardware such as multi-layer insulation (MLI), as required. It is expected that almost all proposers will utilize the standard site, primarily based on the availability of the standard sites versus the large site.

The JEM-EF provides 11 locations for experimental payloads, which are summarized below in Figure 3.2.1-1. However, if you need detailed design-to references, please consult NASDA-ESPC-2900A.¹⁴

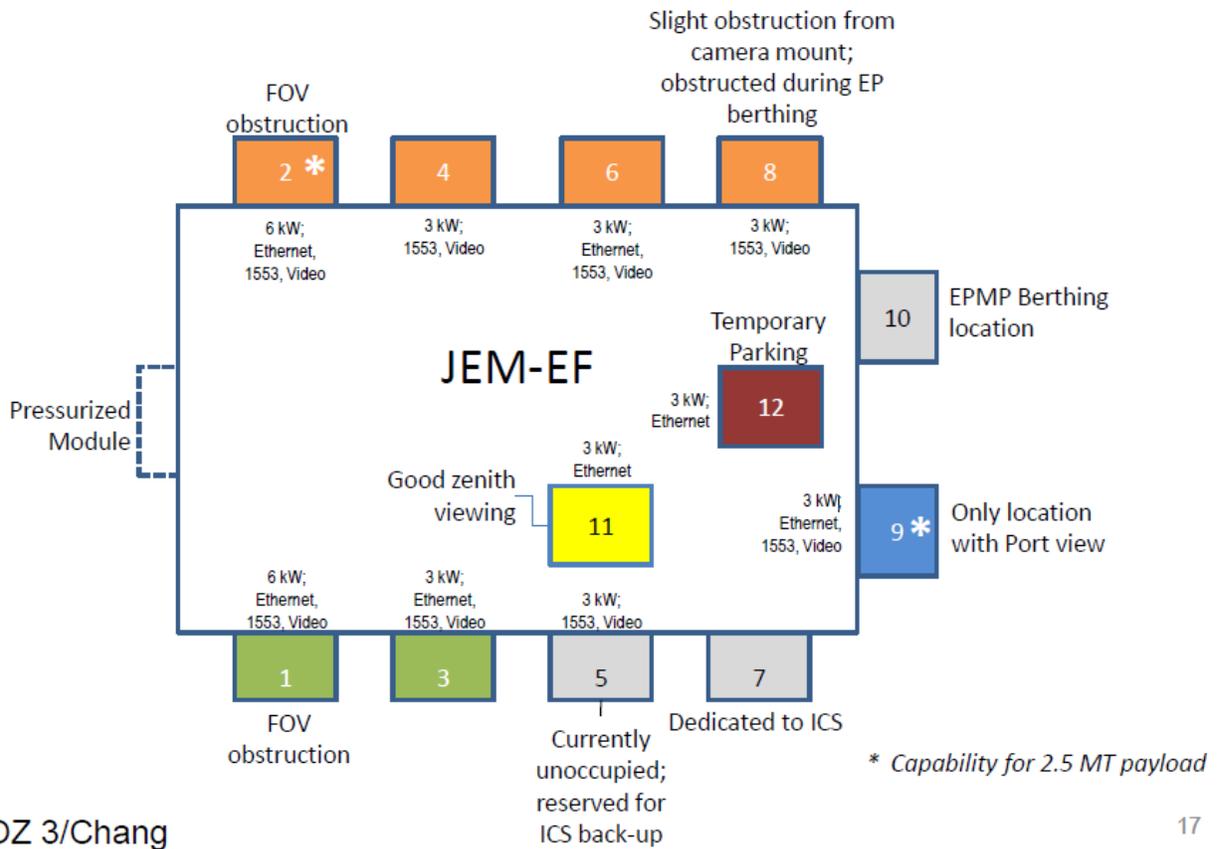


Figure 3.2.1-1. Illustrated JEM-EF 10 locations for experimental payloads.

A more detailed summary for each JEM-EF site is shown in Table 3.2.1-1.

Table 3.2.1-1. Detailed summary for each JEM-EF site.

Location	Viewing	Payload Size	Description / Notes	Power	Data
1	Ram, Nadir, Zenith	500 kg	Ram field of View (FOV) obstruction by JEM module	6 kW	Ethernet, 1553, Video
3	Ram, Nadir, Zenith	500 kg	Clear view	3 kW	Ethernet, 1553, Video
5	Ram, Nadir, Zenith	500 kg	ICS System back-up site (negotiable?)	3 kW	1553, Video
7	Ram, Nadir, Zenith	500 kg	ICS-dedicated	-	-
9	Port, Zenith, Nadir	2.5 MT	Best volumetrically for large payloads (up to 2.5 MT), but not necessarily the best viewing	3 kW	Ethernet, 1553, Video
2	Wake, Nadir, Zenith	2.5 MT	Can hold large payloads, but has an FOV obstruction by JEM module	6 kW	Ethernet, 1553, Video
4	Wake, Nadir, Zenith	500 kg	Clear view	3 kW	1553, Video
6	Wake, Nadir, Zenith	500 kg	Clear view	3 kW	Ethernet, 1553, Video
8	Wake, Nadir, Zenith	500 kg	Obstruction during EP berthing, slight obstruction from camera mount	3 kW	1553, Video
10	Wake, Nadir, Zenith	500 kg	EPMP berthing site	-	-
11	Zenith only	500 kg	Good Zenith viewing	3 kW	Ethernet
12	Zenith only	500 kg	Temporary stowage location	3 kW	Ethernet

Payload Interface Unit (PIU)

The Equipment Exchange Unit (EEU) interfaces payloads to the JEM-EF site. All payload accommodations, including power, data, and coolant, are provided through the EEU connectors. The EEU consists of two units: the Exposed Facility Unit (EFU), located on the JEM-EF structure, and the PIU, which is attached to the payload. Figures 3.2.1-2 and 3.2.1-3 provide an illustration of these units. In addition, NASDA-ESPC-3122¹⁵ contains more detailed design-to-type information for the PD that may be used as additional references.

All JEM-EF payloads must utilize the PIU and the supporting structure for interfacing with the JEM-EF, as this unit serves as your primary interface to the ISS, much as the ExPA serves as the interface for utilizing the Truss/ELC and Columbus external payloads. The structure will be provided by the PD.

The PD should select one of the following PIU types, based on the proposed usage:

1. PIU-Type A: Equipped with fluid interface
2. PIU-Type B: NOT equipped with fluid interface

NASA payloads utilizing the JEM-EF will most likely require active cooling, as this is a major difference between JEM-EF and the other ISS external payload sites. Therefore, most proposers will use the Type A PIU, in all probability.

The PIU is the passive half of the EEU, and is installed on your payload. The PIU is affixed to the structural side of the active part of the EEU on-orbit. The PIU is the critical interface for all accommodations from the JEM-EF, and has the following functions:

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1. Kinetic energy absorption – the PIU damper cushions the “collision” between the payload and the JEM-EF EEU, during robotic installation.
2. On-orbit capturing and alignment – using the JEM arm, the PIU is captured and aligned by three latching arms. The PIU absorbs any positioning error between the PIU and the EFU, and keeps the necessary position and accuracy for on-orbit mating with the active-side of the EEU.
3. Mate/de-mate functions of JEM-EF resources – the PIU simultaneously provides the mating and de-mating capability for physically interfacing, using robotics, the payload with JEM-EF.
4. PIU protection – to protect the quick-disconnect connectors from the exposed space environment, the PIU provides a protective cover which automatically opens during the robotic mating sequence.
5. JEM-EF resources/accommodations interface – the PIU provides payloads with power, data, and coolant resource and accommodations interfaces.

The proposer should be cognizant of the following important interfaces between the PIU and their payload. If a proposer determines the need for additional interface or design-to details, the following provide appropriate interface references from NASDA-ESPC-3122.¹⁵

- (a) Mechanical – Figure 3.2.1-2 to Figure 3.2.1-5 provide an overview of the PIU and payload structural interface envelope. For detailed assembly drawings, refer to Appendix 3 of Reference 15.
- (b) Electrical – Refer to Appendix 3, Table 3-5, 3-6. The references for heaters are contained in Appendix 3, Table 3-7.¹⁵
- (c) Thermal – In addition to the criteria contained in Section 3.2.3 of the Guide, please note that the allowable interface surface temperature of the PIU is 60 °C on the ground, and –54 to 82 °C on-orbit.
- (d) Fluid – The PIU Type A reference, which many proposers will elect to use as it contains the fluid interface, is contained in Appendix 3 in greater detail.¹⁵
- (e) Robotic – Please refer to Appendix 3, Tables 3-8, 3-9, and 3-10 for additional information for robotic attachment details.¹⁵ Remember, the PIU requires a GF for robotic manipulation, which should be included in your payload mass and envelope calculations.

The PD must provide a structural interface “box” that contains the PIU, GF, trunnions, umbilical connect mechanism, and H fitting and is referred to as the JEM-EF Attach Mechanism. The latch mechanisms engage the trunnions for securing the payload’s structural interface to the ISS visiting vehicle for launch. The maximum payload envelope that the PD-developed structural interface hardware must fit within, is provided in Figures 3.2.1-2 and 3.2.1-3. The trunnions represented in Figure 3.2.1-2 are used to attach the payload, and accompanying hardware, to the latch mechanisms and survival heater power connector on the visiting vehicle. It should be noted that the trunnions and latch mechanisms may change with flights on Falcon 9/Dragon from the exact hardware presented in Figure 3.2.1-2. However, because these types of interface hardware will be required, the PD should provide a capable structure. In addition, the Umbilical Connect Mechanism (UCM) will be different from the representation in Figure 3.2.1-2. This connection will be adapted for flight on Falcon 9/Dragon. The PD should contact the ISS RIO for the latest interface configuration(s).

NOTE: The natural frequency requirements of the EF Experimental Payload is ≥ 5 Hz for a standard payload (<500 kg) or ≥ 2 Hz for heavy payloads (500–1500 kg, positions 2, 9, and 10), when fixed at the PIU mating surface in all directions.

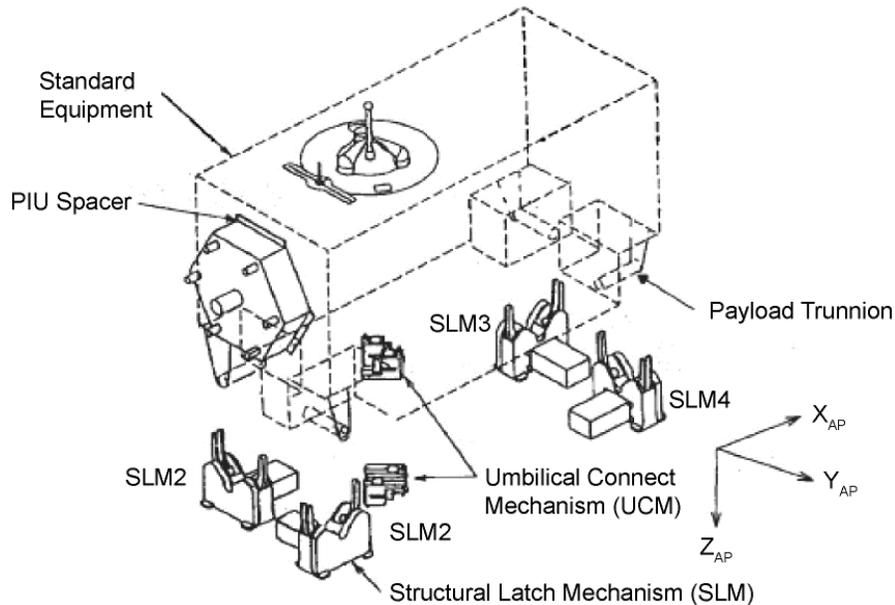


Figure 3.2.1-2. JEM-EF Attach Mechanism.

The PIU, required for all JEM-EF payloads, is supplied by NASA to the PD community, and is represented in Figures 3.2.1-2 to 3.2.1-5. In addition, the following PIU accessories are generally included with PIU usage:

- PIU structural template assembly
- PIU attachment bolts, including spares
- PIU transportation and storage containers
- PIU adapter plate
- Passive flight support equipment (may be supplied by Space X upon request)

The standard payload envelope and location of the required GF are shown in Figure 3.2.1-4.

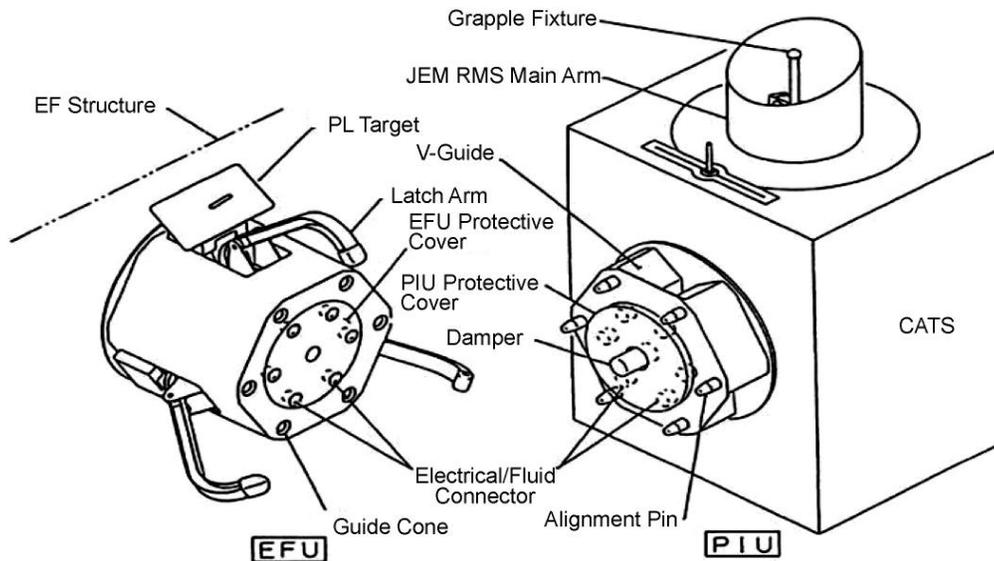
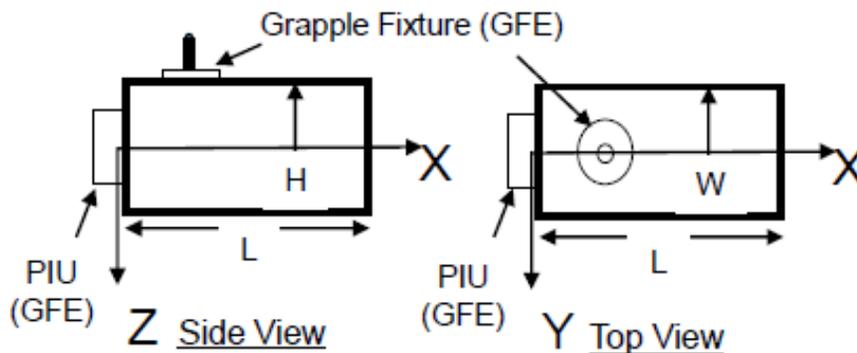


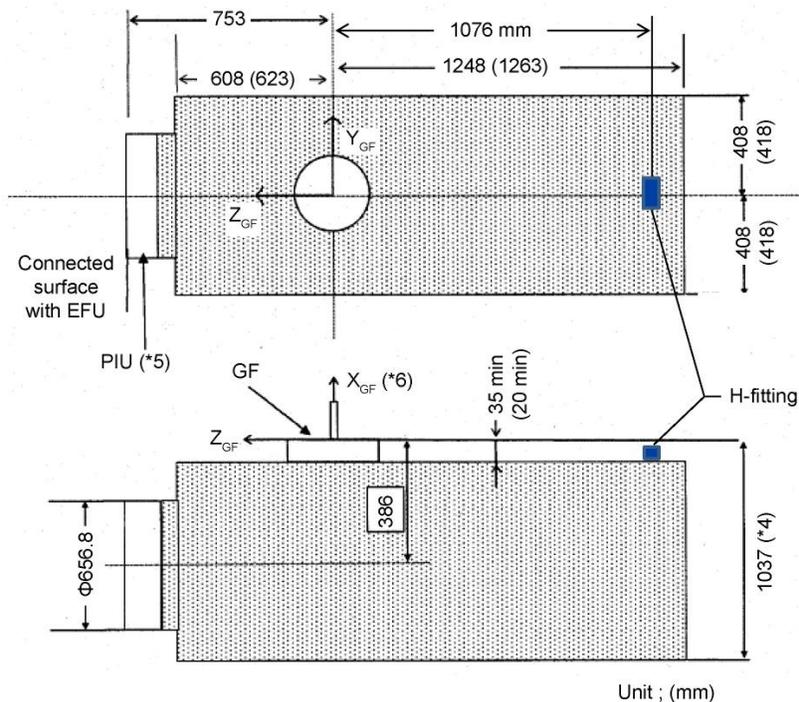
Figure 3.2.1-3. Payload Interface Unit.



Axis	mm	ft	inch
W	800	2	7.50
H	1000	3	3.37
L	1850	6	0.83

Figure 3.2.1-4. Standard payload envelope and location of the required GF.

Figure 3.2.1-5 illustrates the PIU standard envelope, with a standard GF attached.



- (*1) The envelope (hatched area) is defined with respect to the origin of Grapple Fixture coordinate.
 (*2) All are the maximum values (including manufacturing tolerances and thermal deformation) except for standard dimensions and the dimensions specified in "min."
 (*3) The values in brackets are dimensions including MLI.
 (*4) Includes attachment mechanism to a carrier for launch and recovery.
 (*5) The dimension of PIU shall be referred to using Reference 15.
 (*6) The details of GF can be found in NASDA-ESPC-2901.¹⁶

Figure 3.2.1-5. PIU standard envelope and GF.

3.2.2 Robotics

The JEM-EF is serviced by a combination of the ISS robotics system and the Japanese supplied KiBO arm, which is used for JEM-EF only. The JEM's Remote Manipulator System (JEM-RMS) is a robotic manipulator system used to support experiments and logistics conducted on JEM-EF. It is composed of two arms, the Main Arm (MA) (Figure 3.2.2-1) and the Small Fine Arm (SFA) (Figure 3.2.2-2). The MA is used to handle (move, position, berth, de-berth, or maintain position) a large payload and the SFA is used for small items and dexterous tasks.

Payloads that are brought up on the SpaceX Dragon are removed from Dragon and translated to the JEM using the ISS Dextre. (Details of the Dextre robot are provided in Section 3.1.2 of this Guide.) The payload is then grappled by the JEM MA at the FRGF. It is then released by the Dextre OTCM, transferred to the KiBO and then installed by the JEM MA. Upon installation onto JEM, the payload will have survival heater power from JEM, which is a different circuit at a different voltage and allowable power level from Dragon. Analysis must show payload survival without power for at least 7 hours.

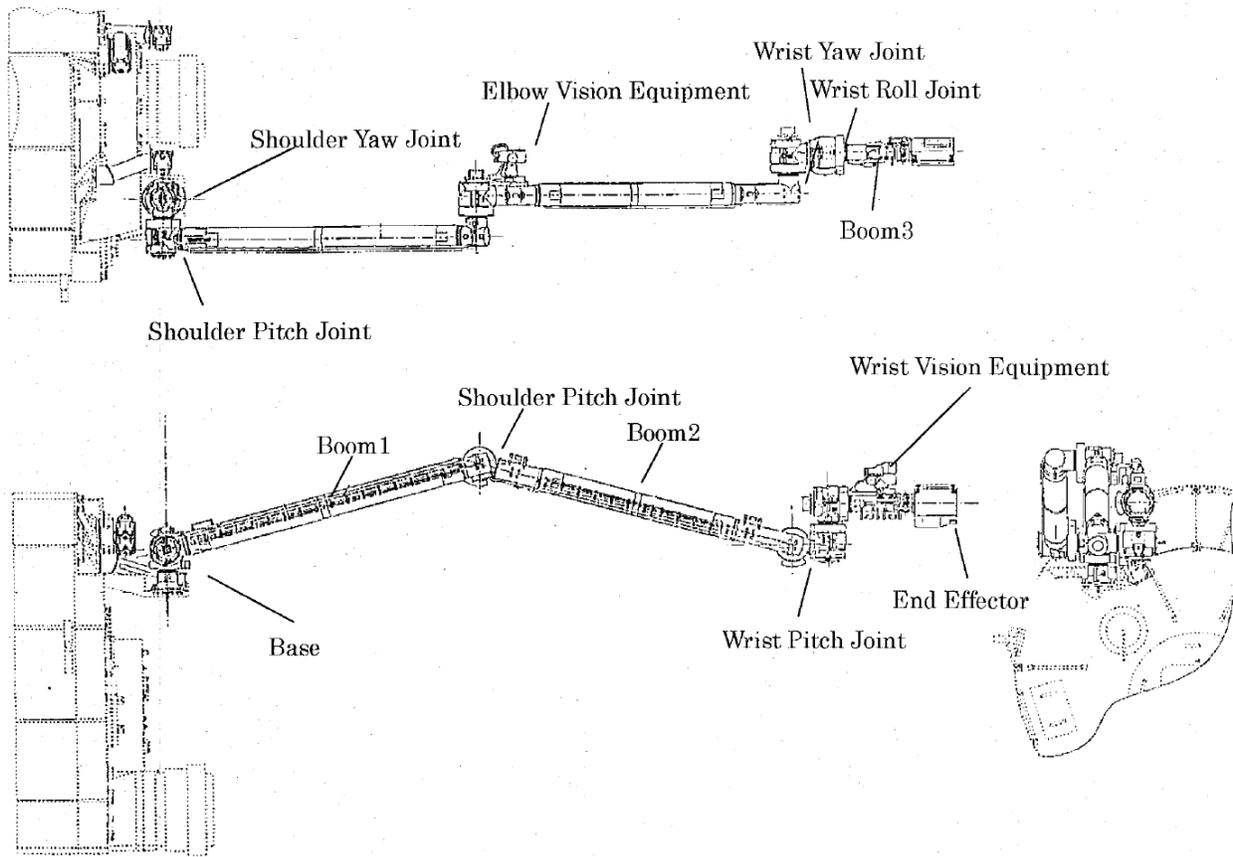


Figure 3.2.2-1. JEM-EF MA configuration.

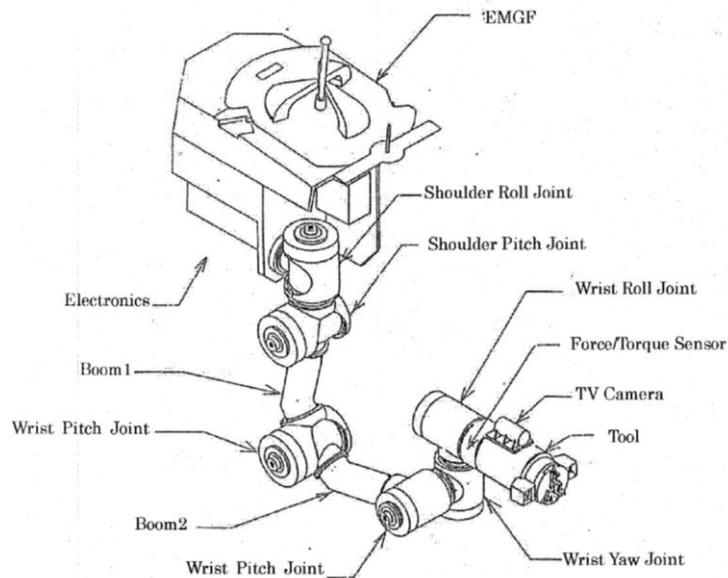


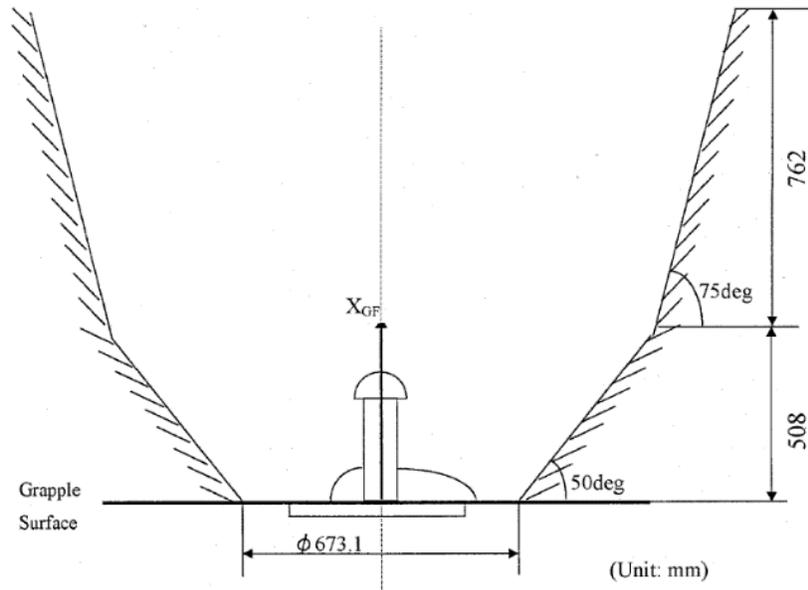
Figure 3.2.2-2. SFA configuration.

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<http://espd.gsfc.nasa.gov/isseppg/>
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All JEM-EF payloads must have a GF that matches the end effector in order for the JEM-RMS to handle the payload. The GF will be supplied by the ISS, but is mounted on a payload-supplied structural interface.

Payloads are grappled by the MA end effector which is the grapple mechanism at the end of the MA. The MA end effector can provide interfaces such as electric power supply, data, and video signals, but this requires the use of a special type of GF which is not readily available.

Clearance for the MA and payload grappling operation shall conform to the illustration in Figure 3.2.2-3.



NOTE: If there is an object higher than 1270 mm, evaluation by element integrator is needed.

Figure 3.2.2-3. Illustration of clearance for the MA and payload grappling operation.

In all cases, the payload interface design is defined and controlled by NASDA-ESPC-2901.¹⁶ The PIU dynamic envelope is shown in Figure 3.2.2-4.

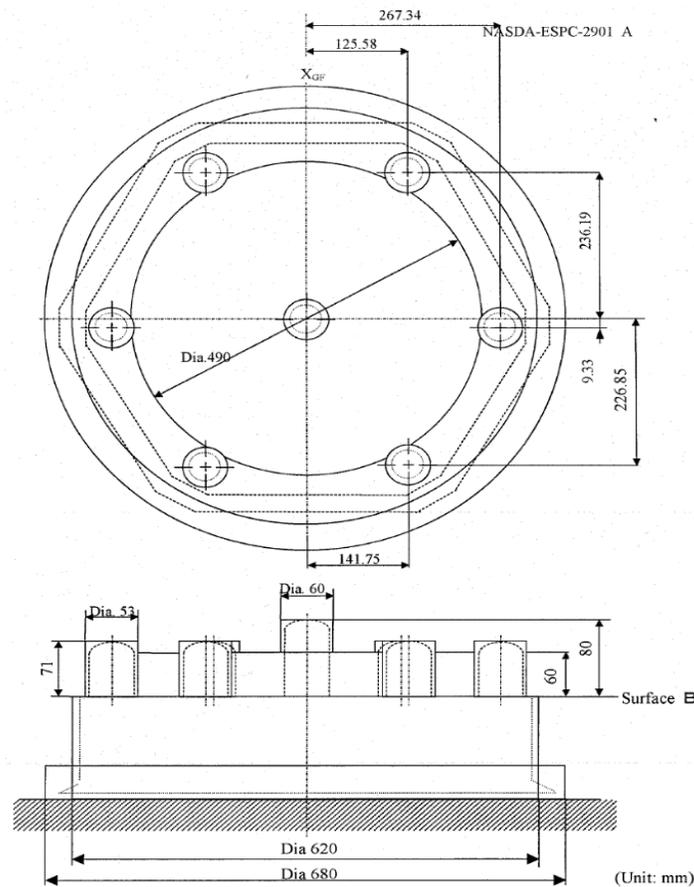


Figure 3.2.2-4. PIU Dynamic Envelope.

3.2.3 Thermal

The thermal interface between the JEM-EF and the experiment payload consists of the Passive Thermal Control System (PTCS) and the Active Thermal Control System (ATCS). There are two thermal system design options available to a payload proposer and each works through a passive half of a PIU mounted on your payload and a corresponding active half attached to the JEM-EF platform. A Type B or Type B-EP PIU have no fluid interface, whereas type A or type A-EP do. The proposer should consider the details involved with either the ATCS/PIU type A or the PTCS/PIU type B, which may require different plumbing hardware configurations within their payloads to meet ISS system requirements. The thermal characteristics related to the PIU are defined in Reference 15. Proposers would need to prepare a math model utilizing the PTCS or the ATCS for a thermal analysis in accordance with JFX-2000073.¹⁷ These analyses need to be submitted to the ISS element integrator for considering interactions with other payloads.

3.2.3.1 Passive Thermal Control System (PTCS) Interface

The PTCS interface consists of heat radiation from the outer surfaces of the EF and of other experiment payloads and heat conduction between the EFU and a type B PIU. There is no fluid coolant in PTCS. The outer surfaces of a proposer's payload that could have heat radiation with the EF and/or with the other EF experiment payloads should be covered with MLI (thermal

blanket). The MLI thermal characteristics the proposer should utilize in their design analysis are defined in Table 3.2.3-1. The interface analysis for heat radiation cannot have heat conductance influence on the EF system.

The thermal conduction design portion for the PTCS is the consideration of heat flow across the EFU to the type B PIU contact interface. The interface surface temperature should fall between -45 to +65 °C, from the moment of EFU/PIU contact, with a EFU/PIU thermal conductance of 10 ±5 [W/K]. In addition, the temperature difference shown by thermal analysis (supported by ISS Element Integrator) between the interface surfaces of EFU and PIU prior to the mating needs to be less than or equal to 80 °C.

Waste heat generated by your proposed payload shall be absorbed and transported by PTCS to minimize the thermal interference between the EF and experiment payload. Heaters are permitted on the payload to preserve the thermal design. The heater power interface provided from the EF is defined in the power interface in Section 3.2.4 of this guide.

Table 3.2.3-1. Thermal-optical characteristics of experiment payload outer surfaces.

Item	Characteristic value
MLI effective emittance	$\epsilon_{\text{eff}} \leq 0.04$
MLI thermo-optical characteristics	$\alpha = 0.31 - 0.6$ $\epsilon = 0.85 - 0.96$

3.2.3.2 Active Thermal Control System (ATCS) Interface

This system involves the use of coolant supplied from the JEM-EF through a piping system through a PIU type A or type A-EP to your payload. The ATCS supplies and collects coolant that absorbs and carries the waste heat away from the proposed payload through the fluid connector of the EEU. A payload ATCS fluid schematic is seen in Table 3.2.3-3. Coolant is not supplied to a payload when it is not operating, so the coolant is not to be considered as the heat source for preserving your proposed payload. Coolant information is available in physical properties of Fluorinert (FC-72-perfluorocarbon) and Space Station Program Fluid Procurement and Use Control Specification, SSP 30573.¹⁸

Coolant Supply Characteristics Summary

Table 3.2.3-2 outlines parameters of the ATCS coolant fluid being supplied to your payload. The coolant system is the key difference that sets the ATCS apart from the PTCS. The coolant temperatures and flow rates cited are those once the operational conditions of the ATCS fluid loop are stabilized for the JEM-EF payloads (stabilized means after a startup time or configuration change). These will depend on the ISS JEM-EF platform operation and configuration (the number of payloads, attached location, heat rejection, etc.), and could even deviate somewhat from these ranges. The interface coolant flow rate is based on a unit heat rejection (including the influence of the payload attaching position). The return coolant temperature range cited is considered acceptable for a payload if it meets three criteria outlined in NASDA-ESPC-2900A.¹⁴ These have to do with the required coolant outlet temperature from your payload to the EF, a maximum amount of coolant warm up from the payload, and a

required minimum temperature of the coolant back into the EF. The payload proposer should confirm all of this with the element integrator for his payload.

The maximum design pressures (MDP), excluding water hammering, cited in Table 3.2.3-2 is guaranteed to your payload based on a two-fault failure internal EF design when there are no factors or no failures that raise the payload internal pressure. The maximum pressure increase allowed in your payload is 392 kPa (4.0 kgf/cm²d) when water hammering occurs. Payload proposers should configure their payload fluid system so that the pressure drop in their payload excluding the PIU, ranges from 52.4 to 57.8 kPa when the flow rate is 190 kg/h (Table 3.2.3-2). The system level pressure drop is ultimately controlled by the EF system through flow control and the actual measured pressure drop tolerance during testing will be considered for verification purposes. Lastly, the fluid pressure in a payload during berthing operation to JEM-EF shall be 785 kPa (8.0 kg/cm²A) or less.

You will need to set the amount of flow rate in your payload to correspond to the maximum payload heat rejection required with a pressure drop based on the Tables 3.2.3-2 and 3.2.3-3 and shown in Figure 3.2.3-1. You should not consider the absorption of heat from the coolant, but instead consider it only for its heat rejection. The amount of heat leakage from the coolant interface heat flow from the coolant loop to the payload in the nominal operation and during non-operation[†] after one failure, should not exceed 49 W.

Table 3.2.3-2. ATCS fluid interface coolant supply characteristics.

Interface Parameter	Coolant Temperature					
	Supplied Coolant Temperature to Experiment Payload, °C	Returned Coolant Temperature from Experiment Payload, °C	Max. Allowable Heat Rejection of Experiment Payload, W	Provided Coolant Flow Rate, kg/hr	Coolant Pressure under nominal operation, MPa (kgf/cm ² A)	Coolant Maximum Design Pressure in system, MPa (kgf/cm ² A)
	16 – 24	16 – 50	900	190 + 18% / -0%	No higher than 0.78 (8.0)	1.57 (16.0)

[†] This is the case that the main electrical power is shutdown while coolant loop flow to the payload remains flowing.

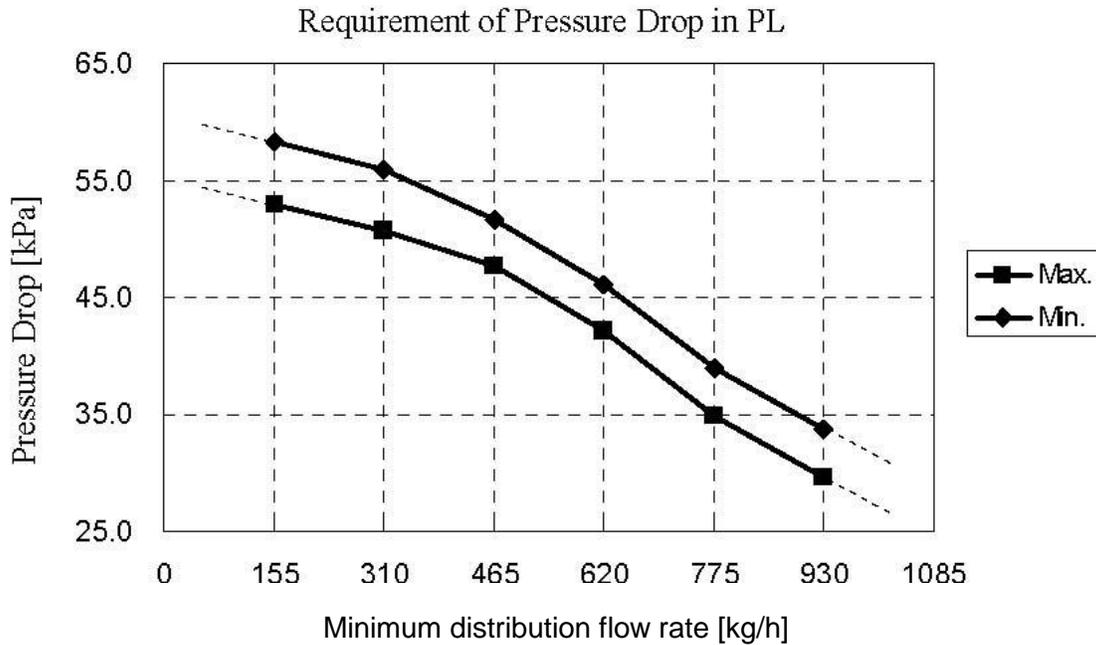
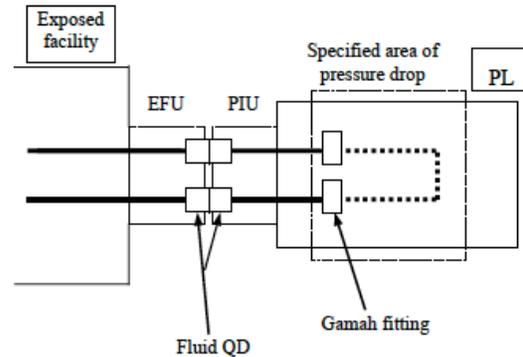


Figure 3.2.3-1. Coolant pressure drop required in experiment payload.

Differential pressure is specified in Table 3.2.3-3.

**Table 3.2.3-3. Differential Pressure and Schematic
 (Note that the pressure drop in the PIU is not included.)**

Heat rejection, kW	Minimum distribution of flow rate, kg/h	Pressure Drop, kPa (kgf/cm2D)	
		Minimum	Maximum
1	155	52.9 (0.540)	58.4 (0.595)
2	310	50.7 (0.517)	55.9 (0.570)
3	465	47.7 (0.486)	51.7 (0.528)
4	620	42.1 (0.429)	46.1 (0.470)
5	775	34.9 (0.356)	39.0 (0.398)
6	930	29.6 (0.301)	33.8 (0.344)



Experiment Payload ATCS Design Requirements

Payloads being proposed with fluid systems that will interface with the ATCS fluid loop of the EF will need to address coolant pressures, cleanliness, capacity, and temperature. Cleanliness requirements are contained in Table 3.2.3-4. The capacity of the fluid system with the coolant inside of a proposer’s payload is 2.0 L or less including the PIU. This capacity excludes a second loop in the payload without a direct interface to EF-ATCS (fluid loop). A filter with the mesh size of 40 μm or smaller is recommended to be installed at the outlet of your payload ATCS loop. Coolant external leakage from the overall plumbing inside your payload is specified in Table 3.2.3-5, while that of the PIU itself is described in NASDA-ESPC-3122.¹⁵

Table 3.2.3-4. Cleanliness requirements for experiment payload fluid systems.

Particle size, μm	Maximum allowable number per 100 ml test fluid *1
0 – 5	Unlimited
6 – 10	3600
11 – 25	1050
26 – 50	210
51 – 100	20*2
101 – 250	2
251 or over	Not allowed

*1. Fiber is included. Note that “fiber” means a non-metallic particle whose length is 100 μm or longer and whose length/ diameter ratio is 10/1 or less.

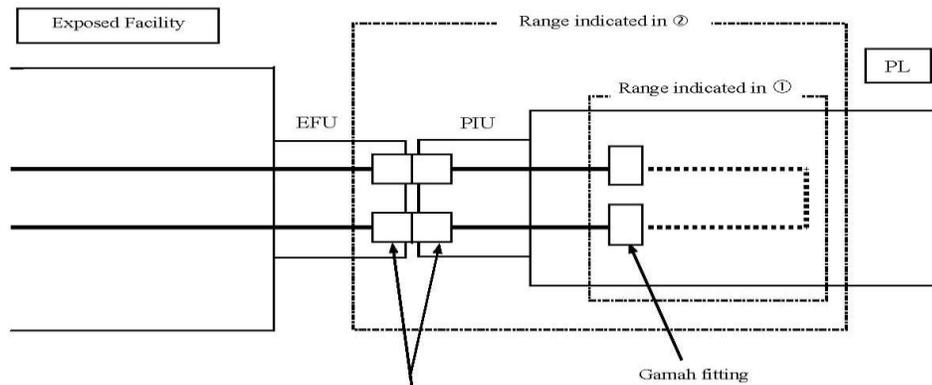
*2. Note that no metallic particle whose length is 51 μm or longer is allowed.

Note 1. The volume of test (sampling) fluid is 100 ml for the unit area 0.093 m^2 (1 ft^2) of required inner surface area. Note that the volume of test fluid is 100 ml when the required inner surface area is less than 0.093 m^2 (1 ft^2).

Note 2. The cleanliness of test fluid (maximum allowable number of particles) shall be 1/10 of the objective level or less.

Table 3.2.3-5. Allowable leak for experiment payload.

	Allowable leak	Remarks
① Fluid QD not included	$1.01 \times 10^{-6} [\text{Pa}\cdot\text{m}^3/\text{s}]$ @ 1.18[MPa] (1×10^{-5} [sccsGHe : @ 12kgf/cm ² A])	Including the leak in the Gamah fitting
② Fluid QD included (Reference data)	$3.04 \times 10^{-6} [\text{Pa}\cdot\text{m}^3/\text{s}]$ @ 1.18[MPa] (3×10^{-5} [sccsGHe : @ 12kgf/cm ² A])	-45 to +16 [°C]
	$7.09 \times 10^{-6} [\text{Pa}\cdot\text{m}^3/\text{s}]$ @ 1.18[MPa] (7×10^{-5} [sccsGHe : @ 12kgf/cm ² A])	+16 to +65 [°C]
	$2.13 \times 10^{-5} [\text{Pa}\cdot\text{m}^3/\text{s}]$ @ 1.18[MPa] (2.1×10^{-4} [sccsGHe : @ 12kgf/cm ² A])	+65 to +79 [°C]



Absorption equipment for pressure adjustment such as an accumulator in a proposed payload shall not be operated (not absorb coolant from the EF system) when the pressure is 471 kPa (4.8 kgf/cm²A) or less, at any thermal environment that the payload is exposed to. A payload proposed requires the ability to detect abnormally high-temperature of inflow/outflow coolant to be in a safe operating condition and limit heat rejection by itself. Abnormally high-temperature criteria are designated as "normal operating temperature +10 °C plus measurement errors" in the payload and will be controlled by commands from the ground.

The coolant supply from JEM-EF ATCS could stop for ISS system operating reasons. The payload proposer should configure his design to be capable of stopping safely to prevent hazards such as the destruction of the EF or other neighboring payloads due to the temperature increase. The proposer should design to prevent the destruction of his fluid system due to the increased pressure of the coolant internal to the plumbing, caused by temperature increases (i.e., equipping the accumulators of the system to absorb pressure increase). Proposers' payload will need to be two-fault tolerant for any MDP (16.0 kgf/cm²A) exceedance.

Visiting Vehicle Transfer

One challenge of experiment payload thermal control design is the safe survival of your payload during the 7-hour unpowered transfer of the payload from the visiting vehicle to the JEM-EF for berthing (as mentioned previously, the fluid pressure in your payload during berthing operation to JEM-EF needs to be 785 kPa (8.0 kg/cm²A) or less). It is important that your payload be thermally conditioned with the survival heater power supplied by the visiting vehicle before it is released. The payload will be powered once it is berthed to JEM-EF. The power from Dragon Visiting Vehicle is supplied at 120 Vdc and is limited to 200 W shared between all the external payloads in the launch vehicle trunk. The launch vehicle, its cargo bay, and the payload itself can be in a combination of orientations during the payload transfer. The thermal interface requirements are found in Reference 14.

Thermal Models

Key thermal considerations that must be addressed by the external PD are the development and delivery of a thermal model in required format and generation of thermal analyses to demonstrate that the payload will remain safe during LTA, as well as applicable on-orbit stowage sites on the ISS. These models will be provided to support required thermal assessment by the Launch Vehicle integrator and the ISS vehicle integrator.

The external PD will be required to develop and deliver a thermal model compatible with Thermal Desktop[®] and TRASYS/SINDA. The model is expected to be comprised of no more than 500 geometric and 500 mathematical nodes that represent all geometric and material properties required for determining critical hardware temperatures. These models will be utilized by the PD to assess the external payload in its on-orbit configuration, including the time periods while stowed in the Launch Vehicle and on the ISS JEM-EF. These models will be provided to the ISS Program in order to support required thermal assessments by the Launch Vehicle integrator and the ISS vehicle integrator.

Thermal Analysis

The external attached payload must remain safe during the LTA phase while stowed in the Launch Vehicle, as well as, during the subsequent phase while it is stowed on an ISS ELC when it is exposed to a space sink temperature of 3 K, the thermal environment parameters

defined in Table 3.2.3-6, the ISS flight attitudes and solar beta angle ranges defined in Table 3.2.3-7 (similar attitude requirements will be defined for the applicable Launch Vehicle) and the thermal interactions with all other on-orbit hardware. Verification of safe operation will require PD generated thermal analysis using ISS Program provided Launch Vehicle and ISS thermal models integrated with a PD developed thermal model of the proposed payload.

Table 3.2.3-6. Thermal environment parameters.

Case	Solar Constant (W/m ²)	Earth Albedo	Earth OLR (W/m ²)	Altitude (km)
Cold	1321	0.2	206	500
Hot	1423	0.4	286	278

Table 3.2.3-7. ISS Flight Attitudes

ISS Attitude Name	Attitude Reference Frame	Solar Beta Range (β)	Yaw	Pitch	Roll	Time in Attitude
+XVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	-15° to $+15^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
-XVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	$+165^\circ$ to $+195^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
+YVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +10^\circ$	-110° to -80°	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
-YVV +Z Nadir	LVLH	$-10^\circ \leq \beta \leq +75^\circ$	$+75^\circ$ to $+105^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
+ZVV -X Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	-15° to $+15^\circ$	$+75^\circ$ to $+105^\circ$	-15° to $+15^\circ$	3 Hours
-ZVV -X Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	$+165^\circ$ to $+195^\circ$	$+75^\circ$ to $+105^\circ$	-15° to $+15^\circ$	3 Hours

3.2.4 Power

Operational Power

JEM-EF electrical power is supplied at a maximum of 3 kW for main (primary) power and maximum 1.2 kW for auxiliary (survival) power as shown in Figure 3.2.4-1. The normal voltage range for main power is 112.5 to 126 Vdc and for auxiliary power is 110.5 to 126 Vdc.

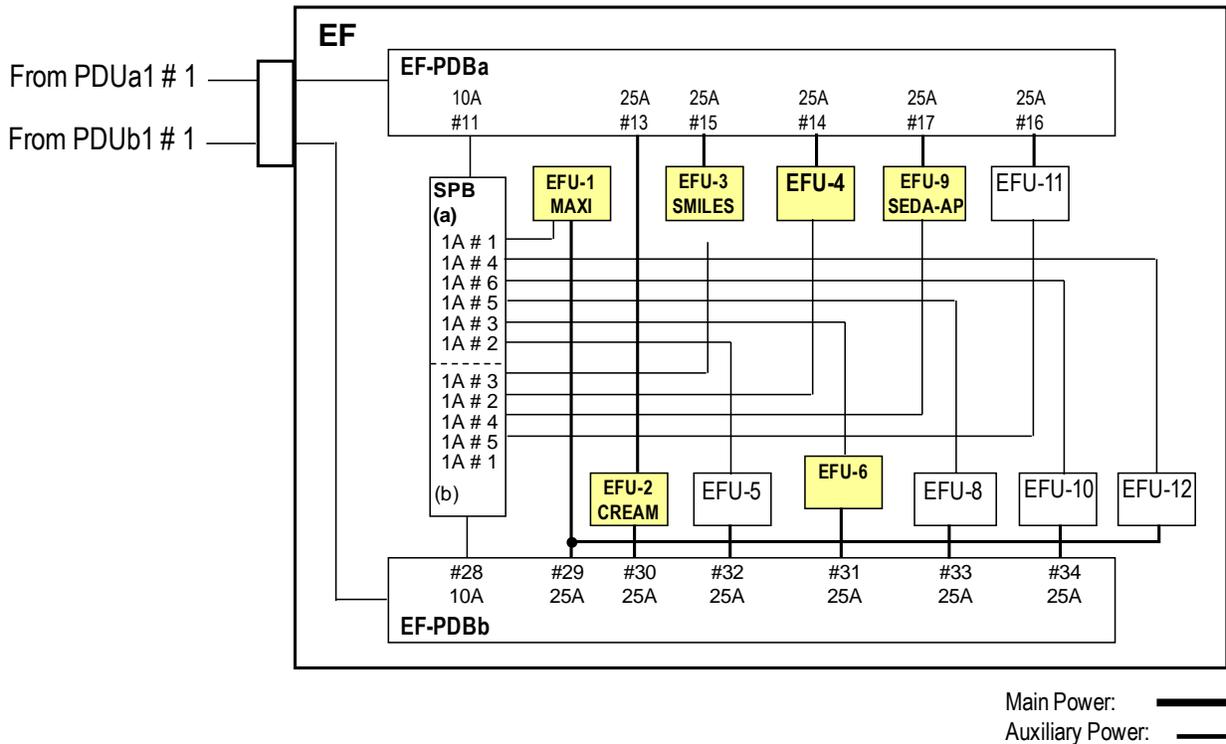


Figure 3.2.4-1. Electric Power Configuration in the JPM.

Survival Heater

Survival heater may be supplied, also, with a channel separated from the experiment channel, and can provide a maximum 120W at 120 Vdc.

Table 3.2.4.-1 also illustrates the power and data connections between the JEM-EF facility and the PIU/payload.

3.2.5 Command and Data Handling (C&DH)

Each payload has the following signal interfaces for the supported C&DH functions (see Figure 3.2.5-1).

Low Rate Data Capability

This is an extension of the payload MIL-STD-1553B¹⁹ data bus primarily used for transmitting the commands to the payload and collecting the low-rate experiment data from the payload. The stub length must be no greater than 2 m, which includes the wiring inside the PIU and the payload equipment. This interface is not available at EFU #7, #10, #11, and #12

Medium-rate Data (Wired) Capability

This interface is part of the MRDL 10/100 Base T network which accommodates both U.S. payloads and JAXA payloads. The JAXA-supplied Layer 2 Ethernet Switch and Multiplexer (LEHX) provide a gateway for science data downlink. The Ethernet frame content conforms

with the LEHX HRDL gateway Layer 2 format per SSP 52050 Rev. K, Section 3.3.6.⁶ This interface is only available for EFU #1, #2, #3, #6, #9, #11, and #12.

Medium Rate Data Link (Wireless)

A MRDL path using a wireless Ethernet data link is available for JEM-EF payloads. It consists of a two-way high data rate communications link using radio frequencies per IEEE 802.11n. This EWC system provides two-way data transfer between the payload sites and the PEHG in the USL. Data rates available to the payload are dependent upon the specific JEM-EF location.

(Although NASA has completed a preliminary coverage map for the EWC System, this information is not included as assumptions made in generating the coverage map may not apply to the JEM-EF external payload utilizing the EWC. External payloads that will utilize the EWC should coordinate with NASA to obtain coverage information. The EWC user should refer to International Space Station (ISS) ELC Wireless Comm (EWC) User Guide, D684-14957-01, for additional information).

High-Rate Data Capability

This one-way high rate data transmission path via optical fiber operates at an encoded signaling rate of 100 Mbps. It uses the TAXI protocol and varying data rates are achieved by parsing the data with sync symbols. There is one channel provided to each payload with the exception of EFU #7, #10, #11, and #12, where there is no such service provided. This has seen limited use, including the NASA Cloud-Aerosol Transport System (CATS) payload.

Video System

A video interface is provided to each payload except for EFU #7, #10, #11, and #12. Due to ISS resource limitation, there are only two channels active for video transmission. As such, one channel will be shared by EFU #1, #4, #5, #8, while the other channel by EFU #2, #3, #6, #9. The video signal interface uses EIA-RS-170A, the National Television Systems Committee color television standard.

Payload Housekeeping Data

There are two analog telemetry monitor channels for each payload interface. Each channel is implemented in primary and redundant signals pair for criticality. They are specifically used for monitoring the temperature and pressure of the payload respectively. The temperature and pressure are critical status monitors when the EF is in preservation mode of operation and/or during the EFU/PIU berthing activities.

Wireless – (ISS Payloads Office will provide definition)

A wireless data link is available for JEM-EF payloads. It consists of a two-way high data rate communications link using radio frequencies per IEEE 802.11n. This EWC system provides two-way data transfer between some JEM-EF payload sites and the PEHG in the USL. Data rates available to the payload are dependent upon the specific attach point and will be provided at a later date. Average payload user data rates of 3Mbps or less typically are easily supportable; higher data rates are possible but require coordination with other payloads.

The NASA-provided EWC Payload Hardware GFE consists of the following items:

1. Payload Antenna Coax Cable Assembly, P/N 684-015637-0001 – Connects EWC Network Interface Card to payload antenna.
2. EWC Payload Antenna, P/N 684-015636-001 – A linearly polarized 17dBi 16° beamwidth antenna, 9 × 9 × 3 inch, must be pointed toward EWC system on USL.
3. EWC Network Interface Card, P/N 684-15644-001 – USB form factor WiFi client, 2x2 MIMO, collects payload data for transmission by the payload antenna.
4. MOXA AWK-4131-M12-CT-T Wireless Access Point – WAP form factor WiFi client, 2x2 MIMO, collects payload data for transmission by the payload antenna.

The payload developer must identify if they are using the NASA provided EWC GFE or if they are developing their own antenna to connect to the external wireless Ethernet. If the payload developer is using the EWC GFE they must define if they choose to interface the wireless Ethernet through either the NIC or the MOXA AWK-4131-M12-CT-T Wireless Access Point (items 3 and 4 listed above).

Finally, it should be noted that all RF operation around the ISS requires advance approval of the transmitting equipment through the ISS RF authorization process. The payload developer will submit the RF system characteristics and planned operating location information via the JSC Radio Frequency Spectrum Management Homepage (<http://ea.jsc.nasa.gov/webapp/fmdb/login.asp>). The initiation of this review process is recommended after SRR and well ahead of PDR. The payload developer is also responsible for obtaining the required regulatory license grant and submit to the JSC Spectrum Management Office to support the completion of the ISS compatibility review process.

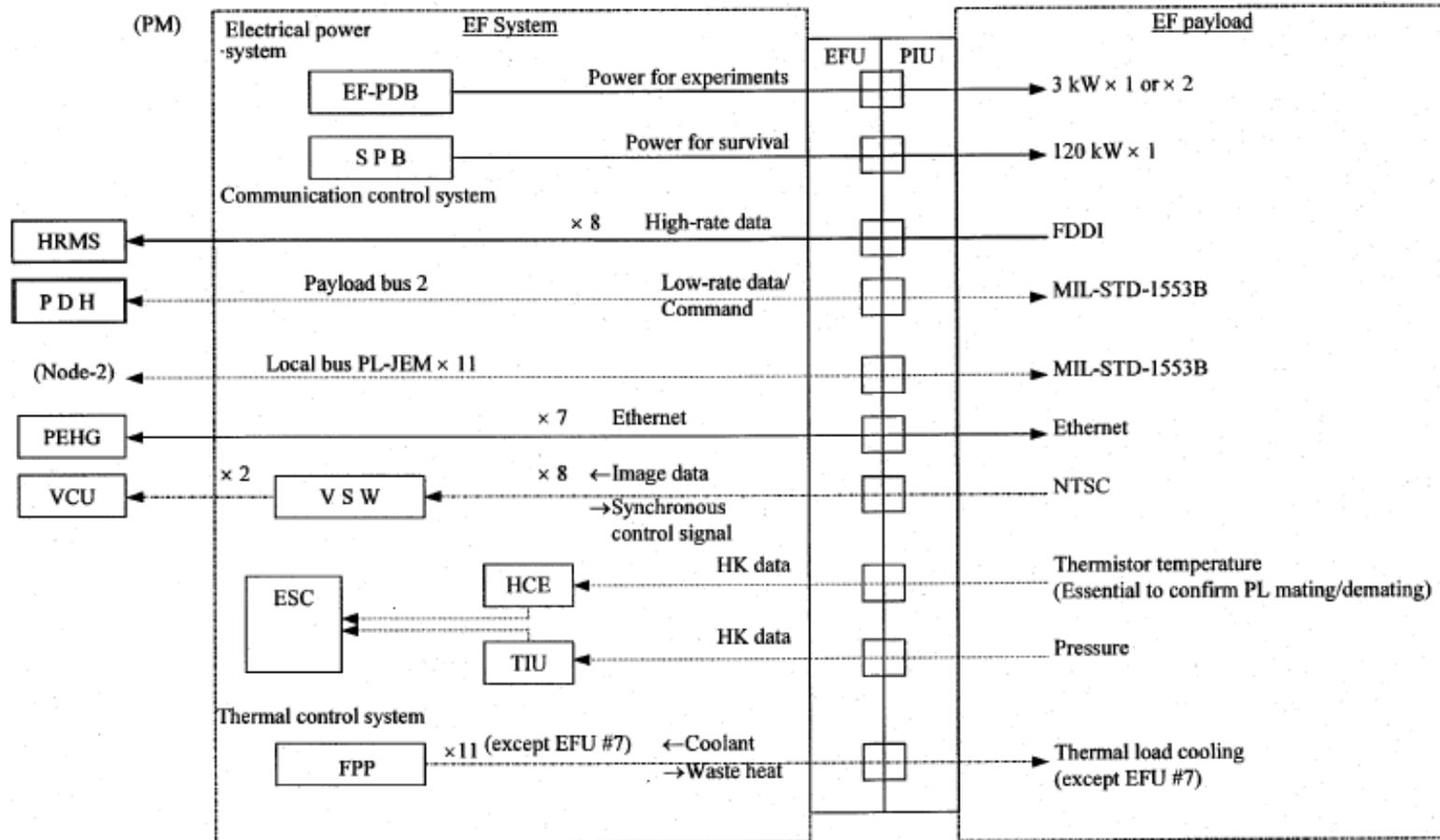


Figure 3.2.5-1. Power and data connections.

3.2.6 Field of View (FOV)

The following provides the proposer a series of fish-eye images of the FOV from those JEM-EF locations available to hosted NASA external payloads, see Figure 3.2.6-1. EFU locations 1 thru 6, and 8, thru 12 are available for payload use. EFU Location 7 is reserved for system use. For each available payload location, a generic payload volume was used, reference Figures 3.2.6-2 and 3.2.6-3. A height (Z-axis dimension for EFU1 thru EFU9) of 40 inches and a width (Y-axis dimension for EFU1 thru EFU8 and X-axis dimension for EFU9) of 33 inches were used to create the payload volumes. The payload volume dimensions used for the EFU11 payload and the payload volume length (X-axis dimension for EFU1 thru EFU8 and Y-axis dimension for EFU-9) are unknown. The viewpoint location for the Forward (ram) images for EFU1 thru EFU8 was the center of each payload volume face normal to that direction. The viewpoint location for the Nadir and Zenith images for EFU1 thru EFU8 was the point 63.64 inches away from the payload interface plane along the center of each payload volume face normal to the image direction. The viewpoint location for the Forward (ram), Nadir and Zenith images for EFU9 was the point 102.64 inches away from the payload interface plane along the center of each payload volume face normal to the image direction. The viewpoint location for the Forward (ram) image for EFU11 was the center of the payload volume face normal to that direction. These images were generated by the MAGIK Team software tools, at ISS/JSC (see Figures 3.2.6-4 through 3.2.6-16). The solid objects in the FOV remain there at all times while the pink shaded areas in the FOV represent the areas swept by the ISS solar (photovoltaic) arrays as they track the sun. A summary of these fish-eye FOV images is provided in Table 3.2.6-1.

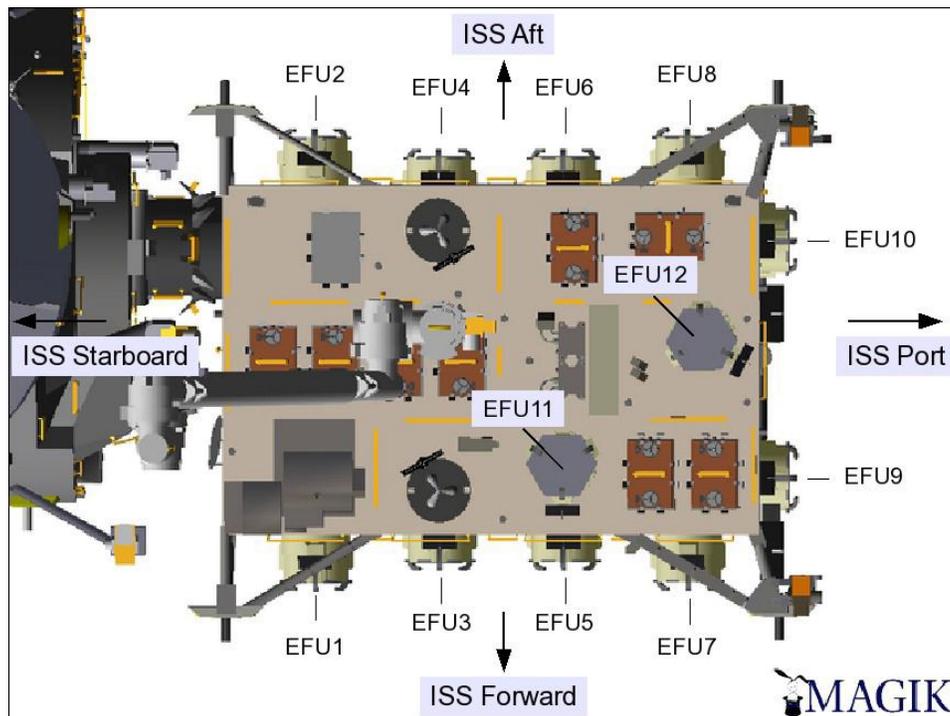


Figure 3.2.6-1. JEM-EF EFU locations and orientations.

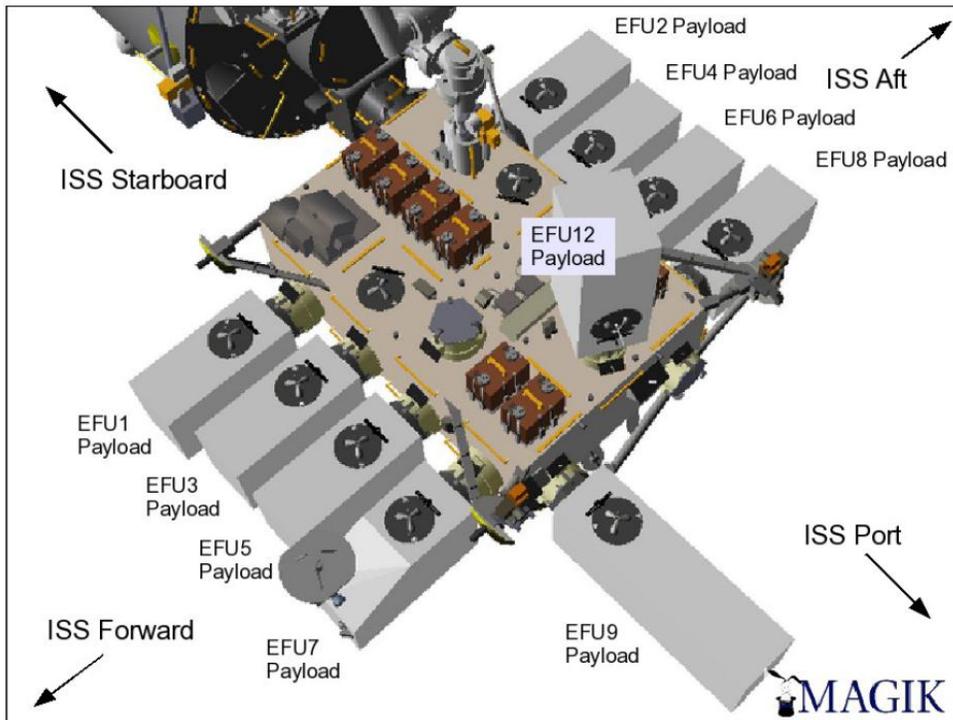


Figure 3.2.6-2. JEM-EF payload volume locations and orientations.

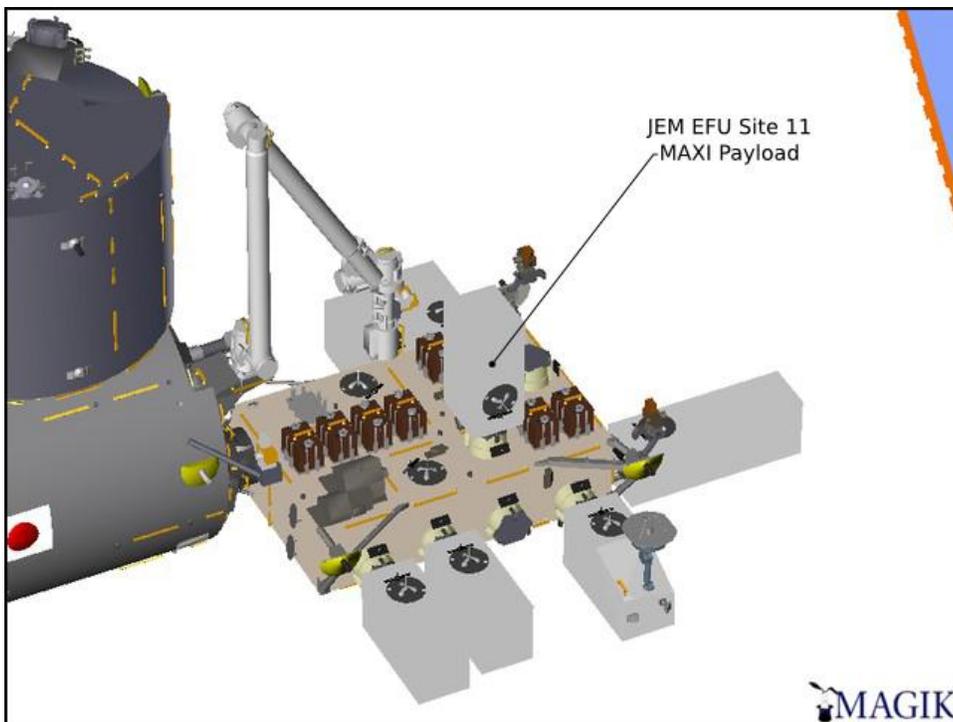


Figure 3.2.6-3. JEM-EF EFU payload volume location and orientation.

Table 3.2.6-1. Summary of these fish-eye FOV images.

Carrier	Location	Optimum Viewing Directions	ISS Aft FOV	ISS Forward FOV	ISS Nadir FOV	ISS Zenith FOV
JEM-EF	EFU1	Ram/Nadir/Zenith		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU2	Nadir/Zenith			<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU3	Ram/Nadir/Zenith		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU4	Nadir/Zenith			<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU5	Ram/Nadir/Zenith		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU6	Nadir/Zenith			<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU8	Nadir/Zenith			<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU9	Ram/Nadir/Zenith		<input type="checkbox"/>	<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU10	Nadir/Zenith			<input type="checkbox"/>	<input type="checkbox"/>
JEM-EF	EFU11	Ram/Zenith		<input type="checkbox"/>		<input type="checkbox"/>
JEM-EF	EFU12	Zenith				<input type="checkbox"/>

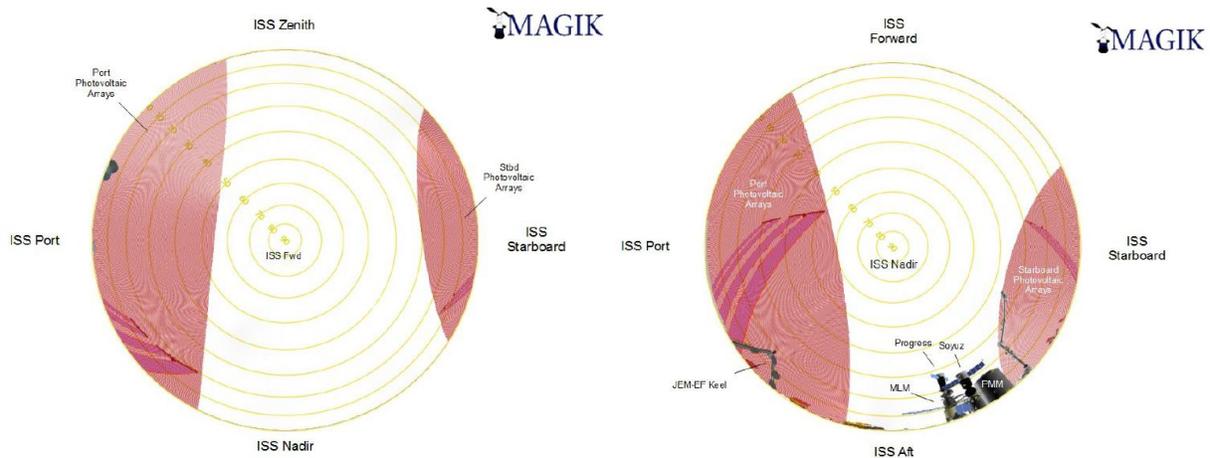


Figure 3.2.6-4. JEM-EF EFU1 P/L accommodation forward (ram) and nadir.

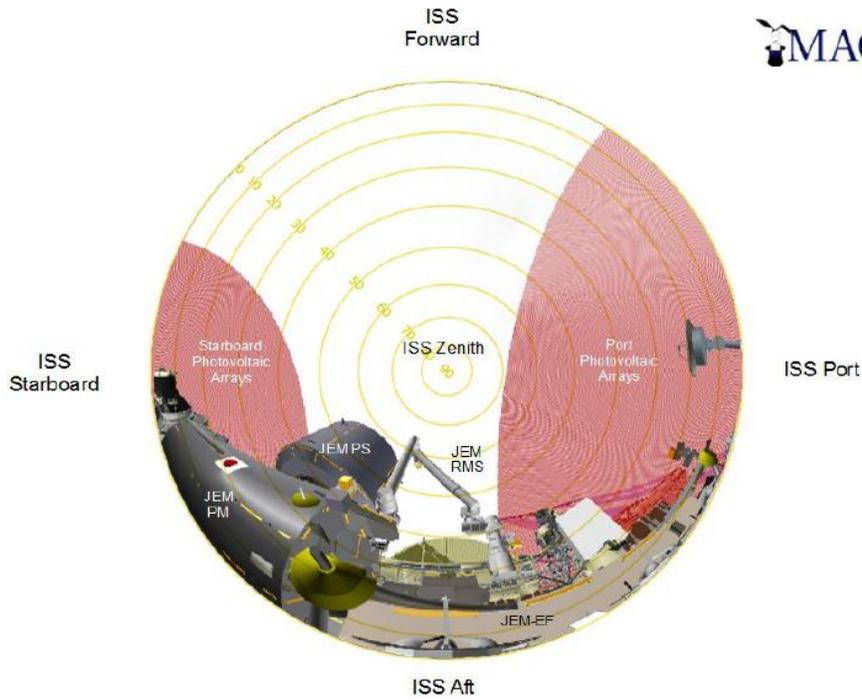


Figure 3.2.6-5. JEM-EF EFU1 P/L accommodation zenith.

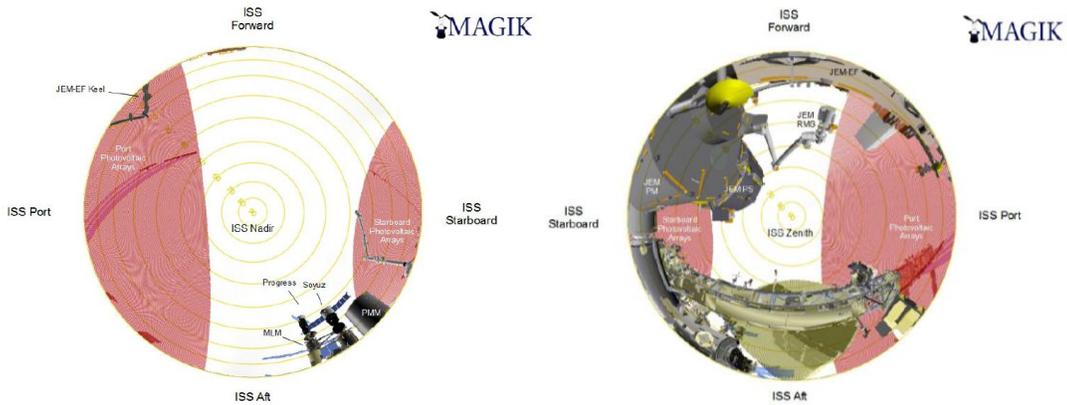


Figure 3.2.6-6. JEM-EF EFU2 P/L accommodation nadir and zenith.

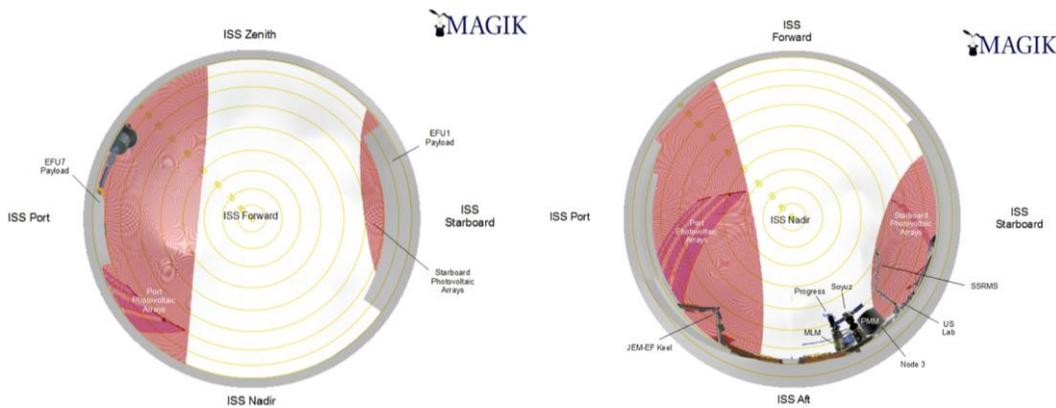


Figure 3.2.6-7. JEM-EF EFU3 P/L accommodation forward (ram) and nadir.

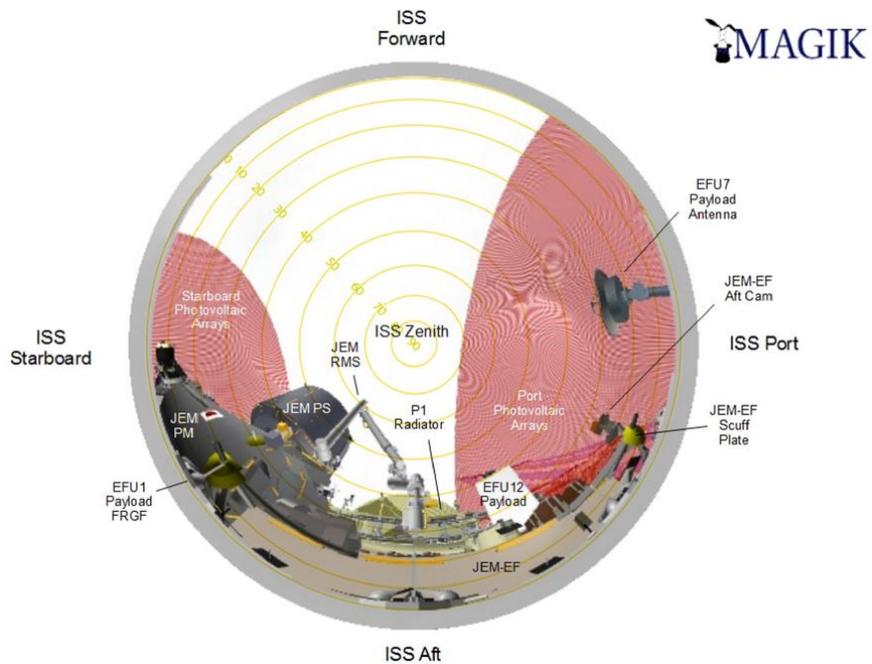


Figure 3.2.6-8. JEM-EF EFU3 P/L accommodation zenith.

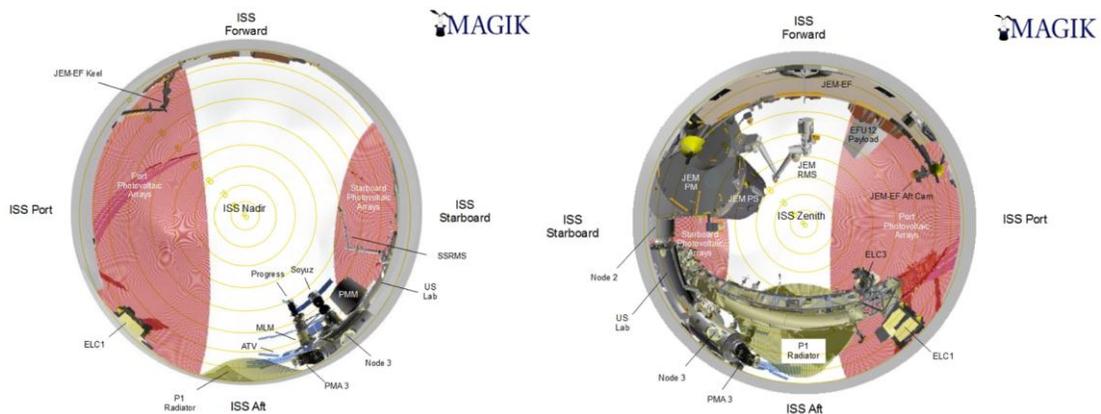


Figure 3.2.6-9. JEM-EF EFU4 P/L accommodation nadir and zenith.

CHECK THE ESP DIVISION WEBSITE AT
<http://espd.gsfc.nasa.gov/isseppg/>
 TO VERIFY THAT THIS IS THE CORRECT VERSION PRIOR TO USE.

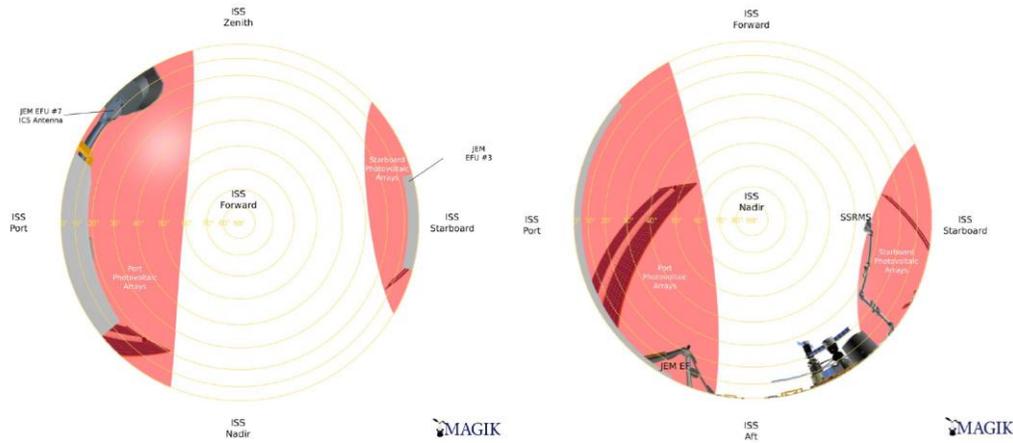


Figure 3.2.6-10. JEM-EF EFU5 P/L accommodation forward (ram) and nadir.

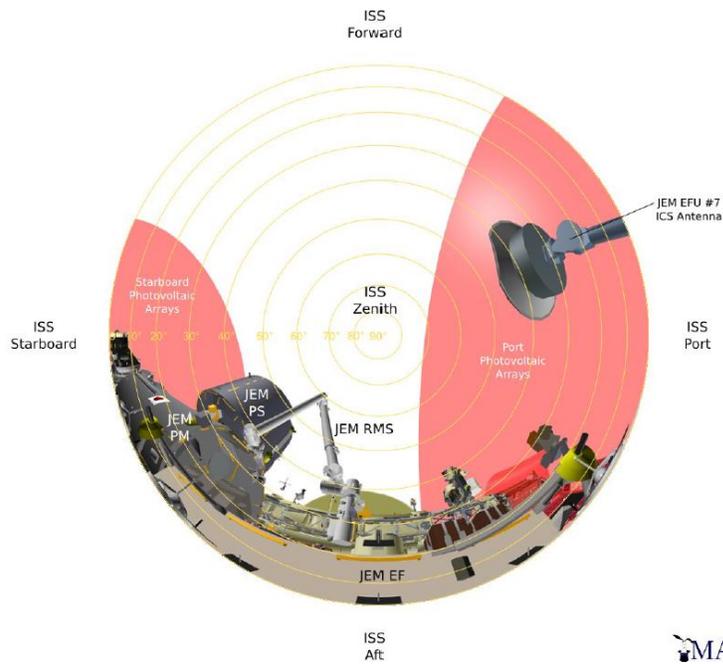


Figure 3.2.6-11. JEM-EF EFU5 P/L accommodation zenith.



Figure 3.2.6-12. JEM-EF EFU6 P/L accommodation nadir and zenith.

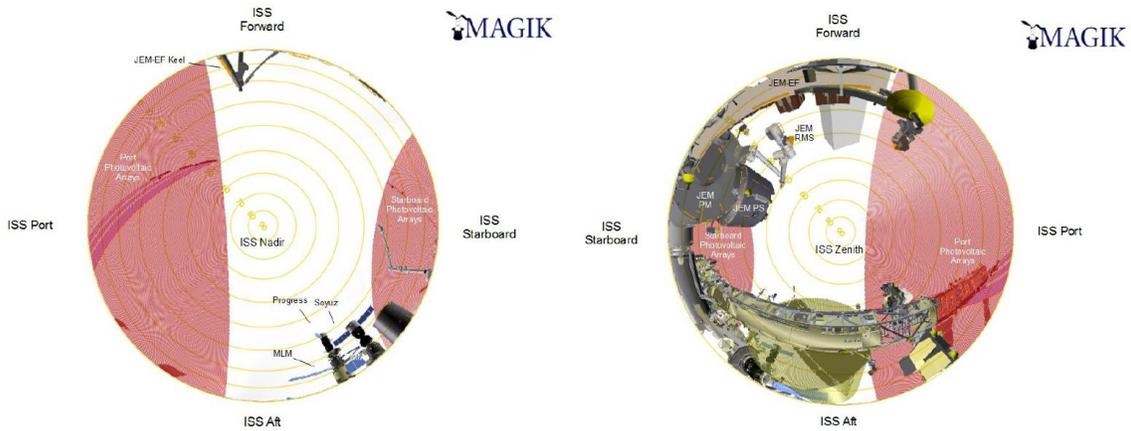


Figure 3.2.6-13. JEM-EF EFU8 P/L accommodation nadir and zenith.

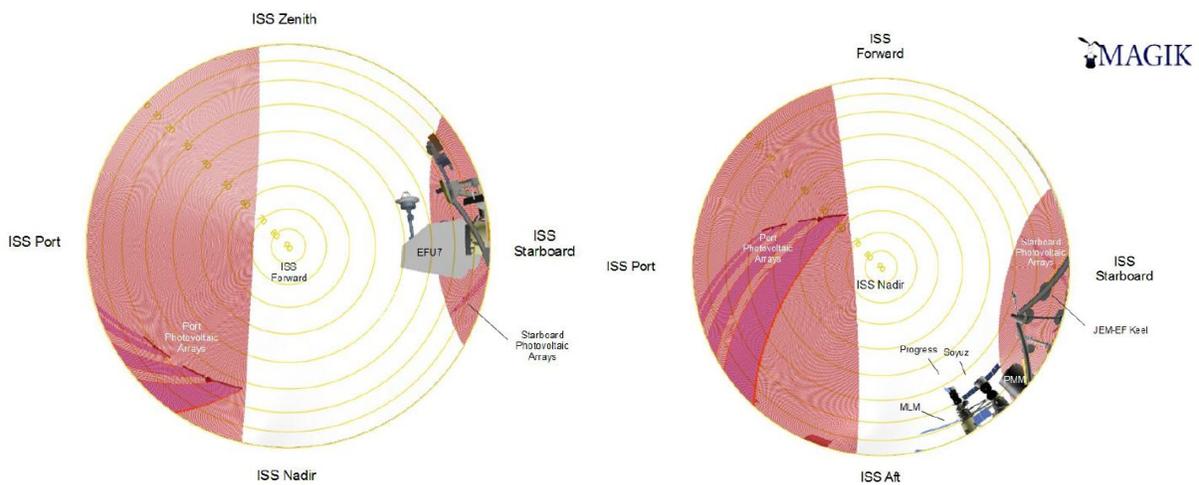


Figure 3.2.6-14. JEM-EF EFU9 P/L accommodation forward (ram) and nadir.

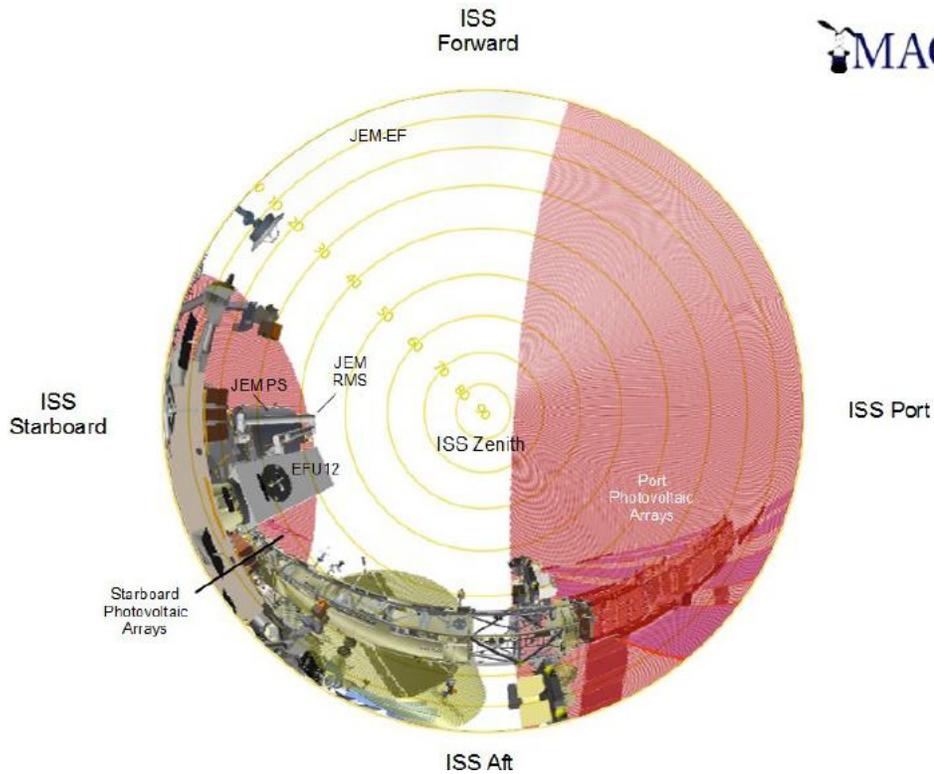


Figure 3.2.6-15. JEM-EF EFU9 P/L accommodation zenith.

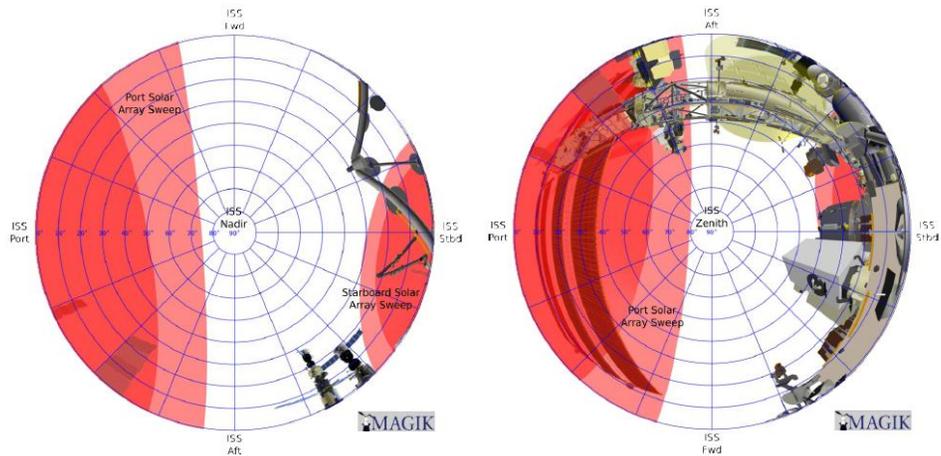


Figure 3.2.6-16. JEM-EF EFU10 P/L accommodation nadir and zenith.

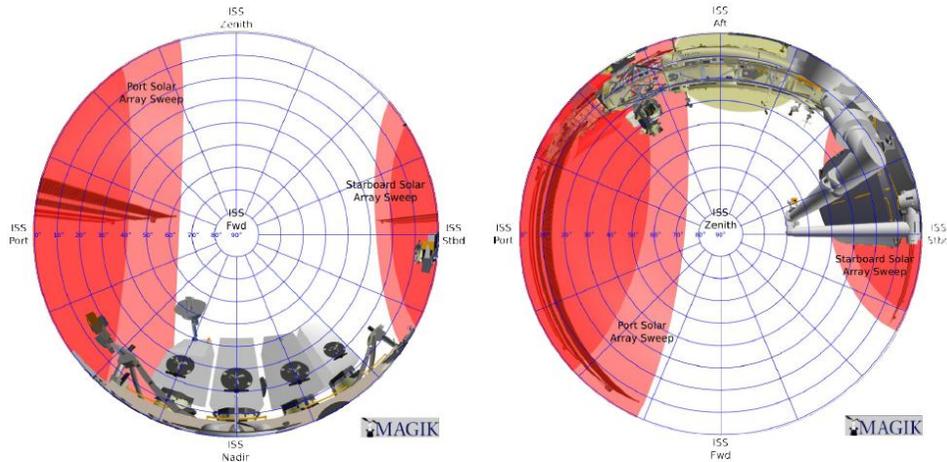


Figure 3.2.6-17. JEM-EF EFU11 P/L accommodation forward (ram) and zenith.

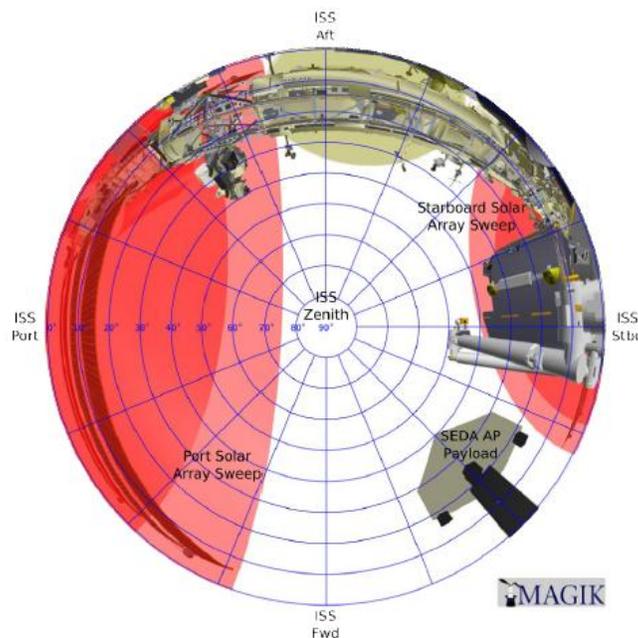


Figure 3.2.6-18. JEM-EF EFU12 P/L accommodation zenith.

3.2.7 Microgravity Environment

The mechanical vibrations generated from all of the ISS systems are transmitted throughout the structure and may affect experiments sensitive to acceleration. Attached payloads are required to restrict their mechanical vibrations so as not to disturb sensitive experiments while the ISS is in microgravity mode. To assure that JEM-EF payloads are able to survive these mechanical vibrations, payloads shall be designed to withstand the on-orbit vibration environment in Table 3.2.7-1. Please note that during nominal operating conditions, additional safety critical loads requirements are levied through your JEM ICD

The ISS Quiescent Mode Truss Vibratory Environment for external payloads is shown for reference, based on data measured on the ISS S3 truss, in Figures 3.2.7-1 and 3.2.7-2. The data were collected while in ISS quiescent mode, with no thruster firings, dockings, EVA, or

robotics operations. It represents typical response, not worst-case, based on a snapshot of three 10-minute data collections, March 2010, with the starboard Solar Array Rotary Joint rotating, three crew, and one exercising. The ELC is predicted to have on the order of 0.03° peak rotations due to jitter, and the JEM-EF should behave similarly.

Table 3.2.7-1. On-orbit random vibration environment.

Frequency (Hz)	Level
10-50	0.0005 g ² /Hz
50-100	+3 dB/oct
100-1000	0.001 g ² /Hz
1000-2000	-3 dB/oct
2000	0.0005 g ² /Hz
Composite	1.3 grms
Duration	10 hr/Year

Note: Accelerations must be evaluated as acting one axis at a time in each of three orthogonal axes.

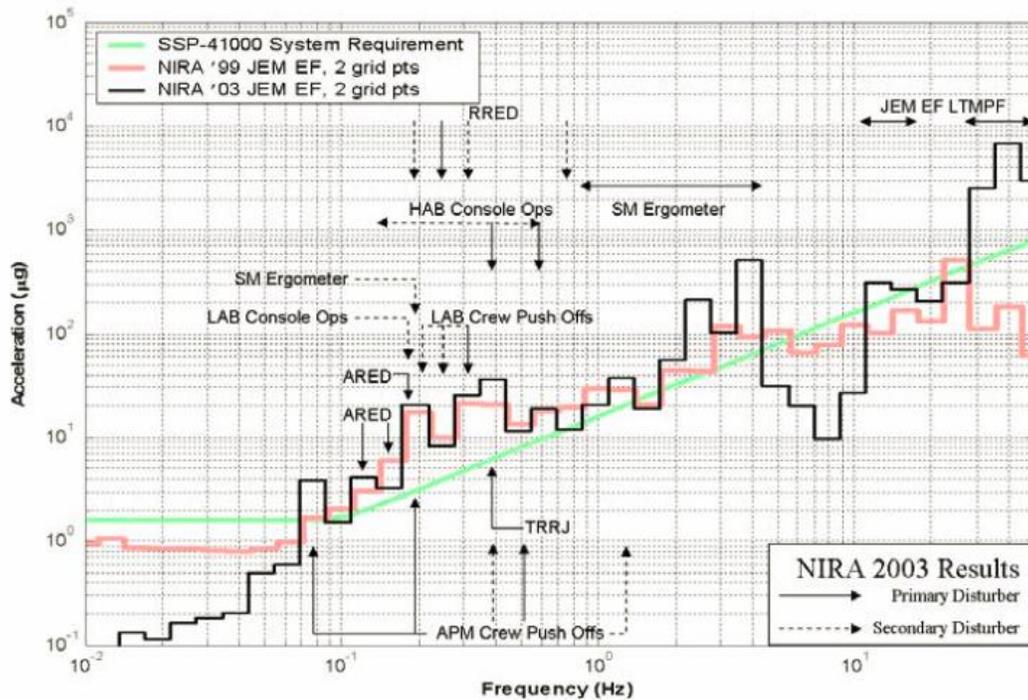


Figure 3.2.7-1. Typical microgravity environment for JEM-EF.

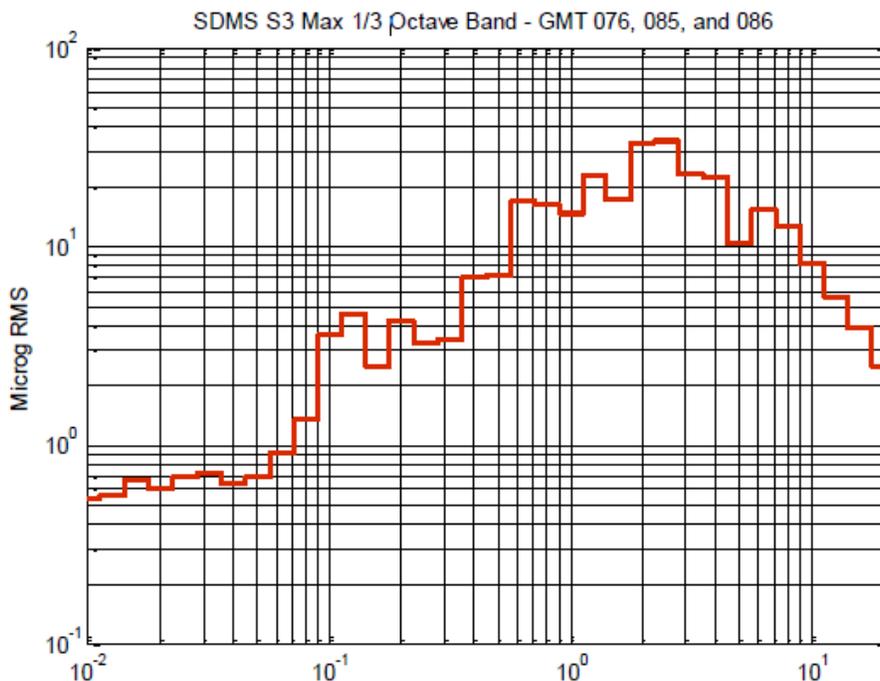


Figure 3.2.7-2. Typical microgravity environment for JEM-EF (RMS).

3.2.8 Electromagnetic Compatibility/Electromagnetic Interference (EMC/EMI)

In addition to the information presented in paragraph 3.1.8, PD should be aware that they must also meet the following radiated emissions limit:

- Frequency Range: 23.54 GHz +/- 1 kHz
- Allowable max electrical field density: -5.9 dB μ V/m

Finally, please note that ISS Management Policy Directive (MD-113) applies to all U.S.-funded intentional transmitting and receiving systems for operation on the ISS. This policy provides the authority and requirement to conduct technical reviews in ensuring RF compatibility of the proposed RF system/usage with existing ISS RF operation.

Regarding IP-managed RF payloads for the ISS, the process for coordination of frequency selection and compatibility assurance will be worked through the JSC Spectrum Manager (designated as the ISS Spectrum Manager) as documented in SSP 50423.¹³ Additionally, regulatory filing and approval for space operation of an IP-managed RF payloads is the responsibility of the IP, consistent with the applicable domestic regulation and process of the IP. The International filing documentation, at the minimum of Advanced Publication Information (API) stage, is to be provided to the ISS Spectrum Manager for final validation of ISS onboard compatibility certification. Upon this validation, the payload will be issued an ISS RF Authorization document for Program ICD and IRD requirements closure.

3.3 Columbus External Payload Facility (Columbus-EPF)

The Columbus-Attached Pressurized Module (APM) is a crew-habitable module attached to the ISS. The location of the APM is illustrated in Figure 3.1.0-1. The external payload sites are referred to as the EPF.

There are four external payload sites suitable for payloads on the Columbus module, two of which are allocated to NASA at any given time. The four sites are designated SOZ, SOX, SDX, and SDN, where the 'Z', 'X', and 'N' denote zenith, X-axis, and nadir, respectively. These sites are depicted in detail in Figure 3.3.0-1. Figures 3.3.0-2a and 3.3.0-2b picture the external payload locations. For Earth Science payloads, the sites SDX and SDN, as shown in Figure 3.3.0-1, are considered optimal. The definitions of the site designations are as follows: SOZ – Starboard Overhead Zenith; SOX – Starboard Overhead X-Direction; SDX – Starboard Deck X-Direction; and SDN – Starboard Deck Nadir.

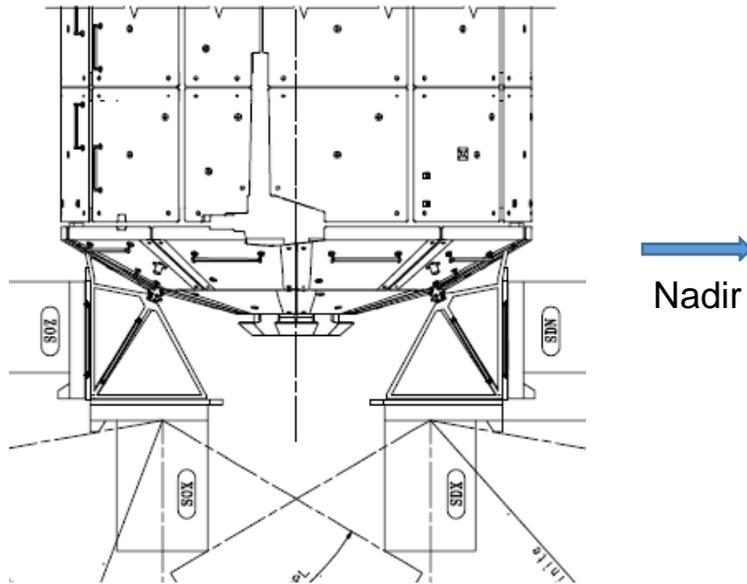


Figure 3.3.0-1. Columbus external payload sites: SOZ, SOX, SDX, and SDN.

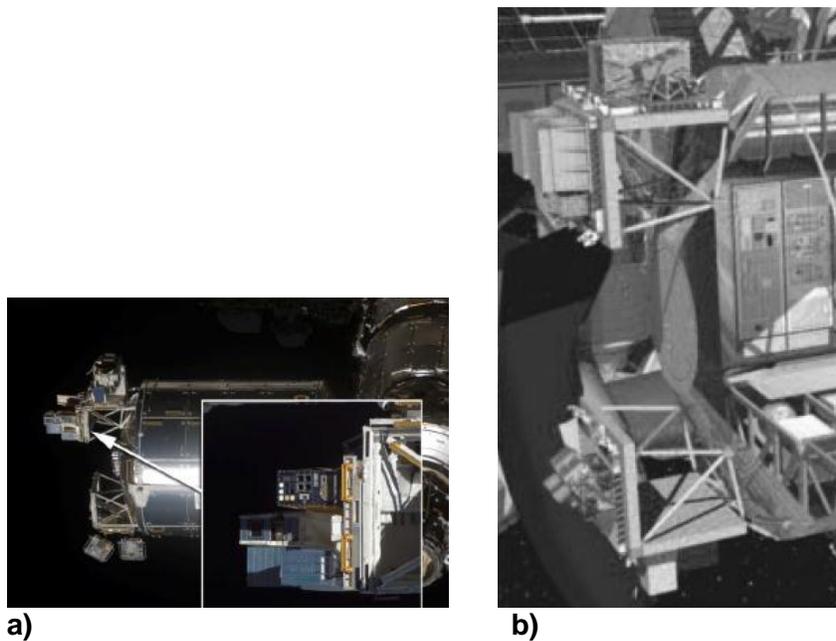


Figure 3.3.0-2. a and b Columbus external payload sites.

The Columbus-APM Coordinate System is shown in Figure 3.3.0-3 and Figure 3.3.0-4.

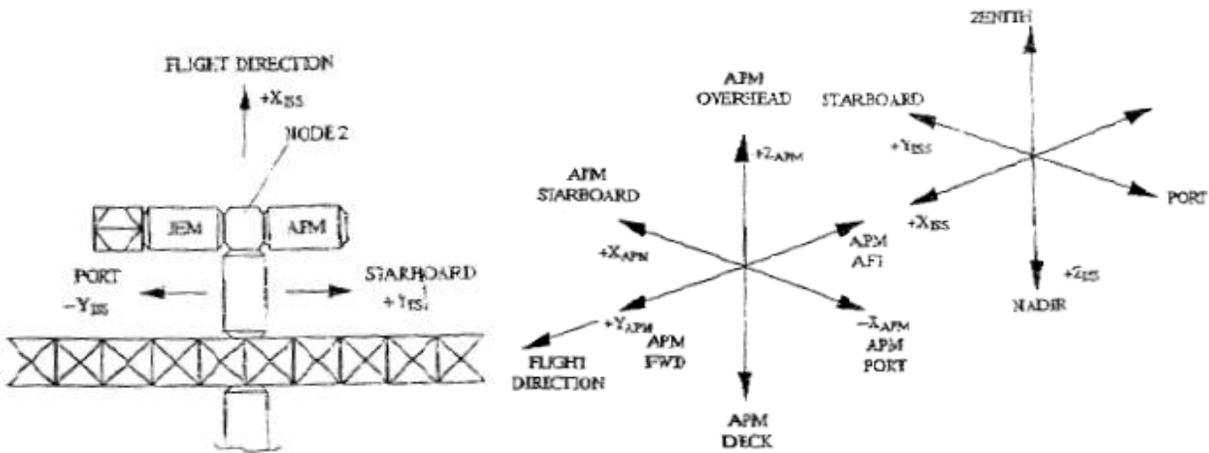


Figure 3.3.0-3. APM/ISS reference coordinate system.

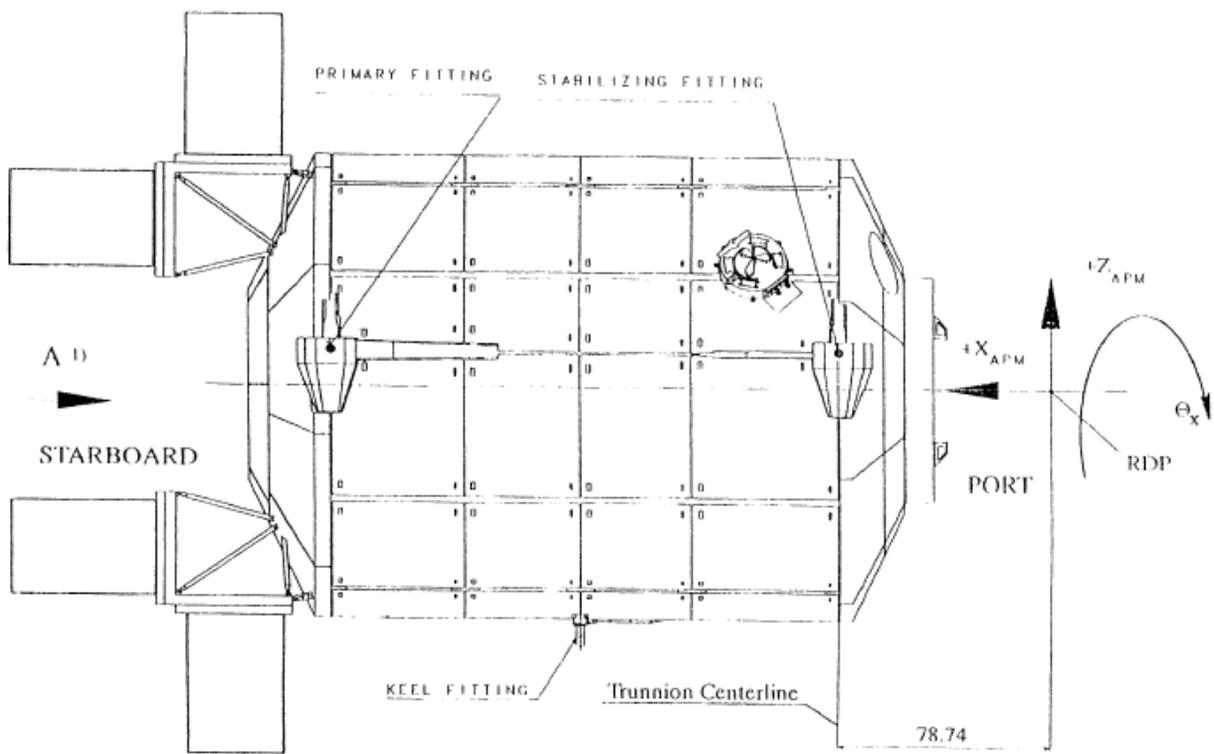


Figure 3.3.0-4. APM coordinate system.

Figure 3.3.0-5 provides the PD with the external payload locations as they are found on the Columbus-APM. In addition, the OTCM locations are provided, which illustrates the position of the DEXTRE (SPDM) robotic interfaces. These robotic interfaces are more clearly defined in Section 3.3.2.

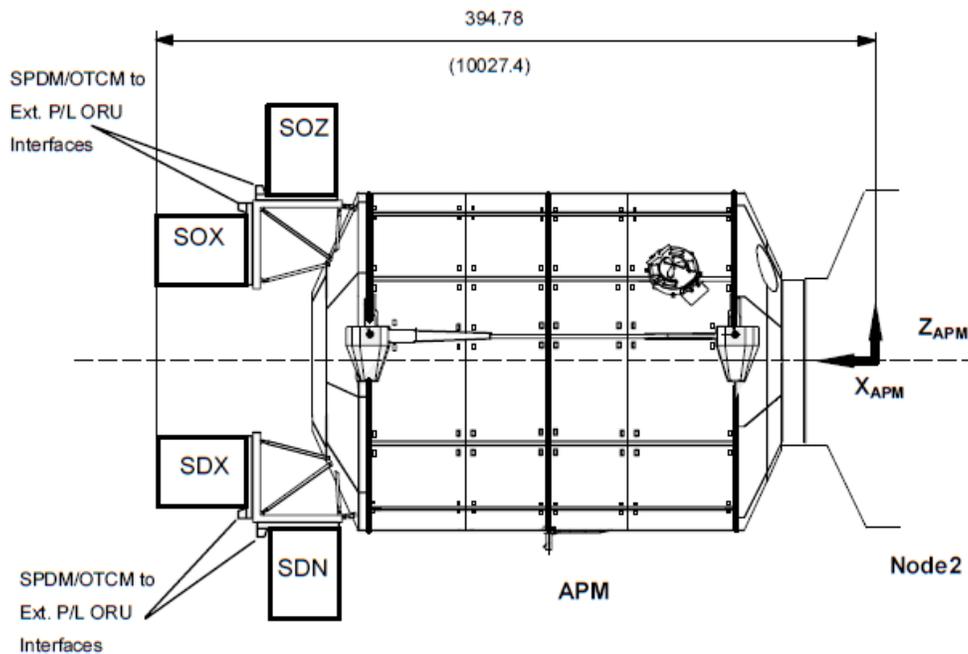


Figure 3.3.0-5. APM with external payloads and OTCM locations.

3.3.1 Physical Accommodations

For attachment to the Columbus-EPF, each external payload will be integrated to a Columbus External Payload Adapter (CEPA), similar to the one used on the ELC for the ITA (truss sites), except for the number of power/data connectors. To accommodate a wide variety of payloads, the CEPA assembly provides standard mechanical and electrical/data interface features. In addition, the CEPA assembly provides standardized structural, electrical bonding, and Ground Support Equipment (GSE) interfaces, that are the similar functions provided by the ExPA for the ELC. The CEPA plate configuration provides the required interfaces for integration with the AFRAM. Each AFRAM is a moving mechanical assembly that consists of close tolerance, precision machined components attached to the bottom of an adapter plate. This arrangement is analogous to the ExPA/AFRAM assembly defined for the ELC in Section 3.1. Figure 3.3.1-1 provides a CEPA isometric view and coordinate system.

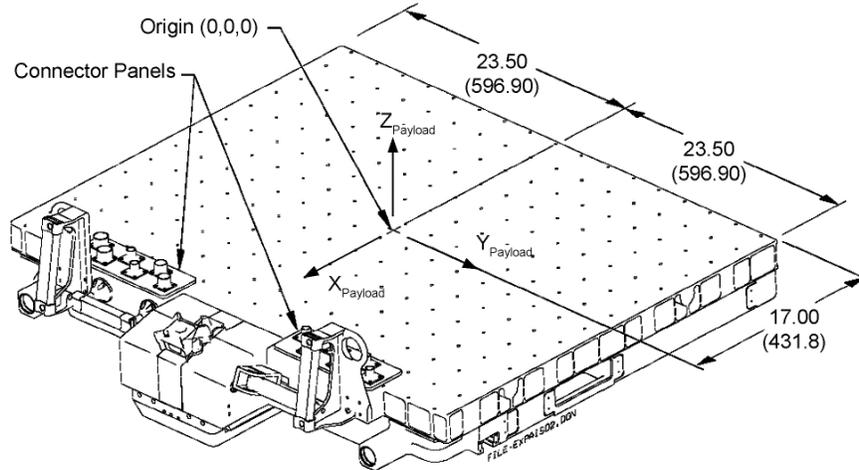


Figure 3.3.1-1. CEPA isometric view and coordinate system.

The CEPA and AFRAM are compatible with the passive FRAM, which is already contained on the Columbus external payload site. Figure 3.3.1-2 illustrates the mechanical attachment. The CEPA interface contains connectors for the electrical interfaces to the Columbus Module. These interfaces are described in Section 3.3.4. You may also refer to Section 3.1 of the Guide for additional information on the ExPA, including views, configurations, and drawing specifics. The structural attachment interfaces for Columbus payloads are identical to those used by ELC payloads, described in Section 3.1. Please refer to D683-97477-01 Revision D²⁰ for additional information.

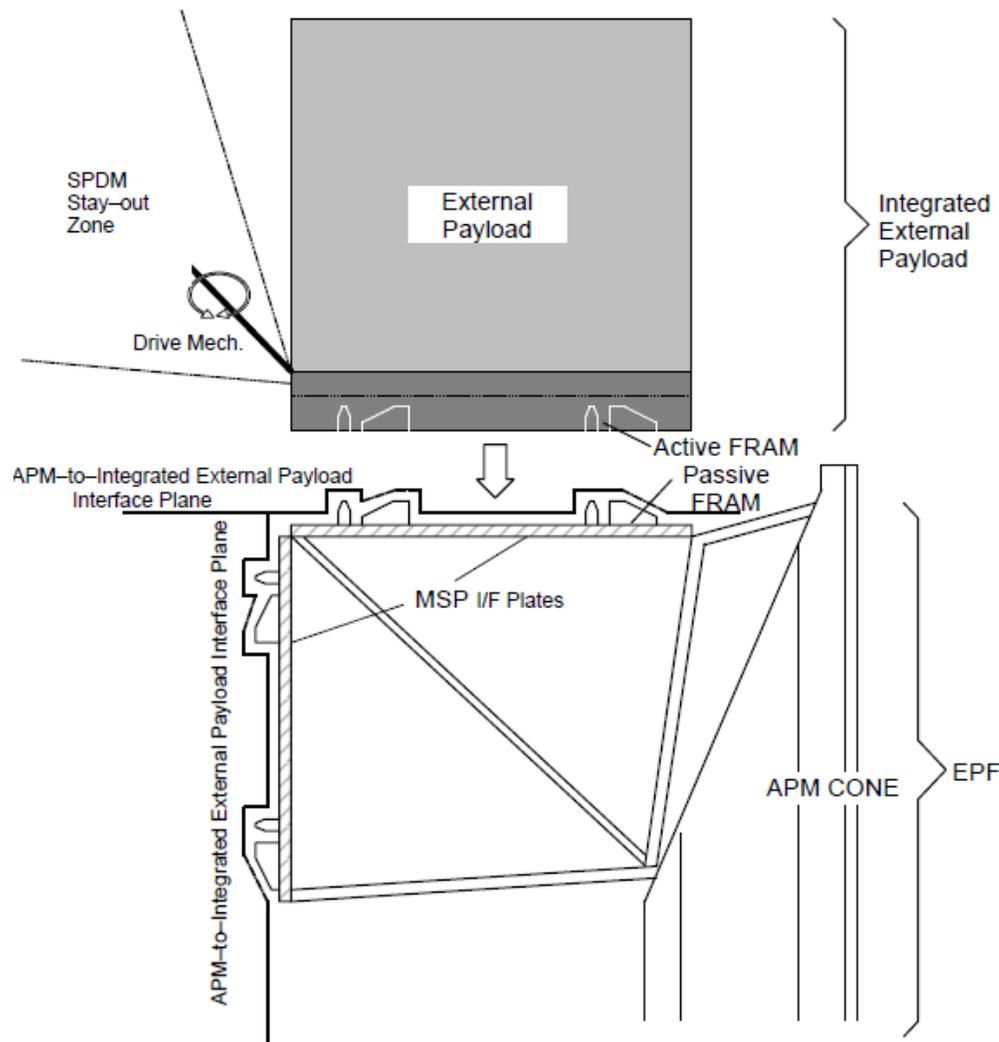


Figure 3.3.1-2. External payload to Columbus attachment.

External Payload Mass Allocation

The maximum mass allowed at each External Payload site is 370 kg including the mass of the CEPA and AFRAM. This leaves approximately 230 kg for the payload itself, or roughly 500 lbs, depending on the specific integrated C.G. location for each payload. The proposer should use 500 lbs as their maximum payload weight.

Payload Envelope and C.G. Requirements

The allowable envelope and center of gravity for the integrated external payload during on-orbit operation (i.e., after docking to PFRAM) are shown in Figure 3.3.1-3 with a maximum height above the interface plane of $z = 600$ mm. The maximum payload envelope ($L \times W \times H$) is also shown in Figure 3.3.1-3.

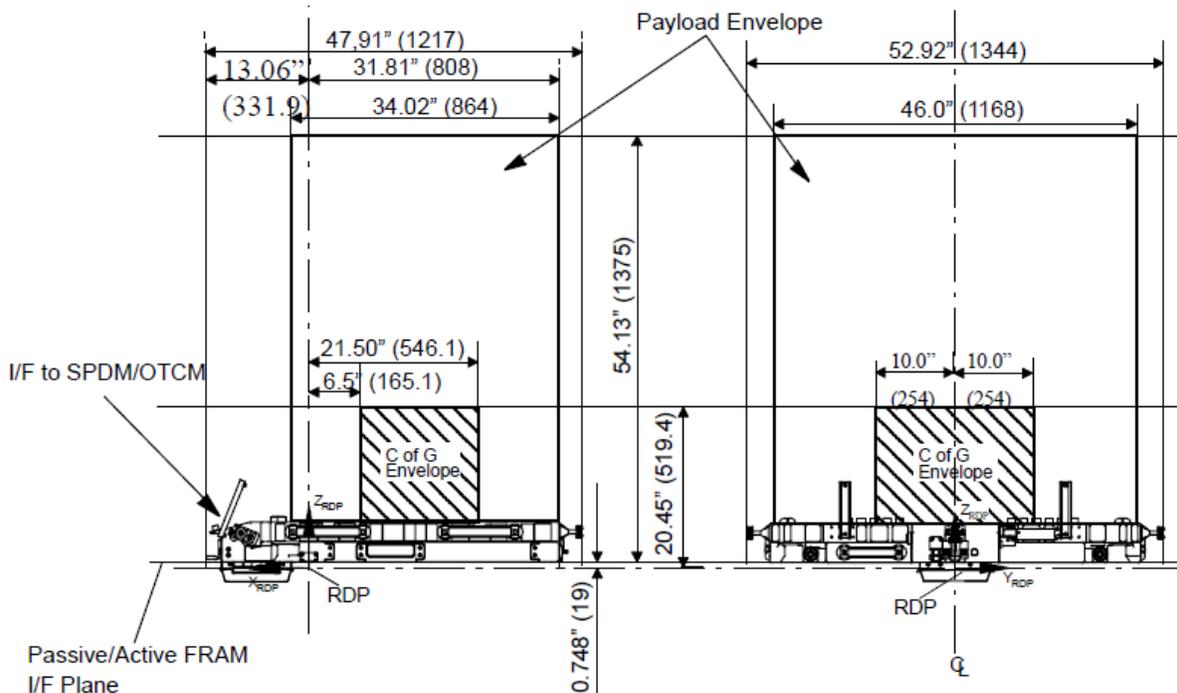


Figure 3.3.1-3. Integrated external payload envelope.

3.3.2 Robotics (SSRMS/SPDM (Dextre))

Please review the robotics section contained in Section 3.0, prior to reviewing the specific robotic requirements/accommodations outlined below.

The SSRMS is used in conjunction with the SPDM, or Dextre, to remove a payload from the trunk of a docked visiting vehicle, transfer the payload to the Columbus-APM location, and perform the installation of the payload onto the external payload location.

Dextre can interface directly with a Columbus external payload, by using the attachments contained on the CEPA. The primary purpose of Dextre is to perform dexterous tasks that require fine, minute control. In the past, such tasks have been performed by EVA crewmembers, so being able to perform these tasks with Dextre removes the risk to EVA crewmembers. Dextre's construction consists of a LEE, the EOTP, a Body Roll Joint, two arms, a tool holster, a PDGF, and four cameras. Figure 3.3.2-1 illustrates Dextre.

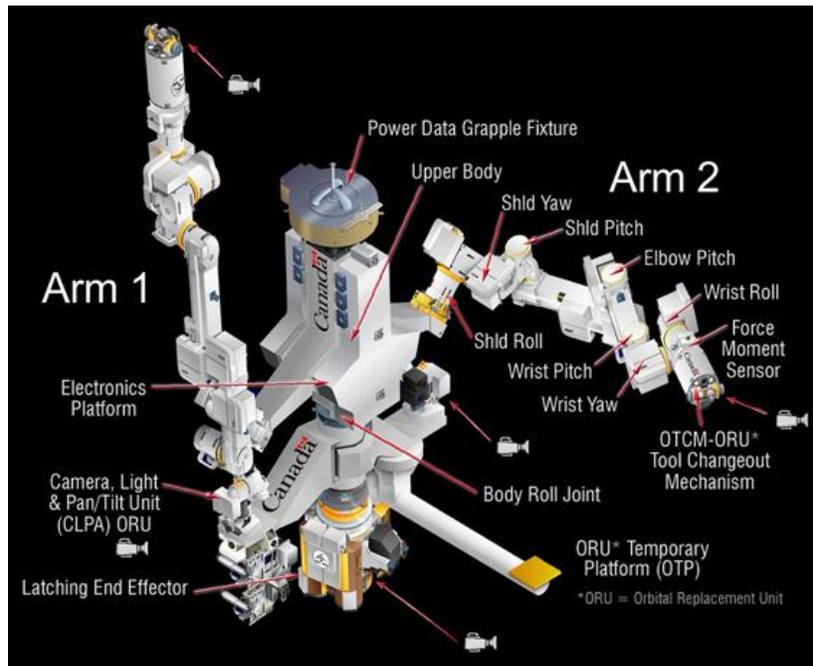


Figure 3.3.2-1. ISS Dextre.

Dextre's manipulation control is provided by two seven-jointed arms, which are both terminated with OTCMs. The OTCMs attach to specially designed fixtures on the CEPA, and feature a retractable motorized socket wrench used to torque bolts, a retractable umbilical connector used to provide electrical, data and video connections to payloads, and a camera and lights used for close-up viewing and to align the OTCMs with the fixtures. Figure 3.2.2-2 illustrates the OTCM.

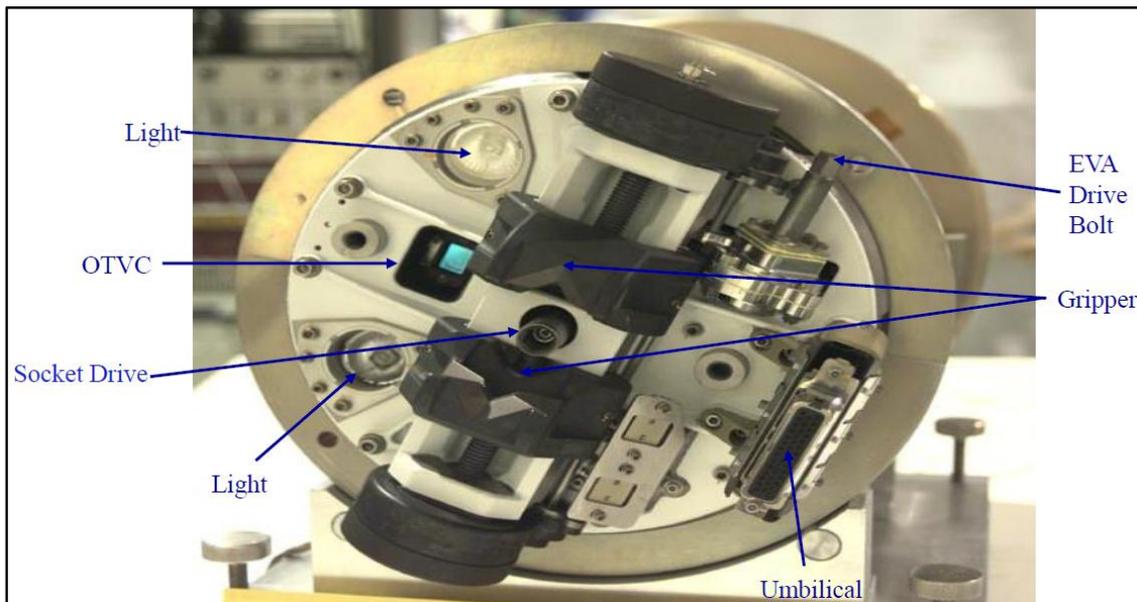


Figure 3.3.2-2. Dextre OTCM.

A SpaceX Dragon carrying a Columbus EPF payload in the Dragon Trunk will be attached at the designated ISS node location. The SSRMs will pick up the Dextre robot, and bring it to near the Dragon. Dextre will reach into the Dragon trunk and an OTCM will engage the fixture on the CEPA (active FRAM portion) of the integrated attached payload. The payload will be released from the Dragon the OTCM driving a bolt with the torque wrench to release the connector and the four clamps, and then Dextre will remove the payload. After bringing the external payload to the Columbus APM, it will be installed onto the appropriate external location. After release of the Dragon connector, the payload will have no survival heater power and must be able to survive a minimum of six (6) hours without power.

3.3.3 Thermal

It should be noted that the Columbus external payload sites do not provide active thermal control interfaces, thus the external attached payload will need to be compliant with the following section. The External Payload Site Temperature range with no Payload attached (EPF/PFRAM) will be between $-72\text{ }^{\circ}\text{C}$ to $+48\text{ }^{\circ}\text{C}$. The mechanical functionality of the active and PFRAM design (see Section 1.2 for interface plane definition) during berthing and un-berthing will be assured if the PD plans for thermally conditioning the integrated external payload to a temperature range between $-21\text{ }^{\circ}\text{C}$ ($-5.8\text{ }^{\circ}\text{F}$) to $+49\text{ }^{\circ}\text{C}$ ($+120.2\text{ }^{\circ}\text{F}$). For analytical integration purposes, the payload should not rely on conductive heat exchange across the integrated payload to EPF interface.

The reflected incident solar and incident infrared (heat) radiation from the integrated external payload onto the surfaces of other on-orbit segments cannot exceed 30 W/m^2 , average on the EPF as a whole and 60 W/m^2 , locally on the EPF, as calculated over the individual area of each node belonging to the ISS TMM for external P/L mounted on the Columbus External Payload Facility (CEPF).²¹ Peak values during on orbit must not exceed twice the above values for both incident solar energy and heat radiation.

The proposer of an integrated external payload will need to consider their payload being exposed to and must be compatible with the thermal solar constants, albedo and Earth OLR environments as defined for all operational modes in Table 3.3.3-1 and 3.3.3-2.

The proposer should consider induced thermal environments for the integrated external payload. It will be exposed to and must be compatible with the induced thermal environments from vehicle(s) docking and docked with the ISS, and thermal interactions with other on-orbit segments. The analysis of the induced thermal effects on the integrated external payload should be based on the beta angle extremes, orbital altitude, and altitude variation about the ISS vehicle axes are provided in Table 3.3.3-2.

For reference of the thermal interface to the ISS and the EPF, the proposer can use the induced thermal environment determined based on the model definitions in:

Integrated ISS Thermal Math Models, Volume 3 Book 2, October 31, 1999

D684-10058-03-02, International Space Station Program, Boeing/ NASA, JSC (R.D. 2.2.9)

APM Simplified Thermal Mathematical Model Description for ISSA/APM/I/F Thermal Analysis, CLT-TN-AI-0038, Issue 6, 24.05.2005 (R.D. 2.2.10)

The corresponding definition of the overall thermal interface model is provided in:

ISS TMM for External P/L mounted on the CEPF²¹, TEC-MCV/2005/3075/In/HFLR,²² (A.D. 2.1.42). A.D. 2.1.42 should be used for thermal modeling of the integrated external payload induced thermal environment.

Thermal Models

Key thermal considerations that must be addressed by the external Payload Developer (PD) are the development and delivery of a thermal model in required format and generation of thermal analyses to demonstrate that the payload will remain safe during Launch-To-Activation (LTA), as well as applicable on-orbit stowage sites on the ISS. These models will be provided to support required thermal assessment by the Launch Vehicle integrator and the ISS vehicle integrator.

The external PD will be required to develop and deliver a thermal model compatible with Thermal Desktop[®] and TRASYS/SINDA. The model is expected to be comprised of no more than 500 geometric and 500 mathematical nodes that represent all geometric and material properties required for determining critical hardware temperatures. These models will be utilized by the PD to assess the external payload in its on-orbit configuration, including the time periods while stowed in the Launch Vehicle and on the ISS Columbus EPF. These models will be provided to the ISS Program in order to support required thermal assessments by the Launch Vehicle integrator and the ISS vehicle integrator.

Thermal Analysis

The external attached payload must remain safe during the Launch To Activation phase while stowed in the Launch vehicle, as well as, during the subsequent phase while it is stowed on an ISS Columbus EPF when it is exposed to a space sink temperature of 3 K, the thermal environment parameters defined in Table 3.3.3-1, the ISS flight attitudes and solar beta angle ranges defined in Table 3.3.3-2 (similar attitude requirements will be defined for the applicable Launch Vehicle) and the thermal interactions with all other on-orbit hardware. Verification of safe operation will require PD generated thermal analysis using ISS Program provided Launch Vehicle and ISS thermal models integrated with a PD developed thermal model of the proposed payload.

Table 3.3.3-1. Thermal Environment Parameters

Case	Solar Constant (W/m ²)	Earth Albedo	Earth OLR (W/m ²)	Altitude (km)
Cold	1321	0.2	206	500
Hot	1423	0.4	286	278

Table 3.3.3-2. ISS Flight Attitudes

ISS Attitude Name	Attitude Reference Frame	Solar Beta Range (β)	Yaw	Pitch	Roll	Time in Attitude
+XVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	-15° to $+15^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
-XVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	$+165^\circ$ to $+195^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
+YVV +Z Nadir	LVLH	$-75^\circ \leq \beta \leq +10^\circ$	-110° to -80°	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
-YVV +Z Nadir	LVLH	$-10^\circ \leq \beta \leq +75^\circ$	$+75^\circ$ to $+105^\circ$	-20° to $+15^\circ$	-15° to $+15^\circ$	No Limit
+ZVV -X Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	-15° to $+15^\circ$	$+75^\circ$ to $+105^\circ$	-15° to $+15^\circ$	3 Hours
-ZVV -X Nadir	LVLH	$-75^\circ \leq \beta \leq +75^\circ$	$+165^\circ$ to $+195^\circ$	$+75^\circ$ to $+105^\circ$	-15° to $+15^\circ$	3 Hours

3.3.4 Power (and Electrical Interfaces)

Electrical Interfaces

The external payload electrical specific system interfaces are contained in Figure 3.3.4-1.

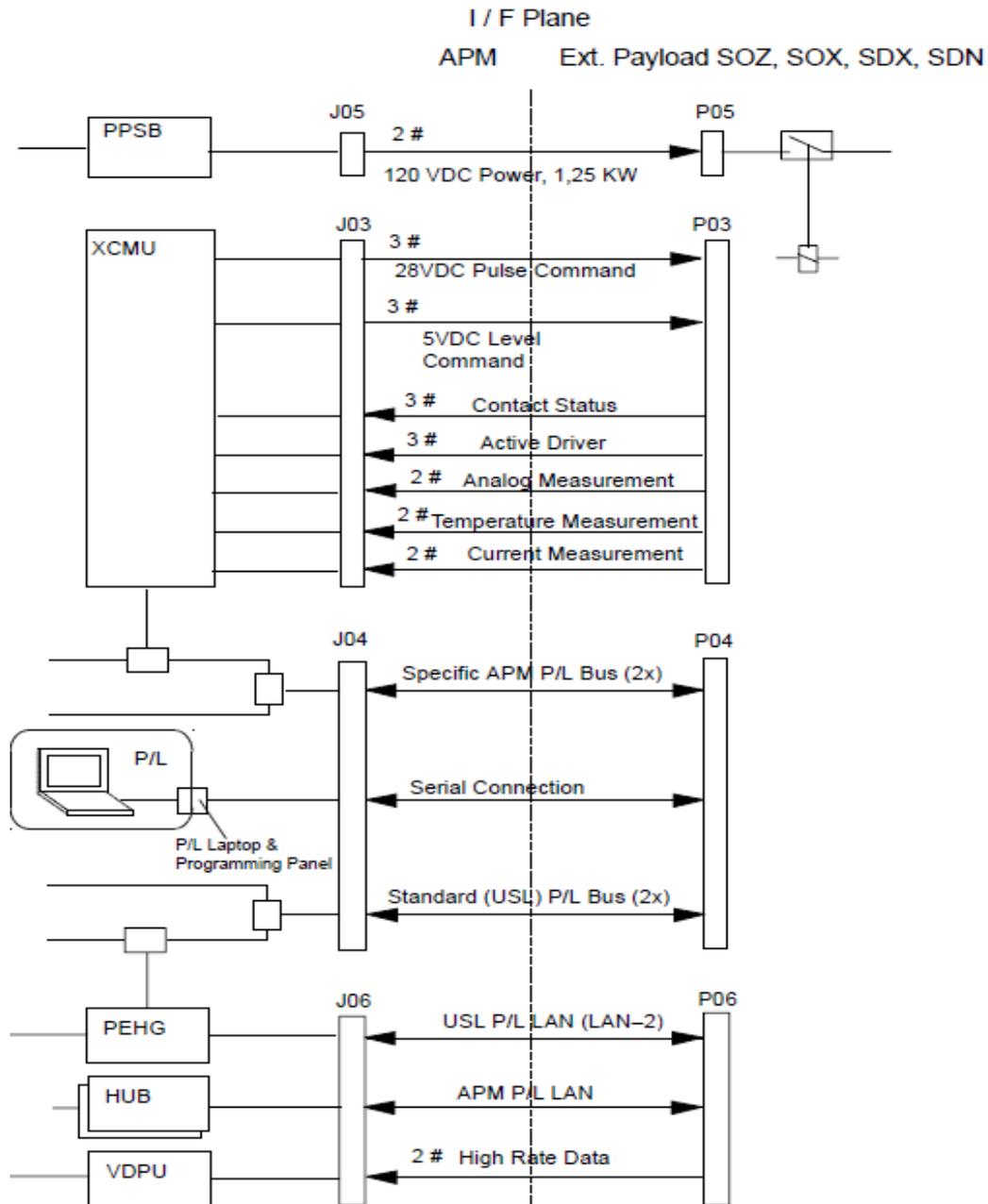
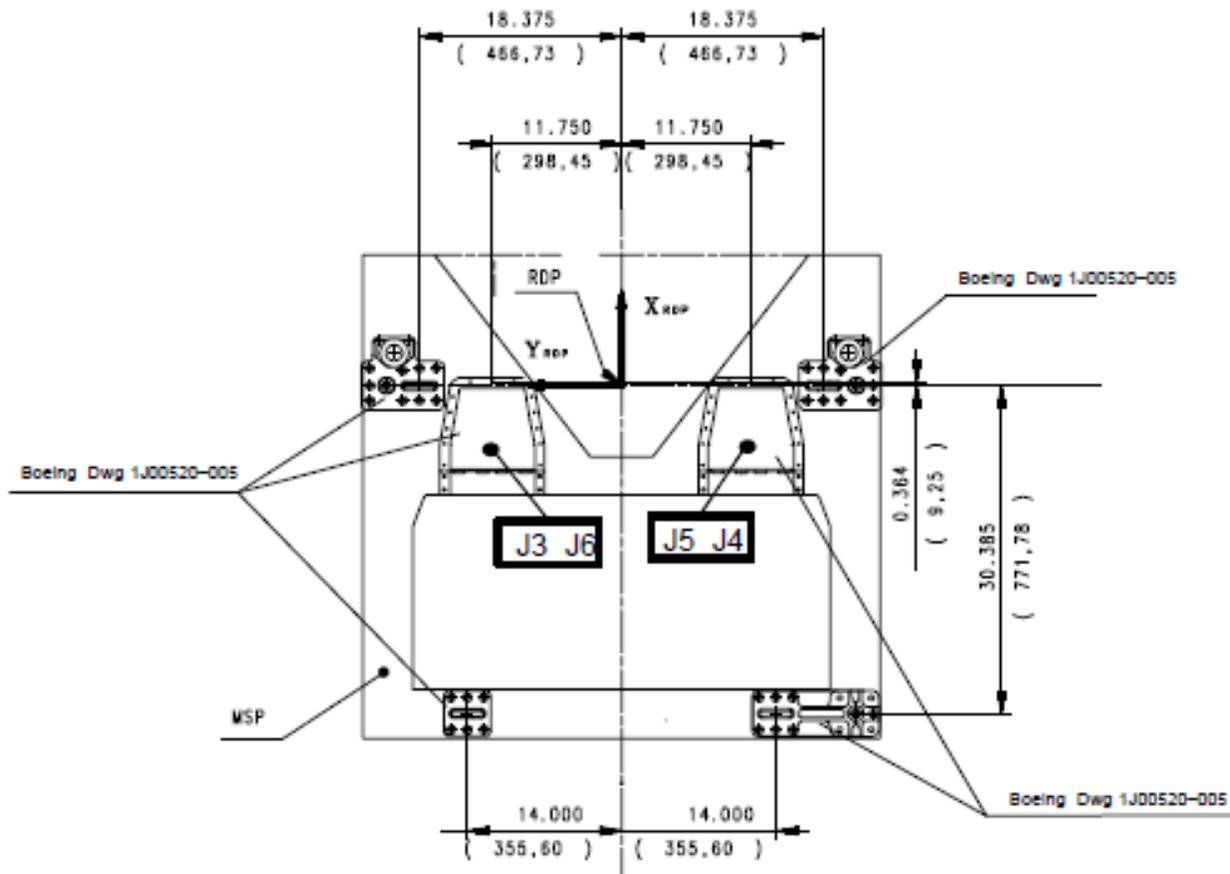


Figure 3.3.4-1. Integrated external payload system interfaces,

These system interfaces are accommodated by four electrical connectors built into the CEPA shown in Figure 3.3.4-2.



- Notes:
- 1) Interface treatment: Alodine acc. to MIL-C-5541 Class III
 - 2) Bonding: metal to metal

Figure 3.3.4-2. Electrical connectors.

Power

The EPF external attached payloads are provided power (via the electrical connectors, J05/P05) as illustrated in Figure 3.3.4-3. Each external payload site is provided two power feeds, Feeder 1 and Feeder 2. Each feed is 1.25 kW with a maximum limit of 2.5 kW. The nominal steady-state voltage range at the external payload interface is 120 Vdc, and ranges from 113 Vdc to 126 Vdc. There are certain limitations and caveats associated with the use of these power feeds. These limitations can be found in the COLUMBUS External Payloads Interface Requirements Document, COL-RIBRE-SPE-0165, Section 3.2.²³ The referenced section also provides additional detailed information on electrical power characteristics and cable, wire, and connector design requirements.

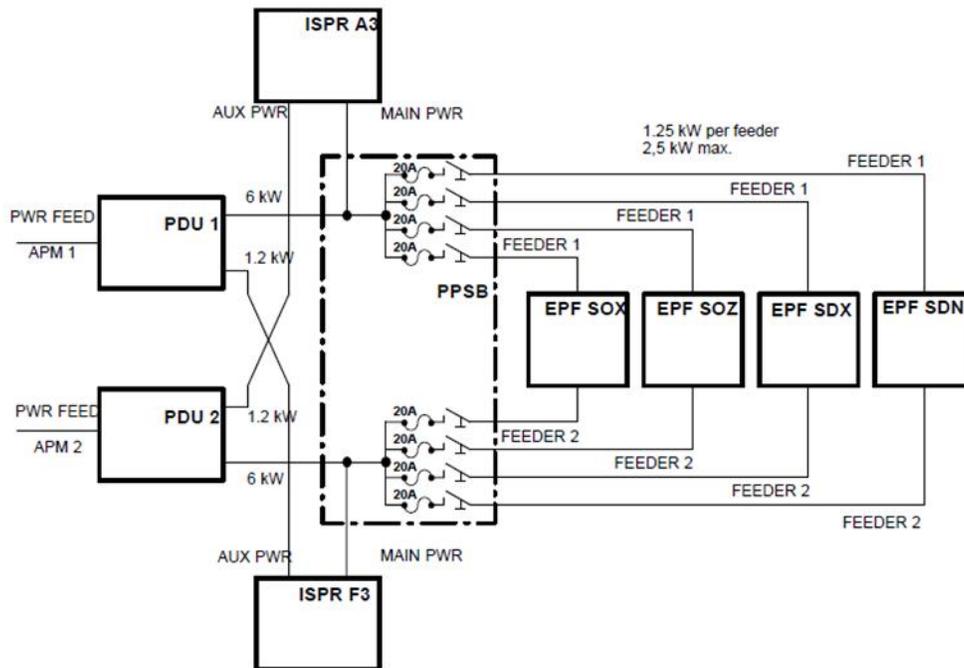


Figure 3.3.4-3. External payload power distribution.

Loss of Power

In the event of a total or partial loss of power, regardless of the availability of the second (redundant) power feeder, the integrated external payload should fail safe in accordance with Payload Safety Policy and Requirements for the International Space Station, SSP 51700 (A.D. 2.1.49).²⁴

Inadvertent Power On/Off Sequence

In the event of an inadvertent APM PDU ISPR outlet power ON/power OFF sequence, the integrated external payload should remain in a safe condition.

3.3.5 Command and Data Handling (C&DH)

For each Columbus external payload there are various data interfaces, which are described below. The C&DH interfaces are provided via connectors J3/P3, J4/P4, and J6/P6 (reference Figure 3.3.4-2). The interfaces include a MIL-STD-1553 data bus, Ethernet, TAXI high rate data, and analog and digital (discrete) signals. The specifics of these interfaces are described in the following paragraphs. Detailed C&DH interface characteristics and requirements can be found in the COLUMBUS External Payloads Interface Requirements Document, COL-RIBRE-SPE-0165, Section 3.3.²³

Low-Rate Data Link

An extension of the U.S. payload MIL-STD-1553B data bus will be used for low rate data transfer, e.g., commanding, time and ancillary data broadcast, file transfer, payload health and status, and payload science data, if chosen. The payload will operate as a RT, and up to 2 U.S. payload RTs are supported. Refer to Columbus External Payloads IRD, COL-RIBRE-SPE-0165, Section 3.3.4.²³

Medium Rate Data Link (Wireless)

A MRDL path using a wireless Ethernet data link is available for Columbus-EPF payloads. It consists of a two-way high data rate communications link using radio frequencies per IEEE 802.11n. This EWC system provides two-way data transfer between the payload sites and the PEHG in the USL. Data rates available to the payload are dependent upon the specific Columbus-EPF location. Average payload user data rates of 3Mbps are easily supportable; higher data rates are possible but require coordination with other payloads.

(Although NASA has completed a preliminary coverage map for the EWC System, this information is not included as assumptions made in generating the coverage map may not apply to the Columbus-EPF external payload utilizing the EWC. External payloads that will utilize the EWC should coordinate with NASA to obtain coverage information. The EWC user should refer to International Space Station (ISS) ELC Wireless Comm (EWC) User Guide, D684-14957-01,²⁵ for additional information).

Medium Rate Data Link (Wired)

This interface is part of the MRDL Ethernet 10/100 Base T network. Downlink via the PEHG gateway per SSP 52050 Rev. K, Section 3.3.5.1 is dedicated to U.S. payloads.⁶

High-Rate Data Link (HRDL)

This one-way high rate data transmission path operates at an encoded signaling rate of 100 Mbps. It uses the TAXI protocol and data rates up to a limit of 32 Mbps can be achieved by parsing the data with sync symbols. Details of this interface are in Section 3.3.6 of the Columbus External Payload IRD.²³

XCMU Interfaces

Each EPF payload site is provided from the Columbus DMS with three contact status inputs, three active driver status inputs, two analog signal inputs, two current measurement inputs, two temperature sensor inputs, three 5 V level commands, and six 28 V pulse commands. NASA payloads may use these interfaces. Detailed interface characteristics may be found in Section 3.3.7 of the Columbus External Payloads IRD.²³

Serial Interfaces

The Columbus module provides for each EPF location a 4-wire communication link. This link may be used by the external payload to transmit serial commands and data between a module internal Laptop and the external payload control computer for reprogramming purposes. The payload must supply the Laptop.

Wireless – (ISS Payloads Office will provide further definition) A wireless data link is available for Columbus payloads. It consists of a two-way high data rate communications link using radio frequencies per IEEE 802.11n. This EWC system provides two-way data transfer between some JEM-EF payload sites and the PEHG in the USL. Data rates available to the payload are dependent upon the specific attach point and will be provided at a later date. Average payload user data rates of 3Mbps or less typically are easily supportable; higher data rates are possible but require coordination with other payloads.

The NASA-provided EWC Payload Hardware GFE consists of the following items:

Payload Antenna Coax Cable Assembly, P/N 684-015637-0001 – Connects EWC Network Interface Card to payload antenna.

EWC Payload Antenna, P/N 684-015636-001 – A linearly polarized 17dBi 16° beamwidth antenna, 9 x 9 x 3 inch, must be pointed toward EWC system on USL.

EWC Network Interface Card, P/N 684-15644-001 – USB form factor WiFi client, 2x2 MIMO, collects payload data for transmission by the payload antenna.

MOXA AWK-4131-M12-CT-T Wireless Access Point – WAP form factor WiFi client, 2x2 MIMO, collects payload data for transmission by the payload antenna.

The payload developer must identify if they are using the NASA provided EWC GFE or if they are developing their own antenna to connect to the external wireless Ethernet. If the payload developer is using the EWC GFE, they must define if they choose to interface the wireless Ethernet through either the NIC or the MOXA AWK-4131-M12-CT-T Wireless Access Point (items 3 and 4 listed above).

It should be noted that all RF operation around the ISS requires advance approval of the transmitting equipment through the ISS RF authorization process. The payload developer will submit the RF system characteristics and planned operating location information via the JSC Radio Frequency Spectrum Management Homepage (<http://ea.jsc.nasa.gov/webapp/fmdb/login.asp>). The initiation of this review process is recommended after SRR and well ahead of PDR. The payload developer is also responsible for obtaining the required regulatory license grant and submit to the JSC Spectrum Management Office to support the completion of the ISS compatibility review process.

3.3.6 Fields of View (FOV)

The following provides the proposer a series of fish-eye images of the FOV from those Columbus-EPF ExPA/FRAM locations available to hosted NASA external payloads, reference Figure 3.3.6-1. For each payload location, a generic payload volume measuring 46 x 34 x 49 inches was used, reference Figure 3.3.6-2. For each payload volume, the 46-inch dimension runs from ISS Forward to ISS Aft. The 49-inch dimension is the height of the payload measure from, and normal to each EPF FRAM mounting surface. The viewpoint location for these images was the center of each payload face normal to the view direction. These images were generated by the MAGIK Team software tools, at ISS/JSC (see Figures 3.3.6-3 through 3.3.6-6). The solid objects in the FOV remain there at all times while the pink shaded areas in the FOV represent the areas swept by the ISS solar (photovoltaic) arrays as they track the sun. A summary of these fish-eye FOV images is provided in Table 3.3.6-1.

Table 3.3.6-1. Index of Columbus EPF fish-eye FOV images.

Carrier	Location	Optimum Viewing Directions	ISS Aft FOV	ISS Forward FOV	ISS Nadir FOV	ISS Zenith FOV
Columbus	SDN	Ram/Nadir		<input type="checkbox"/>	<input type="checkbox"/>	
Columbus	SDX	Ram/Nadir		<input type="checkbox"/>	<input type="checkbox"/>	
Columbus	SOX	Ram/Zenith		<input type="checkbox"/>		<input type="checkbox"/>
Columbus	SOZ	Ram/Zenith		<input type="checkbox"/>		<input type="checkbox"/>

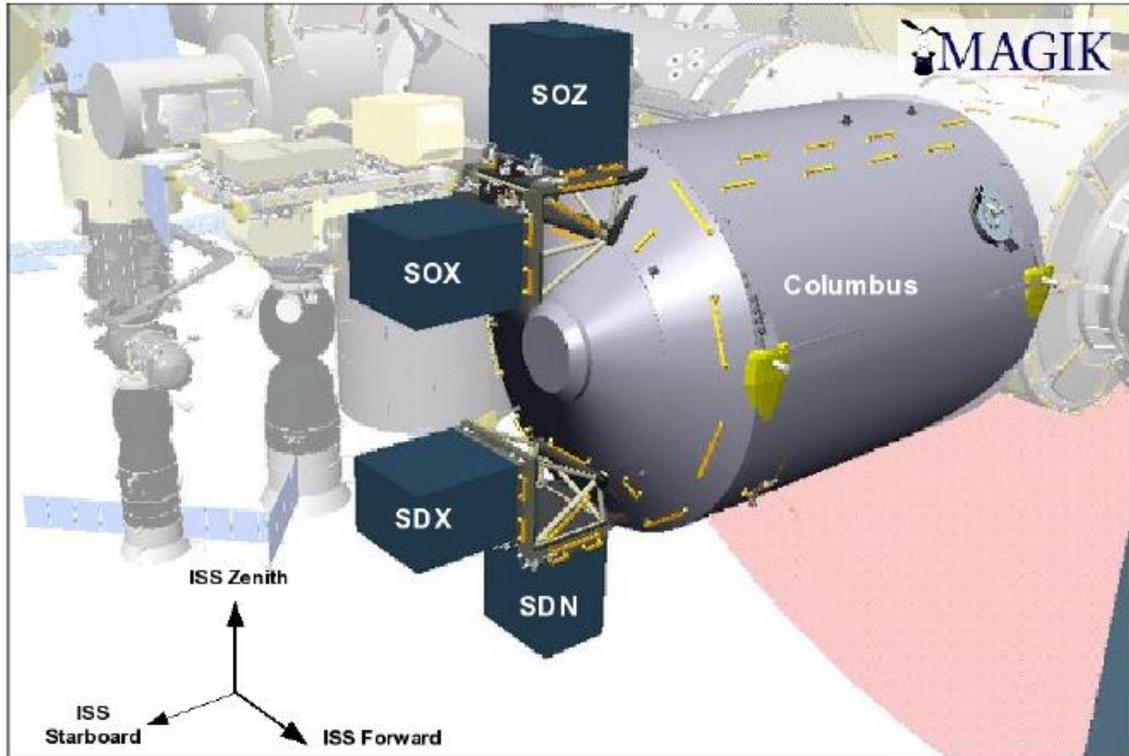


Figure 3.3.6-1. Columbus EPF overview.

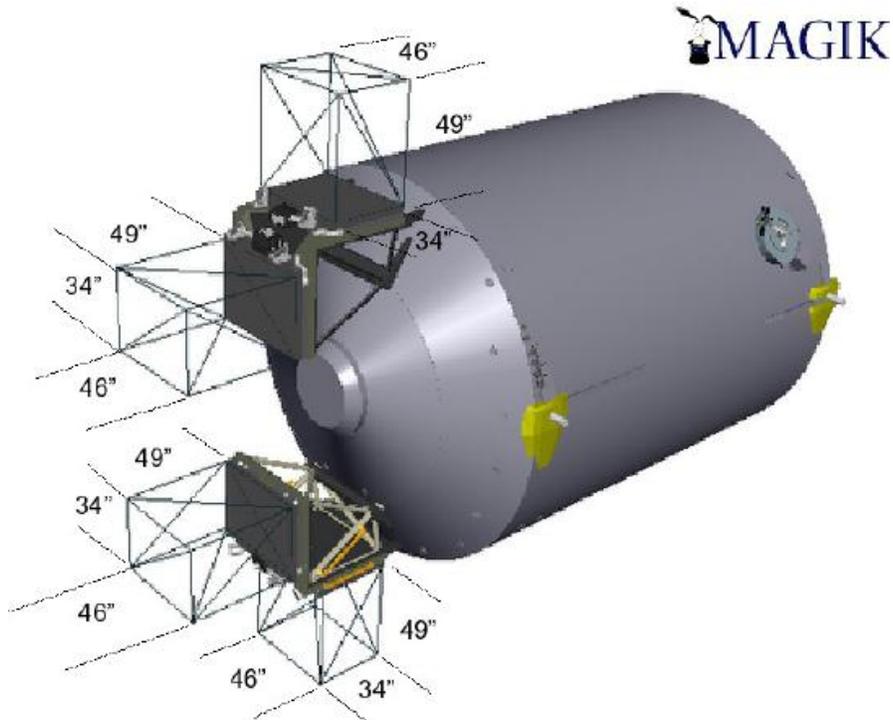


Figure 3.3.6-2. Columbus EPF payload dimensions.

CHECK THE ESP DIVISION WEBSITE AT
<http://espd.gsfc.nasa.gov/isseppg/>
TO VERIFY THAT THIS IS THE CORRECT VERSION PRIOR TO USE.

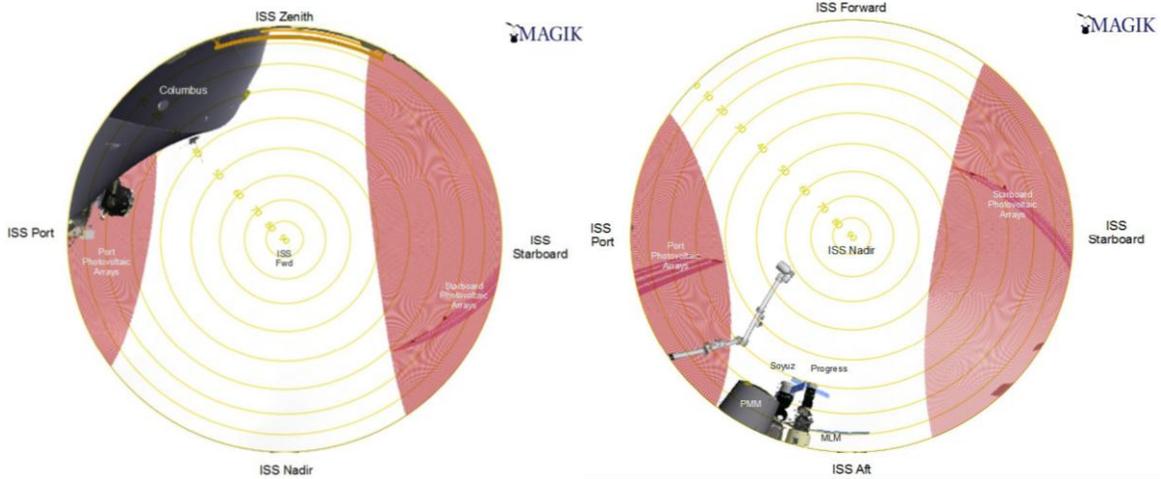


Figure 3.3.6-3. Columbus SDN ExPA P/L accommodation forward (ram) and nadir.

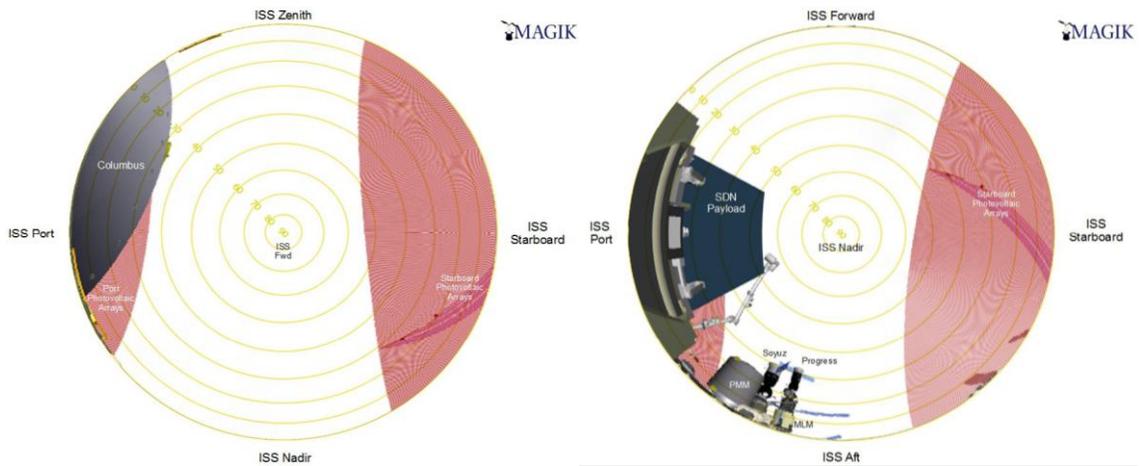


Figure 3.3.6-4. Columbus SDX ExPA P/L accommodation forward (ram) and nadir.

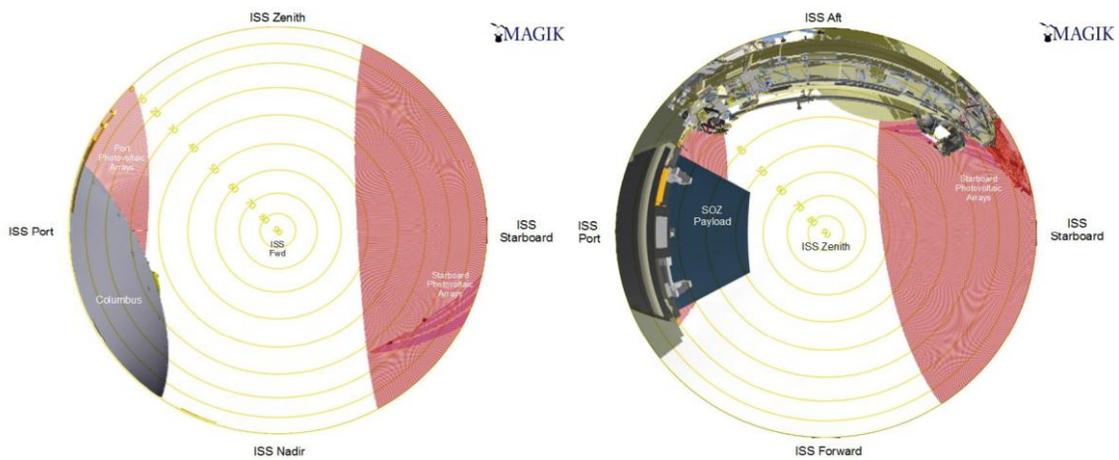


Figure 3.3.6-5. Columbus SOX ExPA P/L accommodation forward (ram) and zenith.

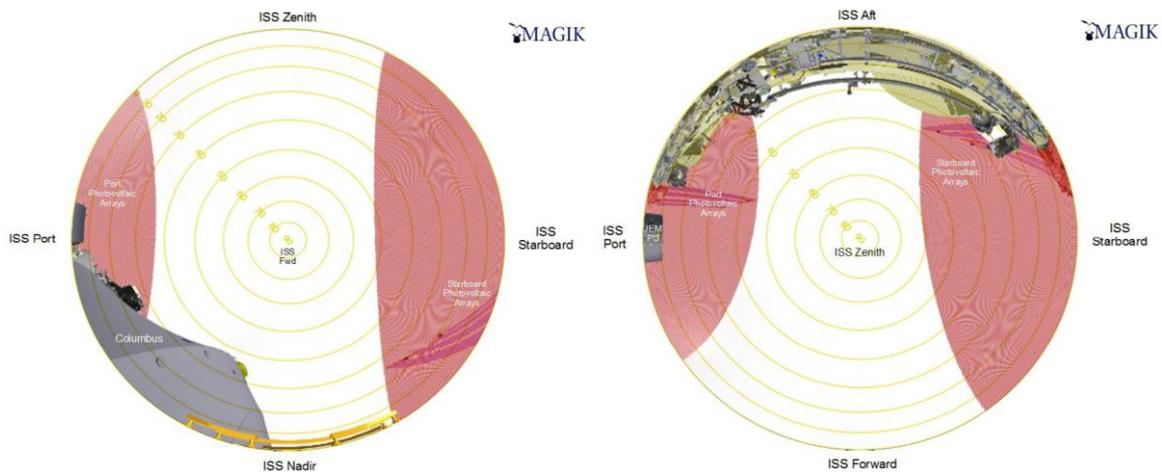


Figure 3.3.6-6. Columbus SOZ ExPA P/L accommodation forward (ram) and zenith.

3.3.7 Microgravity Environment

The mechanical vibrations generated from all of the ISS systems are transmitted throughout the structure and may affect experiments sensitive to acceleration. Attached payloads are required to restrict their mechanical vibrations so as not to disturb sensitive experiments while the ISS is in microgravity mode. To assure that Columbus payloads are able to survive these mechanical vibrations, payloads shall be designed to withstand the on-orbit vibration environment in Table 3.3.7-1. Please note that during nominal operating conditions, additional safety critical loads requirements are levied through Columbus-EPF ICD.

Table 3.3.7-1. On-orbit random vibration environment.

Frequency (Hz)	Level
10–50	0.0005 g ² /Hz
50–100	+3 dB/oct
100–1000	0.001 g ² /Hz
1000–2000	–3 dB/oct
2000	0.0005 g ² /Hz
Composite	1.3 grms
Duration	10 hr/Year

Note: Accelerations must be evaluated as acting one axis at a time in each of three orthogonal axes.

Microgravity Disturbances

Microgravity disturbances induced by external payloads are separated into three categories: quasi-steady for frequencies below 0.01 Hz; vibratory for frequencies between 0.01 Hz and 300 Hz; and transients. These requirements are defined to limit the disturbances to the microgravity environment of other payloads during microgravity mode periods and are not applicable during servicing, crew activities, maintenance, and reconfiguration. The microgravity environment induced by the external payload must not exceed the following levels.

The microgravity disturbances induced by the external payload are not applicable during servicing, crew activities, maintenance, and reconfiguration. The microgravity environment induced by the external payload must not exceed the following levels.

Quasi-Steady Micro-G Requirement

For frequencies below 0.01 Hz, external payload should limit unbalanced translational average impulse to generate <10 lbs (44.8 N-s) within any 10- to 500-second period along any ISS coordinate system vector.

Vibratory Micro-G Requirement

The external payload should limit its interface force induced by its internal disturber to the levels defined in Reference 23, Figure 3.1.4.4.2-1 and Table 3.1.4.4.2-1 for wide- and narrow-band sources. The limit is specified at the external payload to support structure mechanical interface. The requirement is applicable in the 1/3 octave band with coverage over the range of 0.01 Hz up to 355 Hz.

Transient Requirements

A. External payloads should limit force applied to the ISS over any 10-second period to an impulse of not >10 lbs (44.5 N-s).

B. External payloads should limit their peak force applied to the ISS to <1000 lbs (4448 N) for any duration.

Microgravity Environment Profile

Figure 3.3.7-1 depicts a typical microgravity environment profile for the Columbus location on ISS.

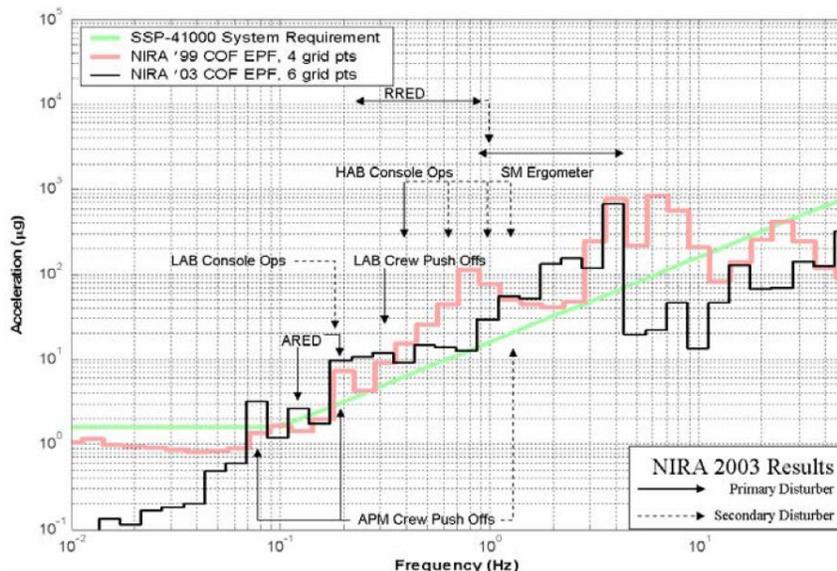


Figure 3.3.7-1. Columbus microgravity environment.

3.3.8 Electromagnetic Compatibility/Electromagnetic Interference (EMC/EMI)

In addition to the information presented in paragraph 3.1.8, the PD should be aware that ESA EMI/EMC requirements are contained in the COL-ESA-RQ-014 (A.D. 2.1.3) document.²³

By agreement between NASA and ESA, a payload can show compliance to either the EMI requirements of COL-ESA-RQ-014 or SSP 30237.²³

Finally, it should be noted that ISS Management Policy Directive (MD-113) applies to all U.S.-funded intentional transmitting and receiving systems for operation on the ISS. This policy provides the authority and requirement to conduct technical reviews in ensuring RF compatibility of the proposed RF system/usage with existing ISS RF operation.

Regarding IP-managed RF payloads for the ISS, the process for coordination of frequency selection and compatibility assurance will be worked through the JSC Spectrum Manager (designated as the ISS Spectrum Manager) as documented in SSP 50423.¹³ Additionally, regulatory filing and approval for space operation of an IP-managed RF payloads is the responsibility of the IP, consistent with the applicable domestic regulation and process of the IP. The International filing documentation, at the minimum of API stage, is to be provided to the ISS Spectrum Manager for final validation of ISS onboard compatibility certification. Upon this validation, the payload will be issued an ISS RF Authorization document for Program ICD and IRD requirements closure.

3.4 Additional Factors Common to All ISS External Sites

The following additional factors that a proposer should consider for their proposal are outlined in the following section. These factors are virtually identical for all three external payload locations. Therefore, they are presented in this section, and do NOT appear in each in Sections 3.1, 3.2, and 3.3.

3.4.1 Attitude and Pointing

External payloads should be designed, and remain safe, while accommodating the ISS attitude and pointing environment described in Table 3.4.1-1.

Table 3.4.1-1. ISS attitude and pointing environment.

Attitude	Duration	Variation
+XVV Torque Equilibrium Attitude (TEA)	95% of Flight Time	Any combination of +/-15° Roll (about X axis) +/-15° Yaw (about Z axis) +15 to -20° Pitch (about Y axis)
-XVV Z-Nadir (LVLH)	< 5% of Flight Time	Any combination of +/- 15° Roll +165° to +195° Yaw +15° to -20° Pitch
+ZVV -X-Nadir (LVLH)		Any combination of +/- 15° Roll +/- 15° Yaw +75° to 105° Pitch
-ZVV -X-Nadir (LVLH)		Any combination of +/- 15° Roll +165 to +195° Yaw +75° to +105° Pitch
+YVV Z-Nadir (LVLH)		Any combination of +/- 15° Roll -110° to -80° Yaw +15° to -20° Pitch

-YVV Z-Nadir (LVLH)		Any combination of +/- 15° Roll +75° to +105° Yaw +15° to -20° Pitch
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Notes:

The attitude variations include variations in the TEA as well as variations in the ISS attitude from the TEA attitude. XYZ axes refer to the ISS coordinate system orientation

Pointing Accuracy

Limitations exist for the degree of pointing accuracy on any payload attachment site. For example, the S3 site, which has the best ISS historical pointing accuracy data, has a fixed alignment error of 0.2 to 0.5°/axis, 3σ. There is also a time-varying error (with a period of the order of an orbit) of ±0.08° for roll and pitch, and +0.23/-0.10° in yaw due to thermal bending of the truss. There is also a random error of 0.001 to 0.02°/axis, 3σ in position knowledge.

Jitter

On ISS, the proposer should be aware that there is a time-varying error (with a period of the order of an orbit) of ±0.08° for roll and pitch, and +0.23/-0.10° in yaw due to thermal bending of the truss. There is also a random error of 0.001 to 0.02°/axis, 3σ in position knowledge.

A depiction of the guidance navigation and control (GN&C) performance regarding jitter is given in Table 3.4.1-2. Please note that the pointing jitter due to microgravity disturbances is estimated to be on the order of 0.03 degrees in quiescent mode (no thruster firings, dockings, EVA or robotics operations, etc.).

Table 3.4.1-2. GN&C performance regarding jitter.

Parameter	Requirement	Typical/Predicted Performance
Attitude rate: non micro gravity	±0.02°/sec/axis	±0.005°/sec/axis
Attitude knowledge at navigation base (S0 truss)	±0.5°/axis 3σ	<0.25°/axis 3σ
Attitude knowledge on the non-articulated portion of the ISS (incl. JEM-EF)	3.0°/axis 3σ	<1.0°/axis 3σ

3.4.2 Micrometeoroids/Orbital Debris

Impacts from micrometeoroids may cause permanently degraded performance or damage to the hosted payload instrument. This guidance provides estimates of the micrometeoroid particle size and associated flux in LEO at an altitude of 400 km. The data comes from the Grün flux model assuming a micrometeoroid mean velocity of 20 km/s and a constant average particle density of 2.5 g/cm³.

Micrometeoroid and artificial space debris flux guidelines are provided separately due to the stability of micrometeoroid flux over time, compared to the increase of artificial space debris. Figure 3.4.2-1 contains the micrometeoroid and space debris flux.

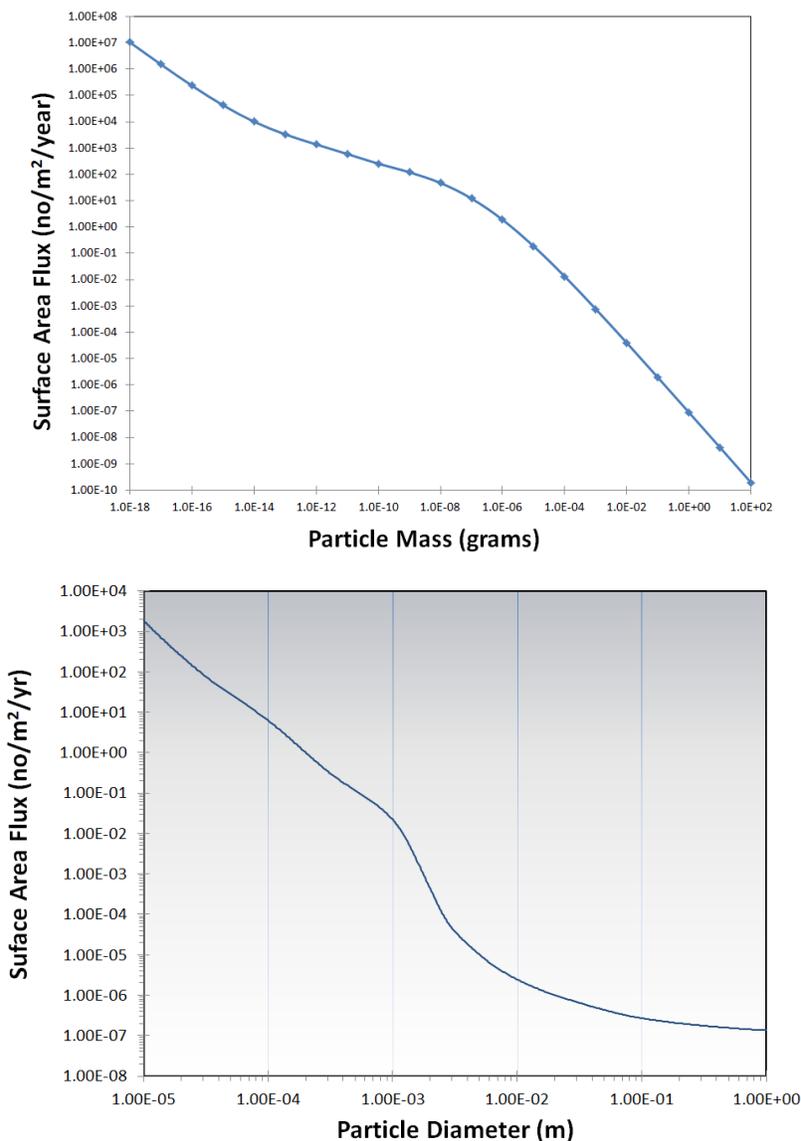


Figure 3.4.2-1. Artificial space debris environment.

Impacts from artificial space debris may permanently degrade performance or damage the Instrument. This guidance estimates the artificial space debris flux and impact velocities an Instrument can expect to experience during the Calendar Year 2014 epoch. Expected artificial space debris flux increases over time as more hardware is launched into orbit. This analysis is for an altitude of 400 km and an orbital inclination of 51.65 degrees.

3.4.3 Magnetic Field and Gravitational Field

The natural on-orbit electromagnetic field and gravitational field environments to which ISS payloads will be exposed are defined in Space Station Program Natural Environment Definition For Design - International Space Station Alpha, SSP 30425²⁶. However, these forces will not be a major factor to the proposer, in all probability. Therefore, no further data are presented in this Guide.

3.4.4 Atomic Oxygen

The average Atomic Oxygen flux of 5.0×10^{21} atoms/cm²/year is expected for the external payload locations. Surfaces exposed 30 days or less will be exposed to an average of 4.4×10^{19} atoms/cm²/day.

3.4.5 On-orbit Minimum Pressure Environment

Payload proposers should plan for their externally integrated and mounted payloads to be able to survive an on-orbit minimum pressure environment of 1.0×10^{-7} Torr (1.333×10^{-5} Pa).

3.4.6 Space Sink Temperature

External payloads will be exposed to and must be compatible with a space sink temperature of 3 K.

3.4.7 Electromagnetic Radiation (EMR)

Electromagnetic noise sources of significance at the ISS extend from direct current to X-ray. Only natural and remote human-made EMR sources are considered here. The categories of noise producers are as follows: Galactic, Solar, Near-Earth natural plasma, and human-made radio noise.

The highest power densities expected to be irradiating the ISS are from the solar radiation in the ultraviolet and visible portions of the electromagnetic spectrum. The ultraviolet radiation can damage materials exposed to it. Space Station External Contamination Control Requirements, SSP 30426, paragraph 7.2, describes the degree of exposure of the payloads to the solar ultraviolet radiation environment.²⁷ Other effects of EMR to be considered include radio noise and field strengths from the natural sources at the ISS. Field strengths produced from quasi-static field structures in the plasma have typical values around 25 mV/m, but can be larger. These values generally occur at latitudes greater than 50°.

3.4.8 Contamination

The ISS has been designed and built with strict contamination control requirements to provide low levels of induced contamination on external payload assets. The ISS contamination environment is one to two orders of magnitude (10 to 100 times) better than the environment provided by the Mir space station. This has been demonstrated by both flight experiments and measurements made on returned hardware.

In order to fly on ISS, payloads must demonstrate compliance with ISS contamination control requirements. The System Specification for the International Space Station, SSP 41000,²⁸ contains the top-level requirements (ISS system level requirements). SSP 41000 Paragraph 3.2.6.1.3 (which calls on SSP 30426, Space Station Contamination Control Requirements, Sections 3.4 and 3.5)²⁷ specify the “design-to” contamination environment of 130 Å/year on contamination sensitive surfaces. Payloads should be designed to perform within this contamination environment.

SSP 41000 3.3.10.3 (which calls on SSP 30426 Sections 3.4, 3.5 and 3.6)²⁷ specify the external contamination releases requirement which limit contaminant deposition to 130 Å/year on contamination sensitive surfaces, from all sources of contamination on the vehicle combined (ISS elements, visiting vehicles and payloads). Hence, the contaminant deposition contribution from a new payload, when combined with contributions from all sources (ISS elements, visiting vehicles and payloads), must not exceed 130 Å/year on contamination sensitive surfaces of ISS, visiting vehicles and other external payloads. This requirement is verified by analysis by

the ISS Program. The payload is responsible for preparing and submitting a characterization of contaminant sources on the payload as required input to the system level analyses conducted by the ISS Program.

In addition to compliance with the ISS system level contaminant release requirements, payloads are required to comply with specific contaminant releases allocations governing integration at specific sites (U.S. Segment attached payload sites, ELCs], the Columbus-EPF and the JEM-EF). As in the case of system level verification, the payload is responsible for preparing and submitting a characterization of contaminant sources on the payload. The intra-site integration analyses are conducted by the site integrator.

Payloads designed for deployment on the U.S. Segment payload sites must comply with contamination requirements detailed in SSP 57003,⁴ SSP 57003-ELC (for ELC-based payloads),³ Attached Payload Hardware Interface Control Document Template, SSP 57004,²⁹ Attached Payload Interface Control Document Express Logistics Carrier (ELC) Cargo Interface Control Document Template, SSP 57004-ELC (for ELC-based payloads),³⁰ and Payload Verification Program Plan, SSP 57011³¹.

Requirements governing integration and verification of payloads on the European Columbus Module are specified in the Columbus External Payloads Interface Requirements Document (COL-RIBRE-SPE-0165)²³

Payloads flying on the JEM-EF are governed by the Exposed Facility/Payload Standard Interface Control Document (JPAH Vol. 3, NASDA-ESPC-2563).³² JEM-EF requirements specify compatibility with the ISS system level requirements but do not make specific sub-allocations for payload-to-payload induced contamination level within the JEM-EF

As previously stated, payloads are required to deliver a characterization of contamination sources on their payloads to support integration and requirements verification activities conducted by the Program. Characterization of payload contamination sources include definition of vacuum exposed materials (all non-metallic materials outside of a pressurized or hermetically sealed environment), vacuum venting (liquids and gases), leakage, thrusters, and sources of particulate releases. Payloads are also requested to identify payload contamination sensitive surfaces and provide operating temperature predictions (contaminant deposition is highly dependent on the operating temperature of receiver surfaces). This data is used for tracking induced contamination on the payload from the vehicle (ISS), visiting vehicles, and other payloads.

Payload data deliveries covering for non-metallic vacuum exposed materials must include identification of materials, location of material application on payload (if application is localized), vacuum exposed surface area, nominal operating temperature range, and outgassing rate data (ASTM E1559). The preferred format for the definition of operating temperature data for payload materials is one that specifies the percentage of time spent under 30 °C, between 30 °C and 60 °C, and between 60 °C and the maximum operating temperature. Outgassing rate test data for the ISS Program is based on Method B of the ASTM E1559 standard, with a minimum test duration of 144 hours and four Thermally controlled Quartz Crystal Microbalances (TQCMs) used for condensable outgassing rate measurements (TQCMs at 80 K, -40 °C, -10 °C and +25 °C).

The Payload is required to deliver a preliminary characterization of contamination sources verification by 24 months prior to launch (L-24 months). The preliminary data delivery at L-24 months is used to identify potential issues and allow for corrective action with minimal impacts

to cost and schedule of payload development and integration. Updates to the preliminary data delivery are required if significant sources of contamination (or significant changes) are introduced prior to final data delivery.

Final verification data submittal from the payload is required 7.5 months prior to launch (L-7.5 months). The final analysis report supporting verification (conducted by the Program) is issued by L-3 months.

From experience from past and existing payloads, the most significant source of contamination from payloads is the use of high outgassing materials (non-metallics). It is critical to control and minimize the use of silicones (e.g., RTV and CV series silicones, silicone-based thermal control paints, cable insulation, cable clamps, connectors, etc.) and hydrocarbon based materials (e.g., Velcro®, epoxies, conformal coatings, Tefzel® cable insulation and encapsulation). Control of outgassing venting paths can be effective in mitigating contamination induced by the payload and the Program can assist with determine preferential venting paths.

As part of ISS payload integration activities, contamination forecast maps are generated for U.S. attached payload sites to support payload feasibility, topology and placement studies. Forecast maps are also used to identify potential integration issues of the ELC payload complement. These predictions are available to Payload Proposers upon request to the Program.

The ELC based attached payload sites are shown in Figure 3.4.8-1. Each ELC contains 2 payload sites in addition to a complement of ISS spares designated as ORUs. The ELC based payload sites are shown in orange in Figure 3.4.8-1. ORUs are shown in teal color (same as the ELC frame). The payload sites on ELCs 1-3 are located on opposing sides of the carrier ELC. ELC 4 offers two payload sites on the same side (ISS inboard side). Figure 3.4.8-1 also shows the ISS coordinate system for reference purposes.

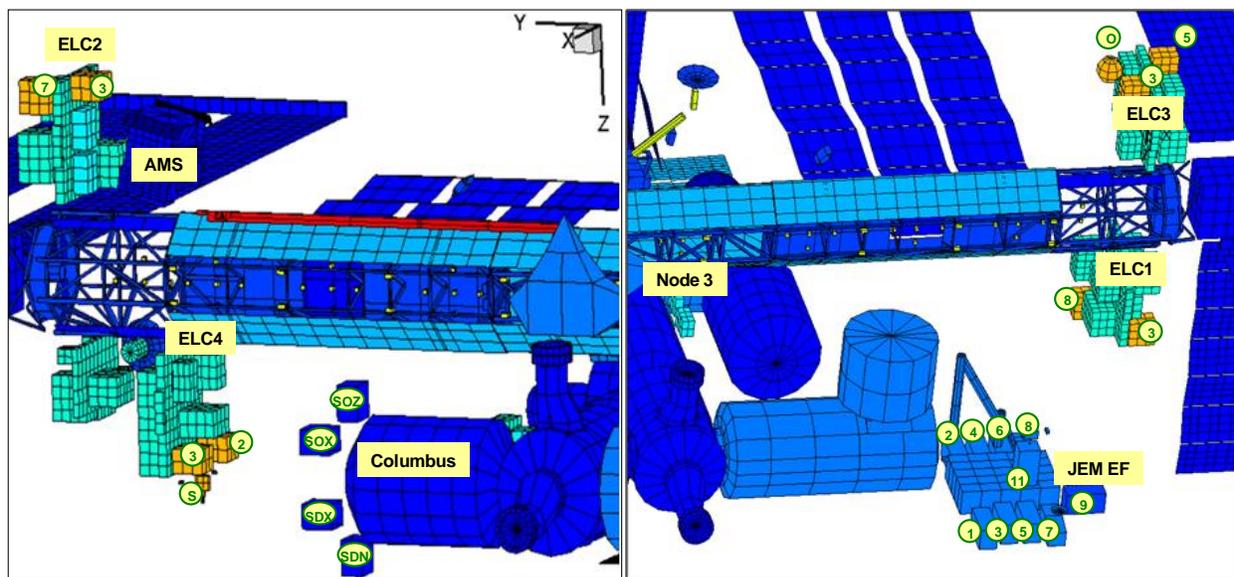


Figure 3.4.8-1. Payload mapping sites (ELCs 1-4).

Contamination forecast maps for 2015-2017 are summarized in Tables 3.4.8-1 through 3.4.8-3. These predicted contamination levels were generated for generic payload envelopes, not actual payload geometries. These figures are meant to provide initial guidance for choosing payload

locations. The Program will work with each payload to obtain more detailed predictions as their designs mature.

Contaminant deposition predictions are provided for 3 receiver operating temperatures (+25 °C, -10 °C and -40 °C). The blue cells represent payload envelope surfaces expected to receive less than 130 Å/year. Gray cells represent the payload interface surfaces with the ELC (the mounting surface). Red cells represent surfaces expected to receive greater than 130 Å/year.

The great majority of red cells are associated with potential contamination from an adjacent ORU on the ELC) and generally only for payload operating temperatures below -10 °C. Further, payload surfaces facing an adjacent ORU have an obstructed FOV (the ORU obstructs the view from that side) and are unlikely to contain contamination sensitive surfaces.

In the process of generating the contamination forecast maps, if a payload surface is identified as receiving deposition levels in excess of 130 Å/year, and is not associated with an adjacent ORU, that surface is identified for forward work (issue resolution).

Table 3.4.8-1. 2015 Contamination forecast map summary.

Payload Site	Contaminant Deposition on Payload Envelope Surfaces (Å/year)																	
	+X _{ISS}			-X _{ISS}			+Y _{ISS}			-Y _{ISS}			+Z _{ISS}			-Z _{ISS}		
	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C
ELC1 - Site 3	9.8	29	67	36	51	74				2.1	3.3	7.3	12	14	18	21	41	98
ELC1 - Site 8	27	64	107	47	56	82	76	110	190				16	19	24	40	67	147
ELC2 - Site 3	6.7	17	33	43	71	86				24	41	76	24	57	98	0.1	0.2	0.3
ELC2 - Site 7	1.0	4.7	16	24	49	65	1.3	3.4	13				22	45	96	0.2	0.3	0.3
ELC3 - Site 3	1.9	16	151	49	69	114	22	38	110				49	99	324	0	0.1	0.1
ELC3 - Site 5	51	104	249	18	19	23				1.1	3.0	11	35	69	229	0.1	0.1	0.1
ELC4 - Site 2	31	57	206	57	68	92				72	102	168	17	21	28	37	58	209
ELC4 - Site 3	2.6	15	45	77	101	242				60	92	162	11.7	14	20	36	203	370

Table 3.4.8-2. 2016 contamination forecast map summary.

Payload Site	Contaminant Deposition on Payload Envelope Surfaces (Å/year)																	
	+X _{ISS}			-X _{ISS}			+Y _{ISS}			-Y _{ISS}			+Z _{ISS}			-Z _{ISS}		
	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C
ELC1 - Site 3	8.8	25	53	30	44	69				2.0	3.1	7.0	8	11	17	19	38	91
ELC1 - Site 8	24	57	92	34	47	79	55	93	174				9	13	22	37	62	135
ELC2 - Site 3	6.0	15	28	40	66	80				22	37	66	22	51	87	0.3	0.4	0.5
ELC2 - Site 7	0.7	3.7	12	22	46	60	1.2	3.2	12				20	42	88	0.4	0.5	0.5
ELC3 - Site 3	1.6	13	89	44	62	103	19	33	87				43	87	245	0	0.1	0.1
ELC3 - Site 5	46	92	220	16	17	22				1.1	2.8	11	31	62	207	0.2	0.2	0.2
ELC4 - Site 2	27	48	174	50	66	101				64	101	171	19	25	37	33	53	190
ELC4 - Site 3	1.2	10	27	67	96	236				52	90	160	12.5	16	24	33	169	317

Table 3.4.8-3. 2017 contamination forecast map summary.

Payload Site	Contaminant Deposition on Payload Envelope Surfaces (Å/year)																	
	+X _{ISS}			-X _{ISS}			+Y _{ISS}			-Y _{ISS}			+Z _{ISS}			-Z _{ISS}		
	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C	25°C	-10°C	-40°C
ELC1 - Site 3	8.0	16	38	29	42	62				1.9	3.0	6.6	9	11	15	18	35	85
ELC1 - Site 8	22	39	69	34	42	59	58	79	131				12	14	20	34	57	125
ELC2 - Site 3	5.7	14	26	37	60	73				20	34	61	21	48	81	0.3	0.4	0.4
ELC2 - Site 7	0.8	3.7	11	20	42	56	1.1	3.0	11				18	39	82	0.3	0.4	0.5
ELC3 - Site 3	1.4	9	70	40	56	94	18	29	76				40	75	208	0	0.1	0.1
ELC3 - Site 5	41	84	198	15	16	19				1.0	2.7	10	29	57	189	0.2	0.2	0.2
ELC4 - Site 2	25	46	162	46	56	77				60	87	141	18	22	29	31	48	174
ELC4 - Site 3	2.0	11	26	62	84	201				49	77	131	11.2	14	19	30	148	282

The contamination forecast maps offer insight into the system level contamination environment for ISS payloads, and support payload placement and topology studies. During the payload integration process, higher fidelity contamination analyses are conducted using actual payload geometries, materials and operating temperature data to support verification and to ensure success of science missions.

Contaminant deposition measurements have been made on returned hardware, and comparisons to analysis predictions have been made to assess performance against expectations. Predicted contamination levels at ISS payload sites are often lower than the system level specification for select surfaces and several contamination sensitive payloads have relied on predicted levels in operational planning. However, the Program recommends designing for performance within the system level specification of 130 Å/year.

Returned materials samples from the Materials International Space Station Experiment (MISSE) 2 flight experiment, which were exposed to the ISS contamination environment for 4 years (from 2001 to 2005), demonstrated low levels of induced contamination from U.S. Segment hardware on ram facing surfaces (wake facing surfaces received contamination from Russian Segment hardware). A comparison of the MISSE 2 predicted and measured contaminant deposition is shown in Table 3.4.8-4.

Table 3.4.8-4. Predicted vs. measured contaminant deposition on MISSE 2.

Experiment	Side	Predicted	Measured
MISSE 2	ram	80 Å	50 Å
	wake	730 Å	500 Å

In 2013, samples returned with the MISSE 8 flight experiment, which was deployed on ELC2 for 2 years, were analyzed with no identification of contaminant deposition. (Shown in Figures 3.4.8-2.)

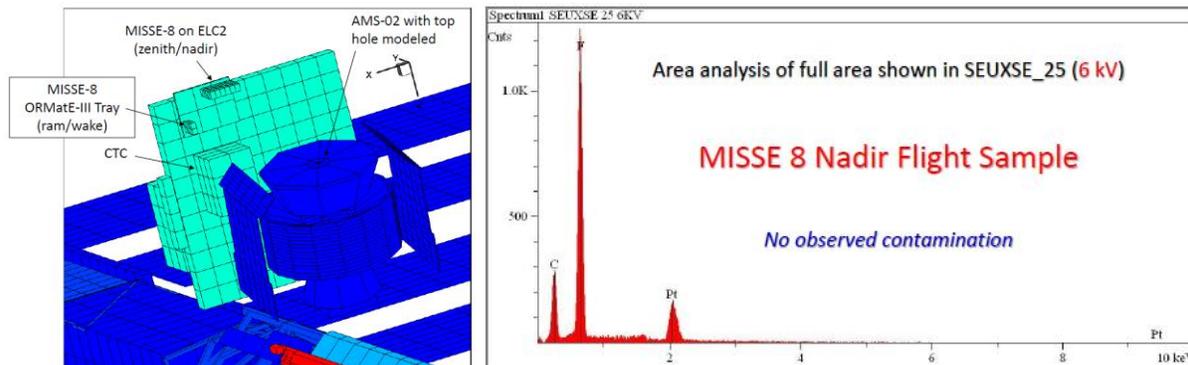


Figure 3.4.8-2. MISSE 8 configuration on ELC2 and sample surface analysis results.

In summary, the ISS offers an exceptionally clean environment to attached payloads. The low levels of induced contamination on the vehicle can support a wide variety of contamination sensitive payloads observing in the visible, infrared, and ultraviolet regions.

3.4.9 Contingency EVA (as required)

For external payloads, only contingency EVA is available, and the PD should not expect contingency EVA to restore mission success. Contingency EVA would normally only be performed to address major issues, such as the case of ISS robotic failure, and other issues concerning vehicle and/or crew safety. In any case, EVA safety requirements must be addressed for your payload design.

3.4.10 End of Mission

Generally, a payload will be removed from its position either shortly before or after arrival of the visiting vehicle that is bringing the replacement payload, according to the ISS manifest. A payload is required to return within the payload's launch envelope. If a malfunction can prevent the payload from returning to within launch envelope by command, the payload is required to have robotic interfaces (hex head bolt(s), micro-square or micro-conical(s), target(s) which will enable an EVR-assisted configuration for disposal of the payload.

Once the payload is configured for disposal and commanded off (and relying on contingency heaters), the removal process begins. Removal is the reverse of the installation procedure, and ends with the payload configured for disposal in the visiting vehicle. However, the order is typically to remove the old payload, set it in a temporary parking position, retrieve the new payload from the visiting vehicle and install it, then retrieve the old payload from its temporary position and configure the payload for disposal in the visiting vehicle. After reinstallation of the payload into the visiting vehicle, the vehicle will reenter the atmosphere, and burn up, based on the current return vehicle capabilities.

4.0 Integration and Operations

The integration and operation of external payloads on the ISS is a vital area that must be evaluated thoroughly by the proposer. Section 4.1 contains an overview of the integration template, engineering and physical integration processes, and Section 4.2 outlines the Operations process. In addition, the safety process is contained in Section 6.0 of the Guide.

The engineering integration and operations areas, for both the ISS and launch vehicle/visiting vehicle, is often underestimated by proposers, but will require a substantial amount of effort during the implementation phase of payload development, and will require a substantial amount of resources. ISS document OZ-10-056,³³ *Payload Developers and Principal Investigators Payload Planning, Integration and Operations Primer*, is an excellent source for in-depth understanding of the manifesting, integration, and operations processes for the ISS.

4.1 Integration

4.1.1 Integration Template (Reference Milestones)

An overview of the ISS critical integration milestones, along with ancillary data, is shown in Figure 4.1.1-1. It is NOT the intent of this Guide to provide detailed explanations of each milestone contained in the Template, nor do you have to familiarize yourself with every nuance contained in the Template. However, it has been incorporated so the proposer may adequately assess the number of data submittals, activities, reviews, and corresponding tasks associated with this critical process, as described in Figure 4.1.1-1.

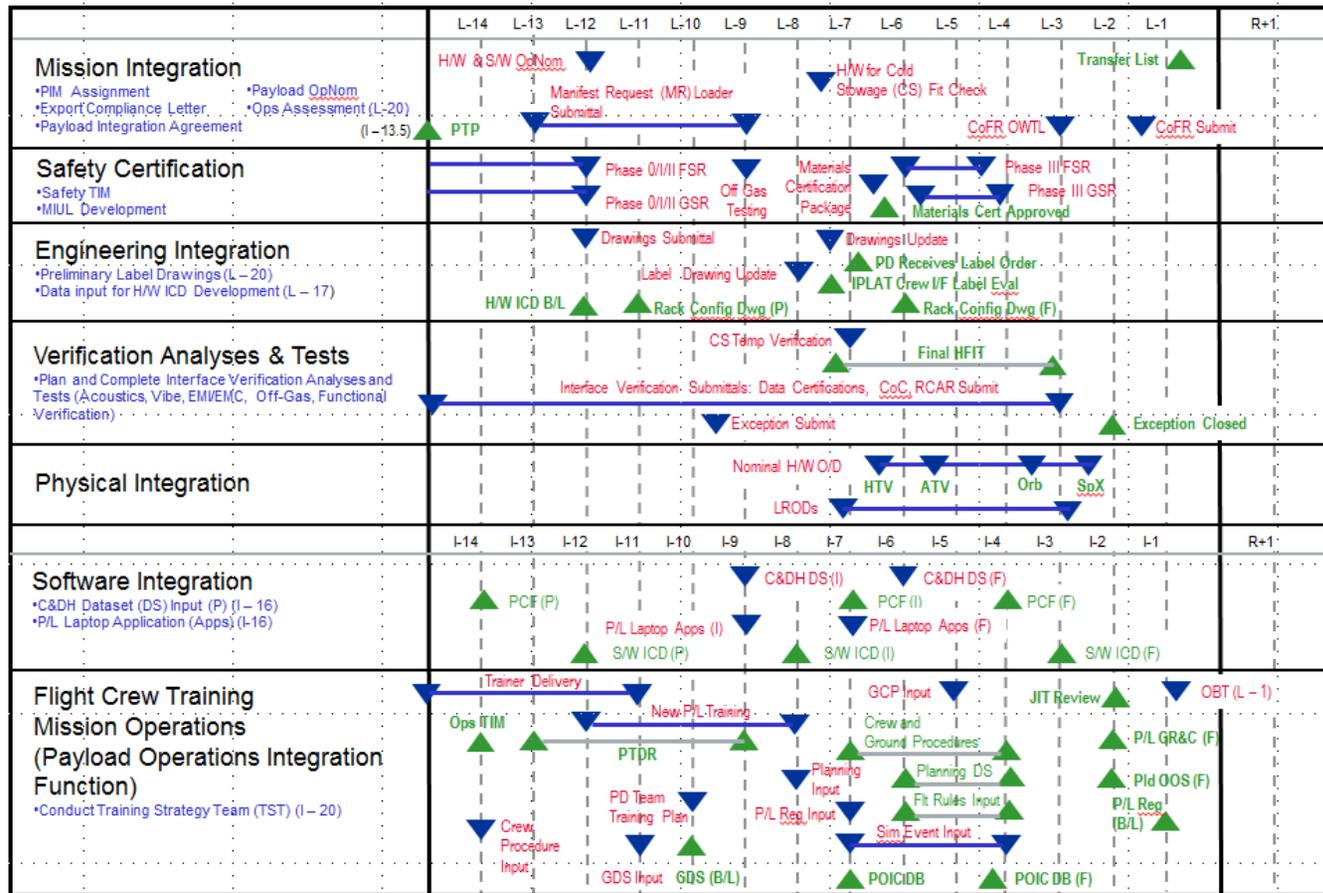
The documents listed in this section will prove crucial to the successful integration of your payload onto the ISS. However, it is the responsibility of the PD to design the payload to conform to the appropriate set of interface requirements specified by the ISS program.

As indicated on the integration template, the PD will be responsible for developing numerous data inputs. For example, inputs are required from the PD the Integrated PDR and Integrated Critical Design Review (CDR), which require significant resources.

The PIM initializes customer interfacing and the development of a specific Payload Integration Agreement (Standard Payload Integration Agreement for Attached Payloads, SSP 57061),³⁴ the payload-unique ICD, a tailored schedule of specific events for your payload, and delivery of payload data products that will be used for safety and verification for flight on the ISS.

Integration Template

The Payload Integration Template as defined involves the integration of a payload with both the ISS vehicle and ground operations. The template is a flight (or increment) driven template in that a payload is assigned a flight on an increment and the PD works the associated milestones to meet that flight/increment's template. Additional template information can be found in SSP 57004-ELC³⁰.



▼ Payload Milestone
 ▲ Payload Integration Team Milestone
 Reference: SSP 57057 - Payload Integration Template

Figure 4.1.1-1. ISS integration template.

4.1.2 Engineering Integration Process

This section primarily outlines the ISS engineering integration process. The integration process of each launch vehicle/visiting vehicle is somewhat dependent on which vehicle is eventually selected, so the launch vehicle/visiting vehicle section of the Guide, Section 5.0, provides a summary of a specific launch vehicle integration process for the existing launch vehicle. Finally, the physical integration, while not specifically part of the EI process, is, nevertheless, briefly covered in the last portion of Section 4.1.

4.1.2.1 ISS Interface Requirements (Integrated Truss Assembly Using the ELC)

The Attached Payload Interface Requirements Document (IRD)-EXPRESS Logistics Carrier, SSP 57003-ELC,³ is the principle source of interface design requirements for attached payloads flying on the ELC (ITA sites). In order for an external payload to be certified for integration into the ISS external sites, the IRD must be complied with.

The IRD levies design interface and verification requirements on ISS external payloads. These requirements are allocated to an external payload payload through the applicability matrix in the unique payload Interface Control Document (ICD). The unique payload ICD defines and controls the design of the interfaces between the ISS and the payload, including unique interfaces. Therefore, for external payloads flying on the ELC, the applicable ISS interface requirements document is the SSP 57003-ELC IRD3.³ All requirements defined in the IRD, including physical, functional, and environmental design, and interface compatibility, apply to on-orbit phases of operation. On-orbit requirements apply to all External Payloads of the ISS located and mounted on the ITA. Examples of interface requirements elements contained in the SSP 57003-ELC IRD are:

- Structural/Mechanical and Microgravity
- Electrical
- C&DH
- Passive Thermal Control
- Environment
- Materials and Parts
- EVR
- Maintainability and Maintenance
- Nameplates and Product Marking
- EVA

In addition, the PD will also develop an ISS Interface Software ICD, and a Payload-to-Launch Vehicle ICD.

4.1.2.2 PD-Developed Interface Control Documents (ICDs)

As described in Section 4.1.2.1, the ELC IRD (for ELC proposers) levies design, interface, and verification requirements on ISS external payloads. These requirements are imposed on a payload through an applicability matrix in the PD developed, payload-unique ICDs. The PD must create 3 payload-unique ICDs for flying on the ISS. The creation of the ICDs is a significant effort imposed on the PD, and should not be underestimated. The PD will develop a hardware ICD and a software ICD, and a Launch Vehicle ICD, which details the requirements implementation methods.

The payload-unique ICDs define and control the design of the interfaces between the ISS and the payload, and the Launch Vehicle and the payload, and serves to establish commonality with respect to interface design, analytical approaches, models, test methods and tools, technical data and definitions for integrated analysis. The Payload-to-Launch Vehicle ICD is covered in Section 5.0.

The engineering integration process includes assigning an ISS ICD engineer to work with the PD on ICD development. The ICD documents the payload interfaces to ISS and identifies the applicable requirements from the ISS that the payload has to meet, the verification method, due date, and verification data for each applicable requirement. The payload-unique ICD documents the implementation of the IRD requirements and ensures that they remain within the interface design parameters. ANY exception to the IRDs must be documented in the payload-unique ICDs. The ICDs will document the specific requirement violated, a description of the existing condition, and a rationale for acceptance. Finally, a crucial component of the PD developed ICDs is the verification matrix.

4.1.2.3 Verification

A major task creation of the ICD is the creation and completion of the verification matrix. Subsequent verification data submittals to the ISS team will utilize the verification methods contained in the ICDs. Verification methods will consist of one or more of the following:

- Analysis
- Testing
- Inspection
- Demonstration.

Verification submittals based on the PD-defined verification matrix are a critical step to flying on the ISS, and requires a substantial amount of effort by the PD. Each item contained in the ICD has a corresponding requirement in the IRD and must be verified by one of the methods listed above. This is a major element to successful closure of the engineering integration tasks, and a vital point for flying on ISS. Please note that the ICD is usually at a draft state for the payload PDR and is baselined at the payload's CDR.

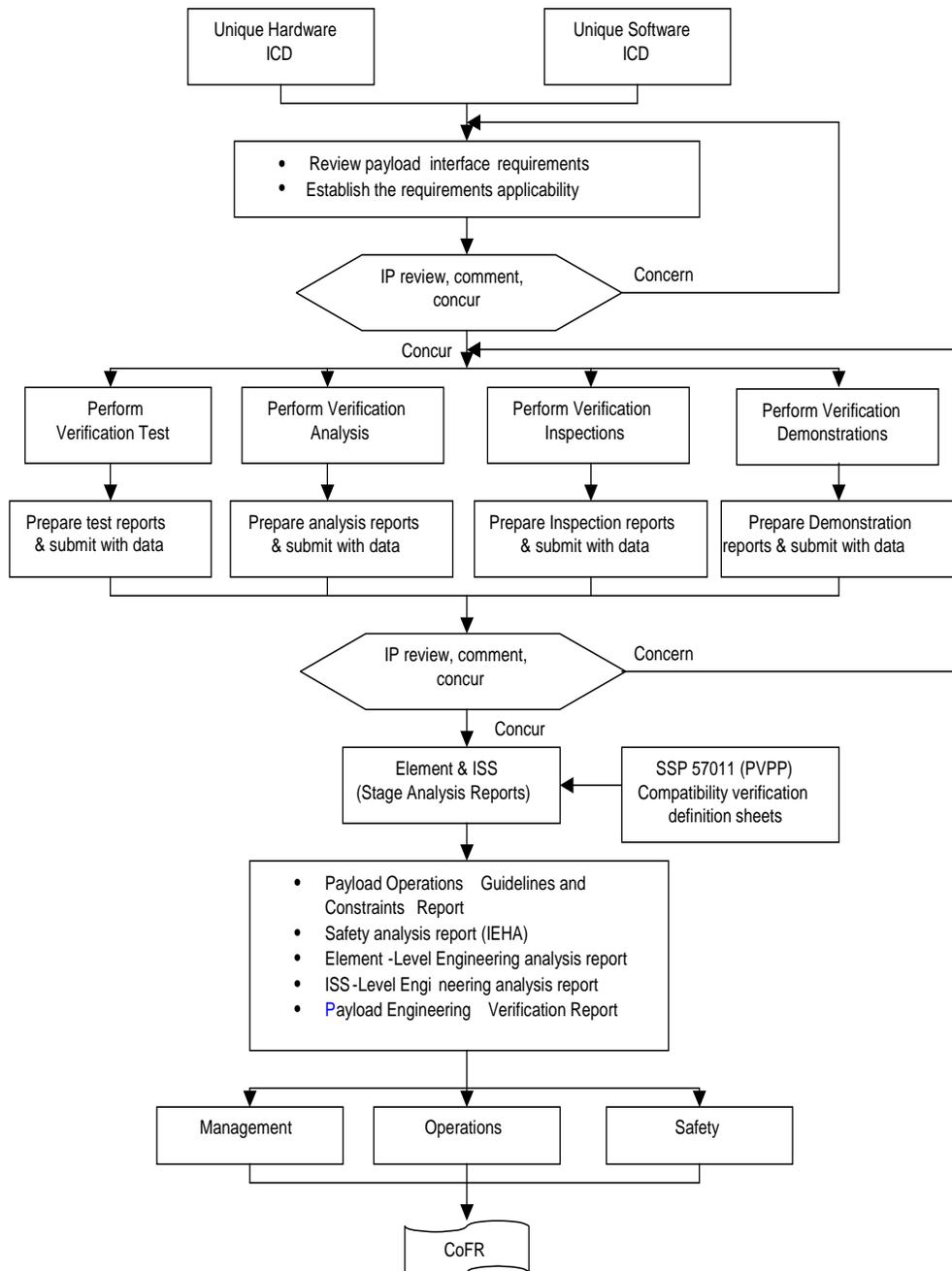
You will need to use the verification methods outlined in the ICD to provide evidence of verification completion to the ISS program. The flow chart, Figure 4.1.2.3-1, provides a summary of the verification process, and is applicable to all ISS external payloads.

During the verification process, the PD will develop a Verification Tracking Log (VTL). The VTL is used for tracking and closing out your payload ICD, system hardware, and safety requirements prior to launch.

Payload verification data are collected from the PD and reviewed by the ISS Payload Engineering Integration team initially. After the verification data have been approved, they create a verification memo for payloads launching and returning on a specific increment. The ISS will conduct numerous reviews of your verification activities, including conducting two extensive paperwork audits of your hardware configuration to your payload ICD and applicable ISS payload requirements interfaces. One is a functional configuration audit and the second is

a physical conformance audit. Together, they verify all of your payload design, manufacturing, assembly, and testing configurations through design, build, and test documentation.

As you can see, the CoFR is the culmination of the verification process. This critical element states that the PD has certified that their payload has passed all verification steps, including hardware and software testing, and is ready for flight. The PD must participate in the CoFR process, which consists of a series of detailed “readiness” reviews. These reviews are performed by NASA, the international partners, and NASA contractors.



NOTE: There are numerous steps included in the total verification process, including performing the actual verification, preparing reports and data submittals, and review of the PDs verification submittals by various ISS organizations, such as operations, safety, and ISS management.

Figure 4.1.2.3-1. Summary of verification process.

4.1.4 Verification Interface Testing (Part of Verification Process)

Most of the PD-developed integration products provided to the ISS are analytical in nature; however, there is a significant verification-testing component to the verification process. When a payload must interface with other facilities or payloads on the ISS, integration and interface testing is performed prior to launch. KSC and MSFC provide the infrastructure for this testing.

Testing can be categorized into payload-unique development testing and ISS verification and interface testing. ISS provides hardware to accomplish both testing phases. The ISS-provided simulators for testing are described in more detail in Section 7.0.

The determination of flight readiness of a payload is composed of: (1) the certification that each hardware and software element meets specific ISS requirements, and (2) the certification of these elements as an integrated payload. The final, integrated verification is accomplished by testing the payload with ground equipment that is equivalent to the facility on-board the ISS.

Testing is also used to perform final functional ISS interface testing which ensures compatibility between ISS and the payload and joint operations with other payloads, as contained in the applicable ICDs. External ISS payloads testing will generally occur at KSC. This testing will determine the readiness of the payload for flight on the ISS.

The overall generic flow for the testing process is shown in Figure 4.1.7-1.

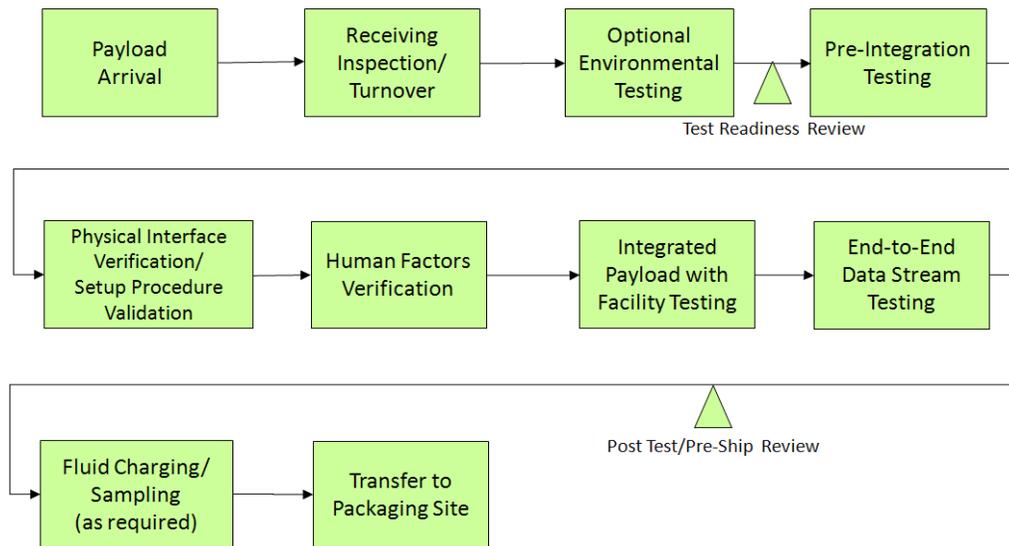


Figure 4.1.7-1. Testing flow.

Along with the testing process, the PD will have numerous prelaunch activities, including off-line preparations; payload integration; test and checkout; and Interface Verification Tests (IVT). These tasks are generally accomplished at KSC. An Integration Readiness Review/Test Readiness Review will be held prior to integrating the payload into the ISS facility for testing.

NOTE: Please remember that it is the responsibility of the PD to conduct payload-unique software testing during the payload development life cycle phase. Software testing can be accomplished using ISS-provided simulators.

Most integrated payload testing for the ELC is performed via the ELC Simulator, and the Active Common Attachment System Simulator, located at KSC. This hardware is configured and checked out by the ISS to support payload verification testing. Environmental testing such as vibration, EMI, off-gassing, etc. should be performed prior to arrival at the test site. If desired, the PD can make special arrangements to perform environmental testing using KSC or MSFC test facilities.

During the final integration phase, the PD's flight hardware/software is turned over to KSC for testing with ISS flight-equivalent hardware and software. This process encompasses the final verification and interface testing for flight on the ISS. After turnover via formal shipping document (e.g., DD1149), KSC has custodial responsibility for the PD flight hardware. These activities include hardware testing, integration, fluid servicing, inspections, fit-checks, etc. as agreed to by the PD and KSC, and uses ISS-provided equipment and simulators. However, the PD must be prepared to support this effort.

End-to-end data stream testing is also performed to ensure data generated from the payload operating in/with the ISS will be properly displayed on the computer screens at MSFC and the PD tele-science centers. KSC and MSFC have implemented the capability to test and verify high-rate data streams using simulators of the ISS external accommodations. These simulators are discussed in Section 7.0 of the Guide.

4.1.4.1 Physical Integration Facilities

The primary area at KSC that the PD may utilize for physical integration is the Space Station Processing Facility (SSPF). External payload ground handling, processing, and ground transportation requirements are specified in KPLHB 50001³⁵. Laboratories in the SSPF building are class 300,000 clean rooms. All labs are equipped with sinks, hot and cold water, cabinets, vacuum system, compressed air, and some have one-ton crane support. The PD should be prepared to support SSPF processing for at least a week, on site.

If a PD requires non-standard services at KSC, they will be documented in the Payload Interface Agreement (PIA). The PD will provide payload-unique technical requirement inputs for these non-standard requirements that are to be levied on KSC. Non-standard services may require PD funding to implement.

4.1.4.2 JEM-EF Engineering Integration Process

The process for flying on the ISS's JEM-EF is virtually identical to flying on the ELC. However, the interface requirements for JEM-EF are contained within NASDA-ESPC-2900A¹⁴. The proposer must utilize this document to determine their specific JEM-EF interface requirements for flying on the ISS. A total of four ICDs will be required for use of the JEM-EF facility:

One ICD will be written using NASDA-ESPC-2900A as the requirements document that covers specific interfaces between the payload and the JEM-EF facility. In addition, another ICD will be required between the ISS/JSC and your payload, as described in Section 4.2.1. Finally, a Software and Launch Vehicle ICD is also required, as outlined in Section 4.2.1.

Interface testing facilities and simulators are provided by the ISS for JEM-EF payloads, such as the Payload Rack Checkout Unit (PRCU), etc. These ISS-provided simulators are described briefly in Section 7.0.

4.1.4.3 Columbus Attached Module Engineering Integration Process

The process for flying on the ISS's Columbus Attached Module is virtually identical to flying on the ISS truss. However, the interface requirements are contained within COL-RIBRE-SPE-0165²⁰. The proposer must utilize this document to determine their specific Columbus interface requirements for flying on the ISS.

Aa total of four ICDs will be required for utilization of the Columbus facility. One ESA hardware ICD will be written as the requirements document that covers specific interfaces between the payload and the Columbus facility, including, but not limited to data, power, cooling, thermal, and C&DH. Also, additional ICDs will be required between the ISS/JSC and your payload, as described in Section 4.2.1. Finally, a Software ICD and Launch Vehicle ICD are also required, as outlined in Section 4.2.1.

Interface testing facilities and testing simulators are provided by the ISS for Columbus Attached Module payloads, such as the PRCU, etc. These ISS-provided simulators are described briefly in Section 7.0.

4.1.4.4 Payload-to-Launch Vehicle/Visiting Vehicle Integration

Launch vehicle/visiting vehicle processing requirements and launch site support may vary with the selected launch vehicle. However, the PD must develop another ICD that complies with the requirements established in the Launch Vehicle Interface Requirements Document. An overview of the launch vehicles'/visiting vehicles' process is outlined in Section 5.0.

4.2 ISS Operations

The mutli-national ISS operations and management functions, including basic functions of the different organizations, are represented in Figure 4.2-1.



Figure 4.2-1. ISS operations and management.

4.2.1 Overview – ISS Integrated Mission Elements

Typically, the integrated mission elements for an ISS external payload include the ISS platform, which will host the Instrument Payload, the Tracking and Data Relay Satellite System (TDRSS) which transmits instrument data from the ISS to the White Sands Complex (WSC), the MCC-H at JSC which manages the American Segment of ISS, The POIC at MSFC which manages all NASA payloads on ISS, and the Payload Developer Payload Operations Center (PD-POC) and data processing facilities provided by the instrument teams. Together, these elements or mission segments ensure successful execution of the mission.

For example, the mission architecture shown in Figure 4.2.1-1 illustrates the integrated mission elements for a typical external NASA payload, the Stratospheric Aerosol and Gas Experiment III (SAGE III).

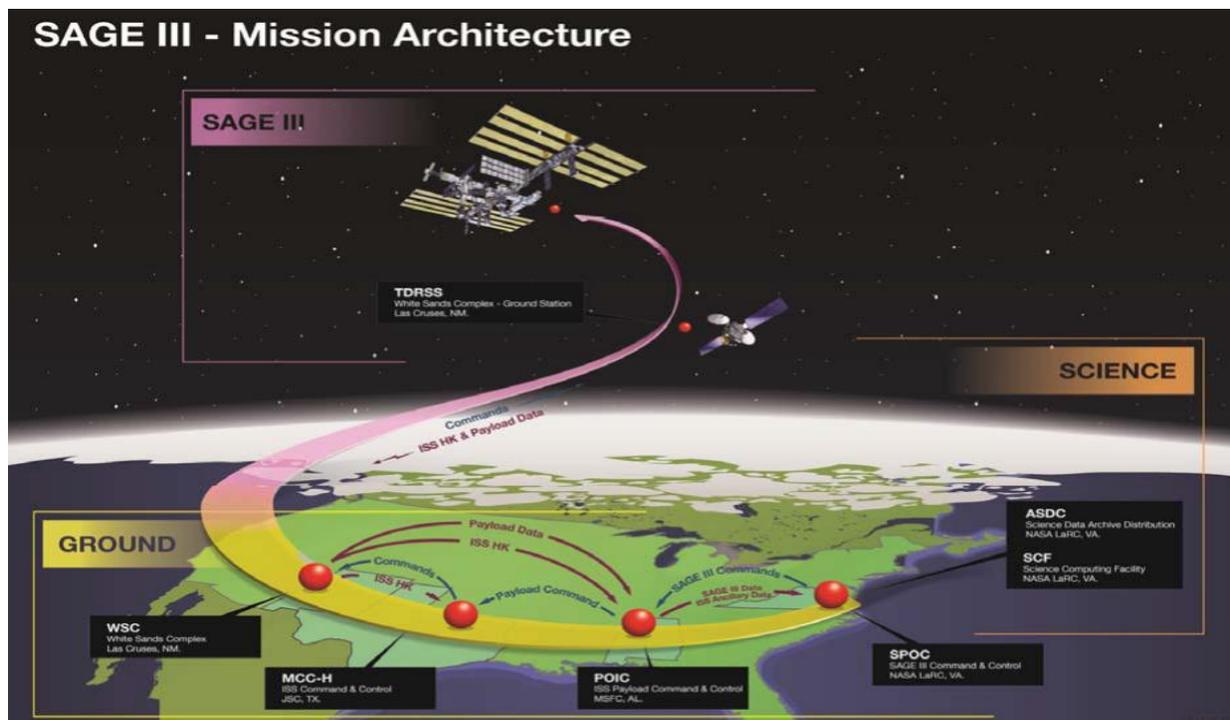


Figure 4.2.1-1. SAGE III on ISS Mission Architecture

ISS Communications

The ISS utilizes the TDRSS Network to command, control, and maintain ISS subsystems and payloads on board the station. The network facilitates the routing of payload data to the different control centers around the world. The communication system provides two-way voice, data, and video communications between the ISS, the MCC-H, the POIC, and payload facilities such as the PD-POC.

ISS communications consists of the TDRSS, the White Sands Ground Terminal, the NASA Integrated Services Network, NASA Communications System, and Internet Voice Distribution System. These resources are primarily transparent to payload operations.

Telemetry

The POIC, located at MSFC, coordinates the transfer of telemetry, and provides short-term (up to 2 years) payload data storage for all payloads. The PD-POC then receives the telemetry in near real-time via bent-pipe approach from the POIC.

Commanding

Commanding of payloads on the ISS requires several steps. First, the PD-POC transmits commands to the POIC. The POIC then transfers the commands to the MCC-H for uplink to the ISS. The MCC-H integrates all ISS commands, including that of the POIC, and sends them to WSC where they are uplinked to the ISS via S-band. Once on-board, the commands are identified by on-board ISS processors and sent to the instrument payload, and the commands are verified through command responses and telemetry. In the event of a failure of the command link between a remote site and the POIC, the POIC may provide backup command capability if necessary. This backup command capability is limited and is provided by the Payload Operations Integration Function (POIF) Cadre. POIF is the Flight Control Team in charge of ISS Payload Operations. A typical command and telemetry path is provided in Figure 4.2.1-2.

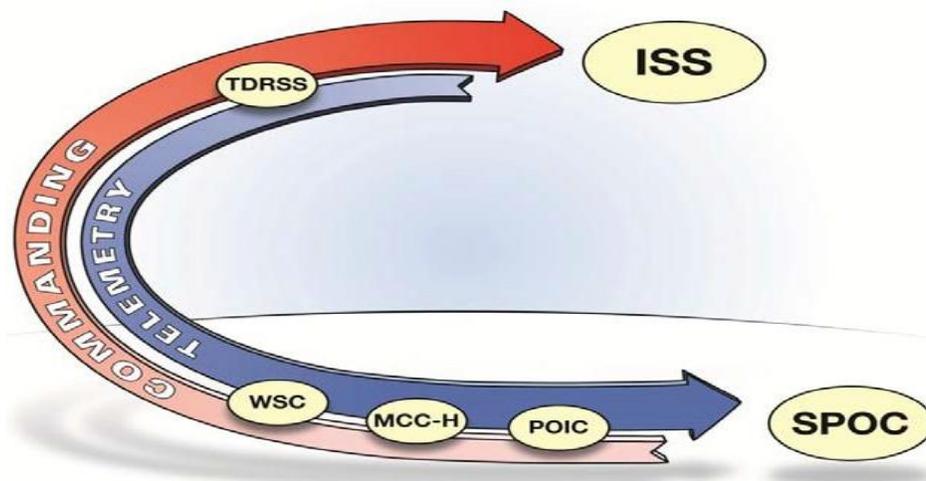


Figure 4.2.1-2. Typical command and telemetry path.

4.2.2 Typical Profile (Day-in-the-Life)

A day-in-the-life for an external science payload is largely dependent upon the mission design, science objectives, instrument design location on the ISS, and level of autonomy of the instrument (on-board autonomy) and ground system (PD-POC). A day-in-the-life for the SAGE III mission has been provided as an example of a typical ISS instrument payload. The following is an excerpt from the *Stratospheric Aerosol and Gas Experiment III (SAGE III on ISS) Mission Concept of Operations, January 14, 2013*³⁶, Section 9.0.

Example: Day-in-the-Life (SAGE III)

SAGE-III, a moderate resolution spectrometer designed to look at the earth's limb, is to be installed via the SPDM at the SAGE III designated location on the ELC-4, Site 3.

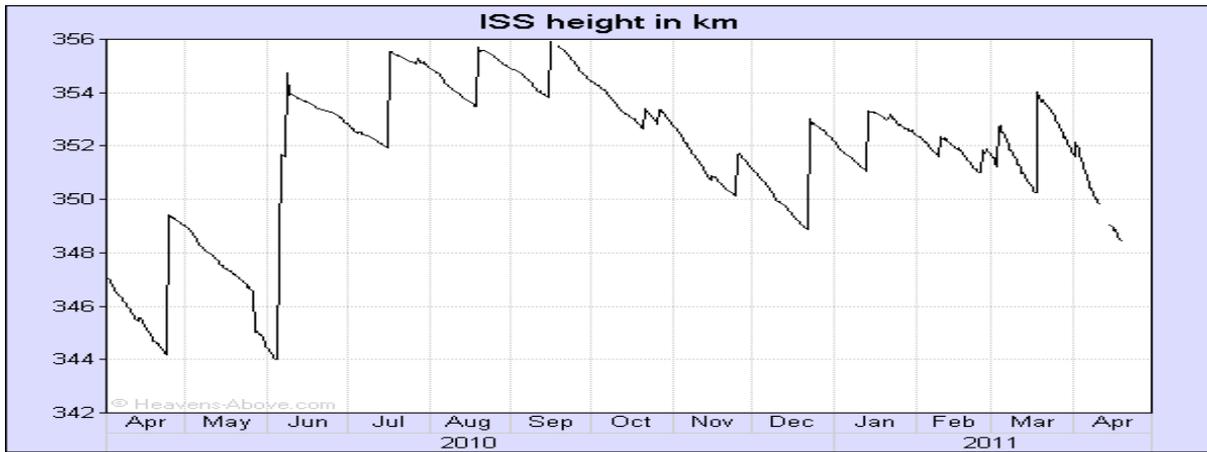


Figure 4.2.3-1a. ISS altitude profile.

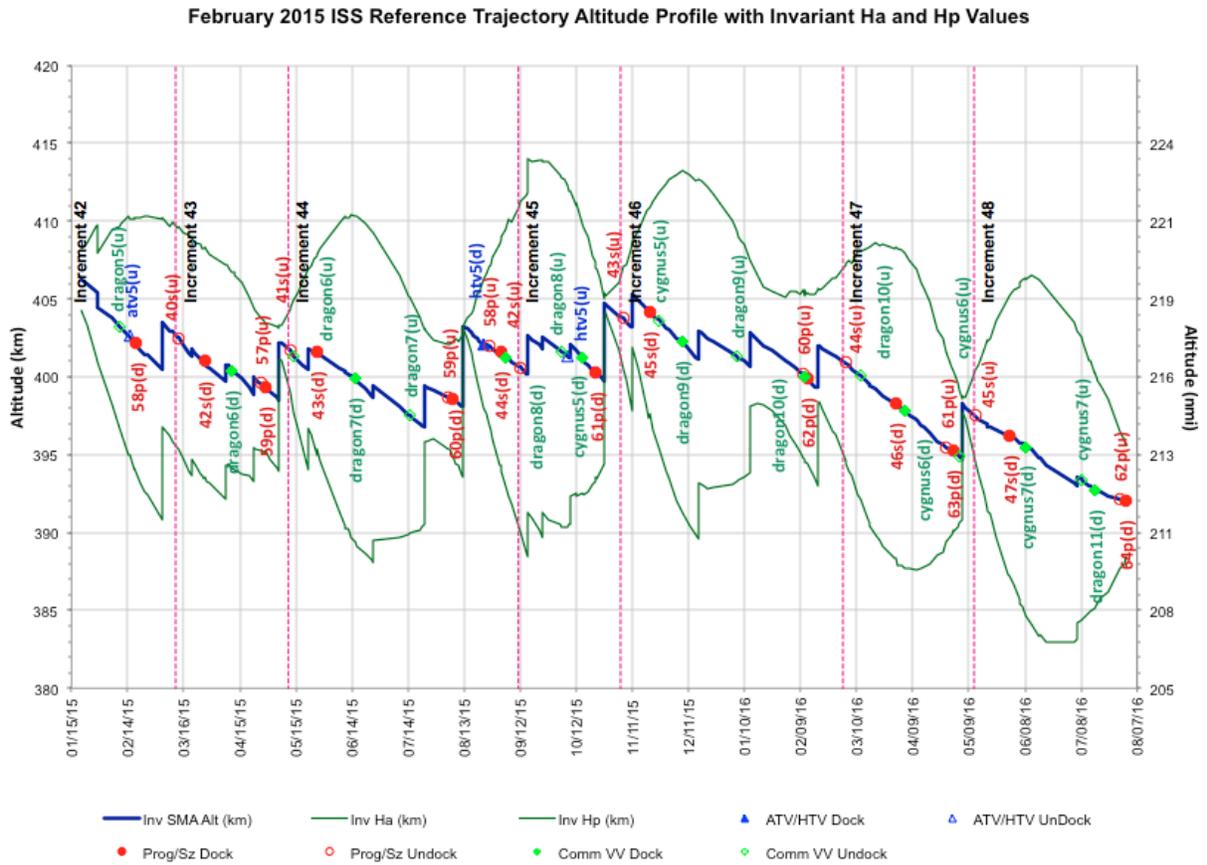


Figure 4.2.3-1b. ISS reference trajectory attitude profile

Figures 4.2.3-1a and b are graphs of the average altitude off ISS during certain months in 2015. This is the average altitude but the apogee and perigee of the orbit can vary as much as 25 km within any single orbit (as shown).

CHECK THE ESP DIVISION WEBSITE AT
<http://espd.gsfc.nasa.gov/isseppg/>
 TO VERIFY THAT THIS IS THE CORRECT VERSION PRIOR TO USE.

The attitude of the ISS is defined by the classical pitch, yaw and roll angles, according to the LVLH coordinate system, and generalized as depicted in Figure 4.2.3-2.

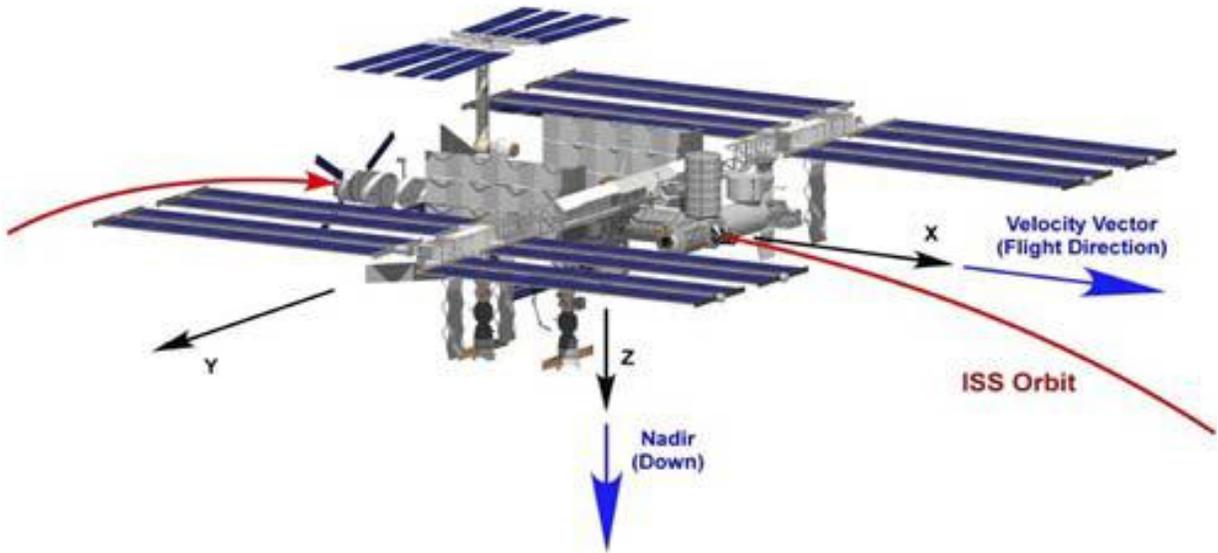


Figure 4.2.3-2. +XVV orientation of ISS.

Attitude & Ephemeris (State Data)

The ISS GN&C flight software provides navigation and pointing data to ISS payloads. Examples of these parameters are:

- Attitude and rate knowledge
- State-vector (position and velocity)
- Mass properties estimates (C.G., etc.)
- Solar array (in-plane) and Beta (out-of-plane) sun tracking angles
- Solar line-of-sight unit-direction-vector components

Data latency COULD be an issue for time-critical payloads requiring real-time ephemeris data, as some users may receive the data a couple of seconds AFTER the actual data measurement is taken. However, the data is time tagged such that the time tag, position, velocity, and attitude data are a time homogenous data set which can be propagated to the time of interest without losing accuracy. Please note that

Broadcast time is the time broadcast by the ISS computers that is intended to be indicative of current time. This is not the timestamp associated with the position, velocity, and attitude data. Broadcast time error is available, and can be used to correct the broadcast time such that the corrected time stamp is accurate to within +/- 55 microseconds. Therefore, seconds and sub-seconds must be added to calculate the total time error.

In summary, $\text{CorrectedTime} = \text{BroadcastTime} - \text{TimeError}$ The ISS can fly in various attitude control modes. The most common operational mode is the Torque Equilibrium Attitude control utilizing a non-propulsive attitude controller and the payload should be prepared to operate in this environment. This mode provides the largest deviations from an Earth reference system

(nadir viewing) and is therefore the most critical for attached payload operation. ISS Attitude and Ephemeris performance data are provided in Tables 4.2.3-1 and 4.2.3-2.

Table 4.2.3-1. Attitude and ephemeris performance data.

Parameter	ISS Requirement	Typical/Predicted Performance
Torque Equilibrium Attitude (TEA) envelope for solar beta < -65°	Yaw: -7° to 15° Pitch: -20° to 15° Roll: -15° to 15°	Yaw: -7° to 0° Pitch: -10° to 2° Roll: -1.5° to 1.5°
TEA envelope for solar beta ≥ -65° and ≤ 65°	Yaw: -15° to 15° Pitch: -20° to 15° Roll: -15° to 15°	Yaw: -7° to 0° Pitch: -10° to 2° Roll: -1.5° to 1.5°
TEA envelope for solar beta > 65°	Yaw: -15° to 7° Pitch: -20° to 15° Roll: -15° to 15°	Yaw: -7° to 0° Pitch: -10° to 2° Roll: -1.5° to 1.5°
Non-Microgravity & Microgravity Yaw: ±3.5° Attitude Control about the TEA Pitch: ±3.5° (Assembly Complete) Roll: ±3.5°	Momentum Minimization	Yaw: ±1° Pitch: ±1.25° Roll: ±0.75°
	Attitude Minimization	Yaw: ±0.13° Pitch: ±0.25° Roll: ±0.75°
	Controller	Yaw: ±0.5° Pitch: ±0.88° Roll: ±0.75°
	Blended Controller	Yaw: ±0.5° Pitch: ±0.88° Roll: ±0.75°

Table 4.2.3-2. ISS position & attitude.

Parameter	ISS Requirement	Typical/Predicted Performance
Attitude Rate: Non Microgravity	$\pm 0.02^\circ/\text{sec}/\text{axis}$	$\pm 0.005^\circ/\text{sec}/\text{axis}$
Attitude Rate: Microgravity	$\pm 0.002^\circ/\text{sec}/\text{axis}$	New controllers will be designed when microgravity is invoked
Attitude Knowledge @ navigation base (S0 Truss)	$0.5^\circ/\text{axis } 3\sigma$	$\leq 0.25^\circ/\text{axis } 3\sigma$
Attitude Knowledge on the non-articulated portion of the ISS	$3.0^\circ/\text{axis } 3\sigma$	$\leq 1.0^\circ/\text{axis } 3\sigma$
Rate Knowledge	$0.01^\circ/\text{s}/\text{axis } 3\sigma$	$0.001^\circ/\text{s}/\text{axis } 3\sigma$
Orbital Position Vector Knowledge (non re-boost)	$3000 \text{ ft } 3\sigma$	150 ft Root Mean Square (RMS)
Orbital Velocity Vector Knowledge (non re-boost)	No requirement	0.23 ft/sec RMS
Orbital Semi-major Axis Knowledge	$1000 \text{ ft } 3\sigma$	50 ft RMS

Table 4.2.3-3. Time distribution of ISS attitude and ephemeris state knowledge and control data.

Parameter	ISS Requirement	Typical/Predicted
Attitude, Position & Timestamp Broadcast Frequency	1 Hz	1 Hz
Broadcast Timestamp Error ± 5 milliseconds	Auto-sync (not typically used) ≤ 0.03 milliseconds	Manual sync (typically used) ≤ 1 seconds

Table 4.2.3-4. Specifications and predictions of attitude rates.

Parameter	Micro-G Spec	Prediction
Attitude Rate (deg/sec/axis)	± 0.002	$\pm 0.02 \leq \pm 0.005$
Attitude Rate (arcsec/sec/axis)	± 7.2	$\pm 72 \leq \pm 18$

*ISS GN&C Performance Requirements and Analytic Predictions

Station Non-propulsive (Microgravity Mode) Attitude Control Requirements introduce the following augmentations to the tables above. However, ISS has yet to utilize microgravity mode, and it is unclear if microgravity mode will ever be utilized. ISS does typically fly several weeks in a row, in between dockings/undocking or berthings/unberthings, in quiescent operations with no thruster firings. The microgravity mode requirements are as follows:

- Maintain non-propulsive control for 30 continuous days
- Provide a total of 180 days/year of microgravity

Position and velocity data are provided to users on board ISS in feet and feet/sec, in telemetry in km and m/sec for the (Conventional Terrestrial Reference System) CTRS position and velocity, and in telemetry in meters and m/s for the inertial J2000 position and velocity. The

CTRS is WGS84 (World Geodetic Survey of 1984 – the CTRS frame used by GPS). The Rotation, Nutation, Precession (RNP) matrix used by GNC is available if needed. It is suggested that payloads use the CTRS information to avoid having to apply the particular Rotation, Nutation and Precession (RNP) matrix being used by ISS GN&C.

It should be noted that Broadcast Ancillary Data (BAD) data are defined in SSP 50540.³⁷ The BAD data are broadcast to payloads, and include additional data such as solar panel rotary joint angles.

4.2.4 Visiting Vehicle Timeline

The operation of the ISS is defined and planned in increments — characterized by time between major events such as spacecraft docking and undocking. Payloads will be required to operate across multiple increments and must plan science and payload operations across multiple docking events. The PD should define an operational procedure which includes these events. A typical visiting vehicle profile is provided in Figure 4.2.4-1.

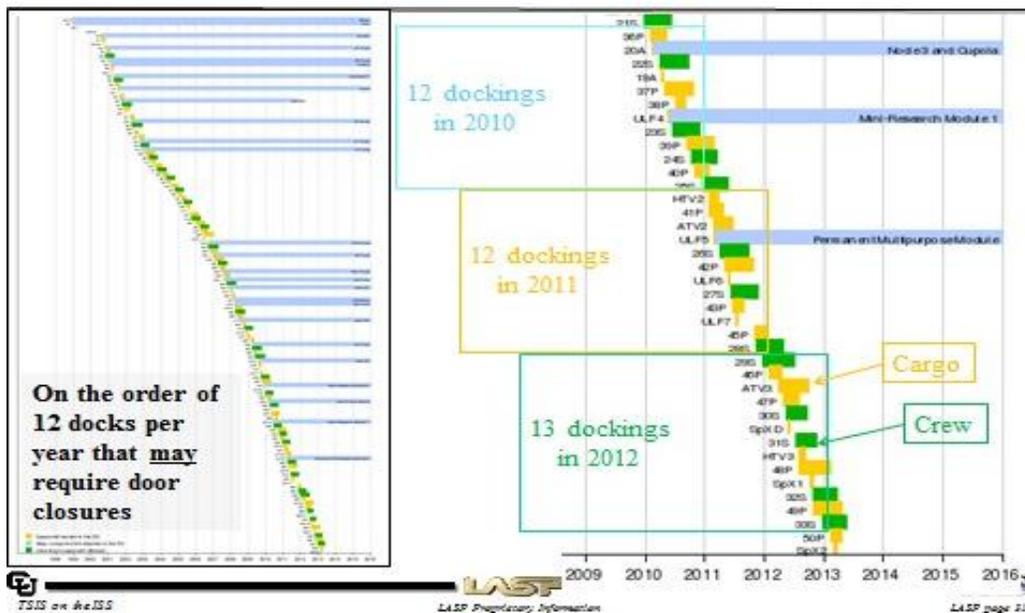


Figure 4.2.4-1. ISS periodic docking profile (historical).

Payload Operations Integration Function (POIF) The POIF is a combined civil service and contractor team tasked to perform payload operations integration for all NASA payloads, located at NASA’s MSFC, that coordinates, integrates, and manages payload operations planning, crew training operations products, payload safety, and ground personnel training. The POIF staffs the flight control team positions in the POIC control center, whereas the POIC Operators are responsible for facility interfaces and internal facility data system operations. The POIF and POIF Flight Control Team (FCT) provides 24-hour a day, 7-day a week (24/7) ISS operations support for real-time command and control of NASA ISS payloads and support systems. Figure 4.2.4-2 provides an overview of the interfaces between the Payload Development Team and the ISS RIO, the Flight Operations Directorate (FOD), and the POIF.

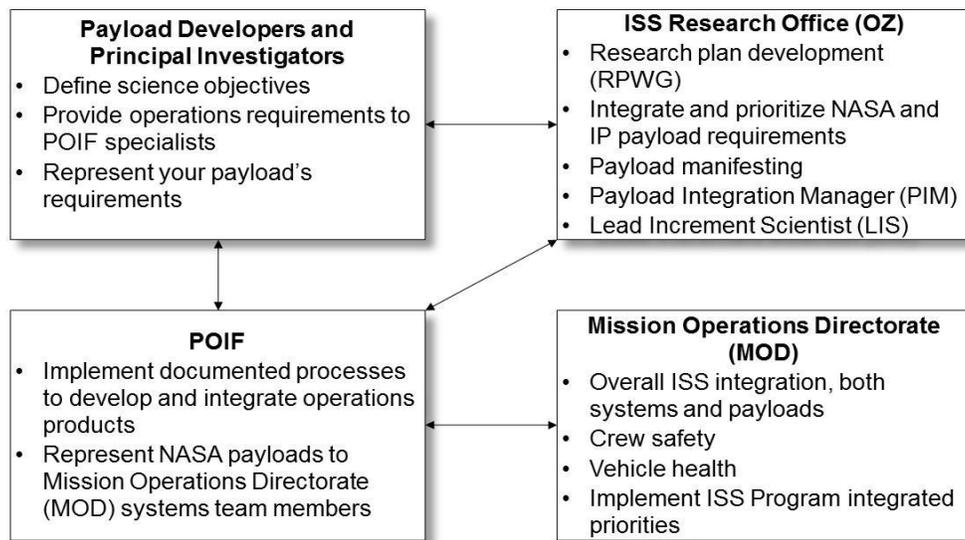


Figure 4.2.4-2. ISS payload operations organizational interfaces.

The POIF manages all NASA payloads on board the ISS. POIF responsibilities include managing ISS payload commanding, monitoring H&S of the payload, and collecting, processing, and routing of payload data to payload control centers. A POIC Customer Service Team (CST) member will be assigned to provide an early understanding of all Huntsville Operations Control Center (HOOSC) ground data services.

The POIC Customer Service Team is responsible for supporting the development of your ground data service (GDS) needs:

- Provide detailed explanation of the POIC's ground services.
- Work with the customer to provide a "best fit" set of POIC ground support services that satisfy the customer's ground operations concept.
- Assist the customer with documenting their GDS needs in the appropriate Program Level and POIC requirement documents.
 - PIA Letter
 - Ground Data Service Blank Book
- Provide cost impacts to optional services or other unique customer GDS needs.
- Coordinate the creation and integration of all the customer GDS needs with customer's host support facilities.
- Ensure Payload Operations team training on POIC GDS.
- Provides a POIC single point-of-contact for GDS needs for the full life cycle of the experiment's operations.
 - Pre-mission
 - Mission
 - Post-mission

The POIC sends payload commands to the MCC-H for transmission to the ISS. The PD is responsible for the generation of payload commands, which are typically generated at the PD-

POC; however, ultimate authority to command the payload will come from the POIF. Figure 4.2.4-3 provides a typical interface diagram and data flow for an ISS payload.

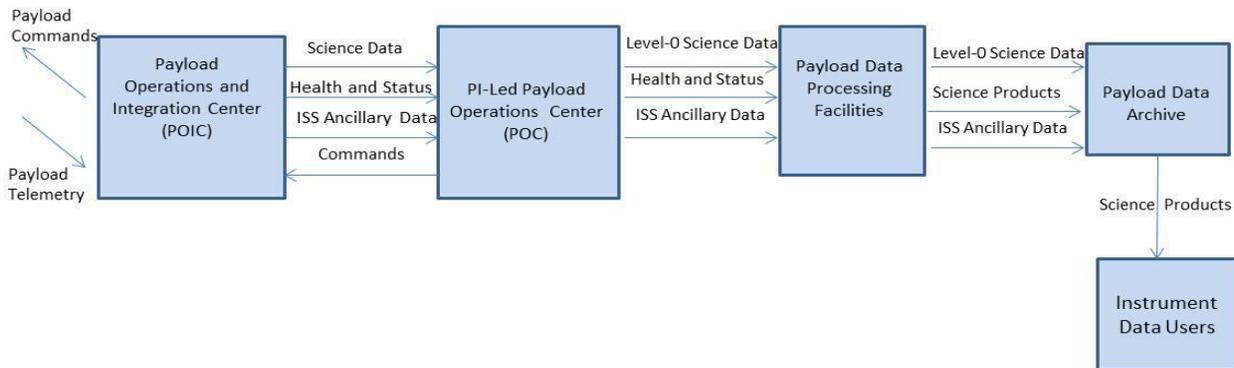


Figure 4.2.4-3. Typical POIF to PD-POC interface and data flow.

The POIF provides command anomaly resolution support for any failed or lost payload commands, and monitors the H&S of all payloads 24/7 (with a smaller, minimal operations team during the off shifts). The POIF will notify Payload Operations Center (POC) operators if an out-of-limits situation is detected or if the payload must be placed into survival mode. The POIC has the ability to store all unprocessed payload telemetry and science data for up to 2 years, and has the ability to transmit real-time science to the POC and perform Level-0 processing on all data.

4.2.5 Mission Operations Functions

Typically, mission engineering and planning for external payloads will be carried out at the PD-POC with support from the POIF for mission-critical functions. The level of autonomy, and thus the level of manpower, required to support on-orbit operations, will determine complexity of both the ground system and ultimately cost of operations. ISS payload operations support requirements for typical ISS external payloads range from 8 to 5 to 24/7 depending on the science requirements and level of ground system autonomy.

Mission Planning

Operations planning is performed by PD operations team. Once again, the level of on-board autonomy will determine the level of effort required for mission planning. An instrument payload with significant on-board autonomy will allow for minimal mission planning during nominal operations.

Typically, PD-POC operators develop and utilize mission-planning aides to perform functions such as determining when science observations may be scheduled. Before every increment, the PD coordinates with the POIF's Payload Planning Managers (PPM) for inputs into the On-board Operations Summary and the Increment specific Execute Planning Ground Rules and Constraints (GR&C). During the increment, the PD-POC works with the PPMs to develop inputs into the Weekly Look-head Plan and the Short Term Plan that includes commanding and data downlink times and durations. The PD-POC also coordinates with the POIF regarding ISS events such as visiting vehicles, microgravity timelines, and collision avoidance, which may impact the ability to perform science on the ISS.

Flight and Ground Software Management

Typically, instrument flight software and the ground processing software are managed and maintained by the PD at the PD-POC, including any modification and testing of the software. All PD-managed software must be under CM control throughout the mission life cycle.

Instrument Payload Commanding

POC operators plan and develop all command loads for the instrument payload. Command loads are developed and tested at the SPD-POC prior to being transmitted to the POIC for uplink to ISS. The command loads are typically developed and tested the week prior to transmitting it to the POIC. The frequency of this process is largely dependent on the science mission and instrument design. For a typical mission, command loads are transmitted to the POIC for transmission to the ISS via the MCC-H and TDRSS.

Status Monitoring and Trending

The PD-POC is responsible for analyzing real-time spacecraft H&S data, and performing long-term trending of housekeeping data. Typical PD-POC tools will automatically select, store, trend, display, and report spacecraft housekeeping data. The PD-POC telemetry monitoring system notifies payload operators whenever an out of limit situation occurs.

Anomaly Ground Response

The first line of ground response for an ISS external payload is the 24/7 POIF/POIC team where the operator's action is authorized in the Payload Regulations for pre-defined anomalies. Once an anomaly has been identified, the next step is to assure the safety of the instrument (or take immediate action to safe the instrument) and then perform an initial report, including logging it into a Nonconformance Reporting System. The PD then develops a recovery plan (if required) and performs the necessary actions to recover the payload to an operational state. The POIF/POIC operators are notified of any instrument or ground issues via red or yellow alarms or ground system displays.

Training

Flight operations personnel receive training on the processes and procedures necessary to conduct the mission activities successfully. The typical training program will begin approximately 1 year prior to launch. The PD and POIF are obligated to develop and administer training on their respective systems prior to launch. Integration and testing activities and the training program begins prior to L-1 year.

Communication

ISS uses TDRSS for its communication, which allows for real-time communications; however, approximately 30% of the time there is no communication with ISS. To ensure no communication is lost due to communication limitations, the PD may decide to include a data recorder in the instrument package or accept some additional risk and elect to use the Ku Comm Unit recording capability for Loss of Signal (LOS) recording.

4.2.6 Payload Developer Requirements for Operations

Throughout the mission life cycle, the PD will work closely with the POIF team to develop the necessary processes, procedures, and operations products necessary to ensure the H&S of the payload, and success of the mission. The key processes and procedures required for a successful launch and payload operations include GR&C, Flight Rules, Payload Regulations, Ground Command Procedures, Command and Telemetry Databases, and Crew Training Package and are described below.

Ground Rules and Constraints (GR&C)

GR&C list specific payload planning requirements, constraints, or other basic ground rules which are crucial for experiment success and the H&S of the payload. They are distributed to PD teams by email for *Informal* review at L-2.25 months and for *Final* delivery at ~L-1 month. GR&C are incorporated into the ISS Increment Specific Execute Planning GR&C which includes ESA, JAXA, and NASA/CSA payloads for each increment. The GR&C requirements and processes are further described in the Payload Operations Handbook, Volume 1³⁸, Section 7.2, *Payload Increment-Specific GR&C Development*.

Flight Rules

Flight Rules are published by the ISS Flight Director Office, enforced in real-time by the Flight Director, and are adhered to by all partner control centers. The Flight Rules, unique to each payload, are defined primarily to maintain crew safety, ISS and visiting vehicle integrity, and document mission success criteria. Flight Rules and requirements for the PD teams are further discussed in ISS Fram Based Payload Common Launch Interface Requirement Document (IRD), SSP 57012, Standard Operating Procedures 6.1.³⁹

NASA Payload Regulations

NASA Payload Regulations are rules and constraints designed to maximize payload mission success and minimize conflicts between/among payloads. Payload Regulations are published by POIC and adhered to by POIC and NASA PD teams. Payload regulations and requirements for the PD team are discussed in Reference 39, Standard Operating Procedures 6.2.

Flight Rules/Payload Regulations Publication

Flight Rules are updated periodically, but not necessarily on a flight-by-flight basis. The POIF collects payload Flight Rule candidates starting at L-7 months. The Payload Operations Directorate will coordinate with MOD to publish the updated payload-specific Flight Rules. The PD team can expect to begin development of Payload Regulations at L-7 months. The deadline for Payload Regulations is L-2 months. Payloads launching on flights not corresponding to increment boundaries, will have their Payload Regulations incorporated at the logical increment mark, or updated by Operations Concept Review.

Flight Rules updates and Payload Regulations are further discussed in Reference 39, Standard Operating Procedures 6.1 and 6.2.

Ground Command Procedures (GCPs)

GCPs contain detailed step-by-step instructions for POIF operations personnel to control and monitor payloads and the ISS systems supporting those payloads. The GCP Book Manager will

work with the PD team and Operations Lead to develop new or update existing GCPs to support the mission. GCPs are under strict configuration control, updates for new payloads are controlled by the Engineering Change Request (ECR) process. The GCP Book Manager submits the ECR at L-3 months.

Inputs required by the PD team include but are not limited to:

- Safing procedures
- Commands delegated to the POIF
- Station resources that may be required
- H&S parameters to be monitored and associated steps to respond
- Unique resources or systems configurations

GCPs and PD team requirements relative to the development and/or update of GCPs are further discussed in Reference 39, Standard Operating Procedures 6.3.

Payload Command and Telemetry(C&T) Databases

Payload C&T databases are built by the POIF using PD team inputs entered in the C&DH data set. C&T database builds are generated on an increment basis; data for the first revision of a database must be in PDL at ~L-8 months.

Preliminary versions are provided to the PD team and the POIF/POIC FCT for testing at ~L-7. Depending on the payload development, integration and test, or operations test schedules, the C&T database may be required prior to L-7. If the C&T database is required for testing earlier than L-7, the PD team must request and coordinate dates through the Payload Software Control Panel All C&T database changes/corrections resultant from testing must be updated in the C&DH Data Set. Data for the flight rev of the databases must be in PDL by L-5 months and available for use/testing at L-4 months. During increment operations, transitions to new databases are pre-coordinated, then scheduled and executed. The POIF Database Lead is the primary point of contact for C&T database questions.

C&T databases and PD team requirements are discussed in detail in Reference 39, Standard Operating Procedure 8.2.

Payload Training

Payload Training for the POIF FCT is conducted by the POIF. The POIF Operations Lead builds the payload training materials which can be in the form of either a computer-based training or an operations summary. The PD team will provide information for the development of training materials during the training package development and baseline phase. Refer to Reference 39, Standard Operating Procedures 10.12 for additional detail.³⁹

4.2.7 Data Flow Planning

Data flow planning is performed by the Data Management Coordinator (DMC) team and coordinated with the mission elements including the POIF and the PD team. ISS payload data flow is limited by the ISS downlink rate as defined in Section 3.0, Table 3.0-1, of the Guide. Data latency requirements are typically defined in a payload's Level-1 requirements and are within the ISS specification for data downlink and latency.

Payloads share bandwidth with ISS systems data downlink requirements. The planned routings and configurations of the on-board data, including ISS payload data, are defined in ISS Data Flow Plan which is used by the DMC to manage these systems for real-time execution.

Data flow analysis is performed real-time and contains the initial plan for data and video system configurations, including start/stop times of data and video producing activities, TDRSS coverage, and recorder management. To plan the routings and configurations for US payloads accurately, payload data and video requirements must be reviewed and understood by the DMC Team during the increment preparation.

4.2.8 Data Downlink Capability

The summary of the current ISS data downlink capability and parameters are provided below, which should be helpful to the PD in assessing and designing their payload data capabilities.

- Current throughput pipe allocated to science data: The system is sized for 259 Mbps (after overhead has been taken into account). Out of that 259 Mbps, about 100 Mbps is typically allocated to systems. The remainder (currently 159 Mbps) is used for science data downlink. This is for all ISS payloads (excluding Russian payloads). Out of that payload available bandwidth 12% is allocated to JAXA payloads; 8% of it is allocated to ESA payloads; the remainder is for NASA payloads.
- ISS onboard storage size, data rate recording and downlink rate: The sustainable record rate for ISS Ku Communication LOS recording is around 100-120 Mbps. At this rate with appropriate reserved playback bandwidth the system can record and playback continuously. If there were that much recording occurring, it would need something on the order of 45-60 Mbps reserved just for playbacks based on 70% Ku coverage and 30% LOS (record time). Current trends shows requirements for recording much lower than that (about 5-30 Mbps is the typical range as of early 2015). The required recording rates vary with the complement of activities scheduled for operations on ISS.

The Recording capacity is 300 GB of space. This space is currently divided ~ 20/80 to systems/payloads activities. This division is configurable based on coordination in planning between systems and payload operations teams to determine where space is needed to support scheduled operations.

- Frequency and percentage of data from the ISS onboard recorder that is downlinked from ISS when TDRSS is available: The Ku Communication Unit recorder is a Communications Outage Recorder. This means its design was based on recording data gaps (during LOS) only. During nominal operations POIC utilizes what is called "Auto Ops" mode recording which automatically stops recording during communication coverage windows and starts recording when communication coverage is lost.

If necessary due to failures of the on board communication system, TDRSS, or ground systems, POIC can place the Communication unit in manual record for the duration of the issue or until storage runs out. To date there has never been any incident where the onboard storage ran out of space due to high rates, or a failure.

ISS science data recorded during LOS is downlinked during every viable Ku coverage window. In practice, this means anything recorded typically reaches the ground within 3 hours. The vast majority is downlinked in the first viable pass after it is recorded (typically within 30 to 45 minutes).

- Amount of science data (on average) within a 24-hour period: It averages out to be around 25 to 30 Mbps stream of actual throughout on a given day as of Early 2015. This is subject to the complement of experiments going on at any given time. There is also a significant discrepancy between the actual rates observed and the required rates that are scheduled. For example, at present, based on requirements POIC probably has more like 50-75 Mbps scheduled support on average.
- Typical latency for science data from ISS to a Principal Investigator's location: The round trip for the JSL-2 (HPEG) is approximately 700 ms. The one-way trip should be about half of that, so the latency is about 350 ms.
- Typical daily estimate of Ku (TDRSS) coverage: A realistic Ku coverage estimate is about 70% on average. During high beta it can drop to as low as 50%. During crew wake periods it is often as high as 85%.
- Health and Status data downlink summary:
 - Health and Status data is always in the downlink and is transmitted immediately once Ku coverage is resumed.
 - Payload Health and Status is currently limited to 256 Kbps (from all active Payloads connected to Payload local buses). Payload health and status is downlinked at 1/sec
 - Health and Status latency time from ISS to Principal Investigator's location is approximately 270 ms

4.2.9 Summary

In summary, the PD team works closely with the POIF and POIC in the development of operations products for testing and on-orbit operations. As such, it is crucial that the PD team provide timely inputs to operations products such as:

- Planning data set
- Crew procedures
- Displays
- Crew training requirements
- Ground command procedures
- Payload regulations

The POIF is the PD team's interface to the ISS program during increment preparation, ground testing, and real-time operations. It is crucial that the PD team work with POIF and POIC personnel to understand fully the operations concepts, hardware, software, and payload specifics including system Interfaces, commands, data, video, unique resources, and hazard controls. The PD team must provide a GDS HOSC CST Representative to identify ground support service needs and GDS Blank Book completion as well as GDS support from the payload's host operating location(s). This support service includes network management, systems management, IT Security, and ground services testing.

Figure 4.2.9-1 provides a list of data input products, and a relative time schedule for tasks/product completion.

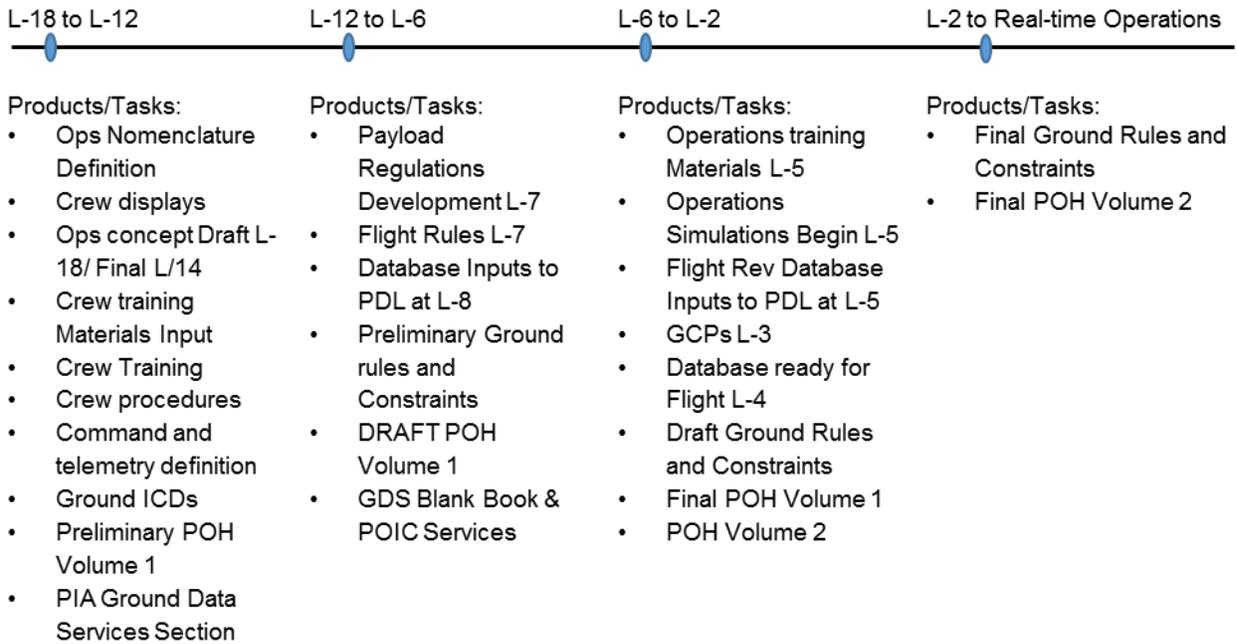


Figure 4.2.9-1. Operations products timeline.

5.0 Launch Vehicles and Visiting Vehicles Supporting ISS External Payloads

5.1 Overview

Currently, only one vehicle provides launch services to the ISS for ISS external payloads: SpaceX's Falcon 9 v1.1 (see Figure 5.1-1). (The HII Transfer Vehicle, jointly developed by JAXA and Mitsubishi Heavy Industries, is nearing retirement and is not included in the Guide.)



Figure 5.1-1. SpaceX's Falcon 9 v1.1.

It must be noted that the ISS program is currently conducting a procurement for additional ISS flights, and this procurement is being competed, as of the time this Guide was developed. Therefore, the selection of a visiting vehicle COULD change for future ISS flights. Please confer with the ISS Payloads Office for the most current vehicle availability.

5.1.1 Launch Vehicles/Visiting Vehicles

Table 5.1.1-1 provides a brief summary of the Dragon, which is the cargo carrier assigned for external payloads (unpressurized portion), referred to as the Dragon trunk.", and that is integrated onto the Falcon 9 v1.1 launch vehicle.

Table 5.1.1-1. Accommodations Table.

	Dragon
Height	69.2 m (206 ft)
Maximum Carrying Mass	13,150 kg (28,990 lbs)
Fairing Size	Diameter: 5.2 m (17 ft) Length: 13.9 m (45.6 ft)
Fairing Volume	295 m ³ (10,400 ft ³)
Launch Site	Cape Canaveral Air Force Station, FL

The Dragon capsule flies aboard the Falcon 9 v1.1 launch vehicle, and is separated from the vehicle on-orbit. The Dragon is the “spacecraft” that will actually rendezvous with the ISS. Currently, all external payloads that fly on the ELC (ITA), Columbus-APM, or the JEM-EF can be accommodated in the Dragon trunk. At present, the Dragon is the ONLY serviceable spacecraft that has this capability (following the HII HTV retirement). Therefore, external payloads should plan on using the Dragon/Falcon 9 v1.1 vehicle now. All external payloads will be mounted within the Dragon trunk, which is unpressurized.

The only service provided by the Dragon, excepting structural attachments, is survival power, listed in Table 5.1.1-2. Currently, there is no command and data capability, or other services, for external payloads available in the Dragon capsule trunk. NOTE: This list DOES NOT include all payload constraints for the vehicles. Other conditions, such as vibrations, acoustics, and pre-launch environments can be obtained through the launch vehicle/visiting vehicle User’s Guides, ISS document SSP 57012³⁹, and by contacting the sponsoring organization, and your PIM.

Table 5.1.1-2. ISS visiting vehicles.

	Dragon
Height	5.1 m
Payload volume	Unpressurized: 14 m ³
Payload mass	Unpressurized: 3,000 kg (total, including ISS resupply)
Power output	20 W
Flight thermal	≤93 °C (200 °F)
Data	N/A in Dragon trunk
Attachment Capabilities	Can use ExPA/FRAM, JEM-EF attachment or mission-specific attachment mechanism

5.1.2 Docking

Vehicles flying towards the ISS will park ~10 m from the ISS. The two vehicles that physically dock with the ISS are the Dragon, deployed from the Falcon 9, and Cygnus, which is deployed by the Athena launch vehicle. The ISS robotic arm SSRMS will grab the vehicle and attach it to a designated attachment site. There are three available attachment sites: Node 1 Nadir, Node 2 Nadir, and Node 2 Zenith. The ISS robotic arm will then transfer the payload from the vehicle to the destination attachment site. During this time, the payload must be able to survive without electrical and thermal accommodations.

This process may take 6 hours for robotic transfer to the ELC or CEF from the Dragon. However, transfer of a payload from a visiting vehicle to the JEM-EF attachment sites may take 7 hours because it requires a handoff from the SSRMS robotic arm to the JEM robotic arm.

5.1.3 General Launch Milestones

Important milestones, such as dates by which critical documents must be submitted to organizations, can be found in Section 4.1, Integration Template, or OZ-10-056⁴⁰. Table 5.1.3-1 provides a top-level view of important milestones. However, each launch vehicle/visiting vehicle has different milestones depending on the type of payload and specific mission details. The milestones listed above are representative of general LV milestones. These milestones may change, to some degree, based on unique flight parameters.

Table 5.1.3-1. Major launch vehicle milestones.

Baseline and Procure Launch Services	~L-3-2 years
Launch Site Operations	~L-6 months to L-10 days
Launch	~L-10 days to Launch
Post-Launch	~To L+3 months

5.2 Vehicle Unique Requirements and Procedures

5.2.1 Falcon 9 v1.1

NASA currently has a Commercial Resupply Contract with SpaceX which guarantees at least 18 launches to the ISS. This contract is well underway, and a follow-on contract will be issued, probably by the time you fly. You must be aware that it is uncertain which vehicle or vehicles will be chosen for future ISS launches when a new NASA contract is issued. For now, plan on using the Falcon 9/Dragon (trunk section) as your primary launch vehicle/visiting vehicle. A compartmentalized sketch of Dragon is contained in Figures 5.2.1-1 and 5.2.1-2.

Currently, external payloads will fly on the SpaceX Falcon vehicle. SpaceX uses a “ship and shoot” methodology at its launch center. The Dragon, a visiting vehicle that rendezvous with the ISS, has a lifespan of ~2 months, after which it is filled and released from the ISS. The trunk is jettisoned and burned in the atmosphere, while the pressurized section returns up to 3,300 kg in 10 m³ of space. Dragon will land off the coast of California. More information about the mission overviews can be found in the *Falcon 9 Launch Vehicle Payload User’s Guide, Rev. 1*,⁴¹ Section 2. In addition, a picture of Dragon on-orbit can be found in Figure 5.2.1-3.

For payloads launched on a SpaceX vehicle, the payload’s launch site ground processing requirements will be levied and controlled by the SpaceX flight-specific launch site support document. Payload ground processing requirements at the SpaceX launch complex shall be negotiated between the PD and the ISS. Requests for power and/or data during transport on the SpaceX vehicle shall be documented in the PIA. In general, SpaceX will provide a payload processing facility (PPF) and launch site. In general terms, the unpressurized cargo owner will arrive at PPF, be moved into facility, where Payload Developer can perform short duration offline work. Upon completion, the payload will be turned over to SpaceX who will integrate payload into trunk. Currently, vehicle flight support equipment manufacturing begins at L-18 months. PD hardware must be delivered to KSC at L-3 months for ISS inspections, and then must be delivered to SpaceX at L-2 months, and payloads will be loaded and integrated onto the vehicle at about L-9 days. Late-load accommodations are also available.

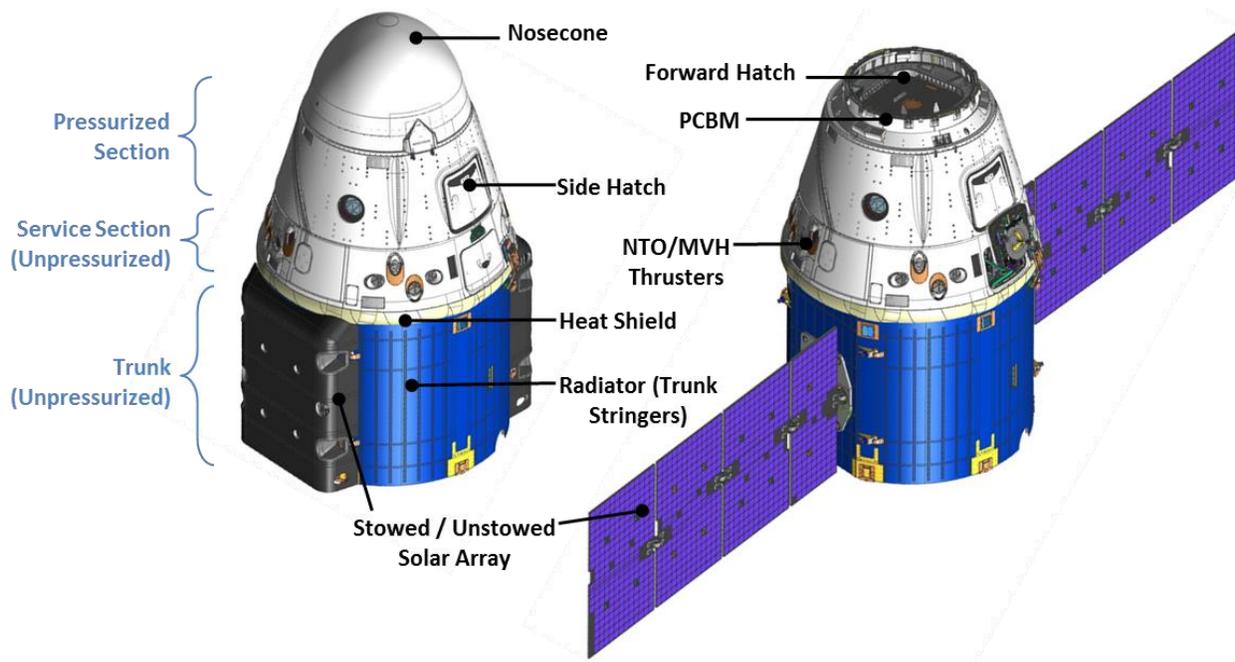


Figure 5.2.1-1. Dragon compartments.

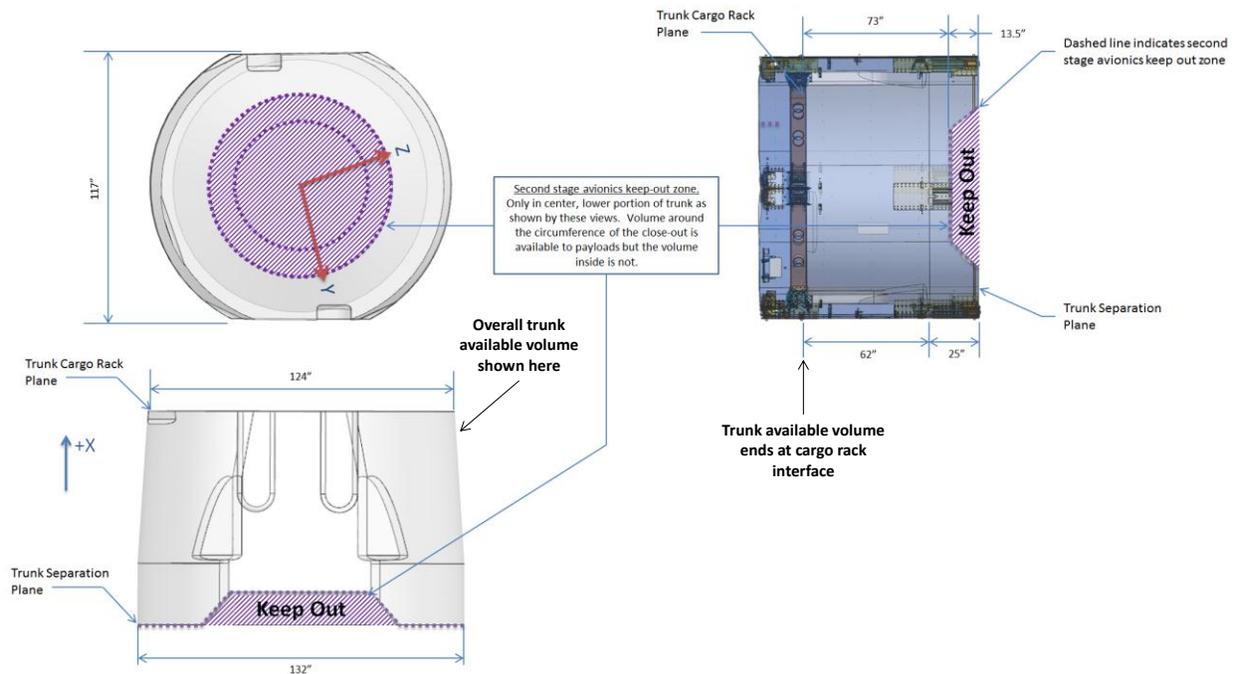


Figure 5.2.1-2. Dragon major components.



Figure 5.2.1-3. Dragon.

5.2.2 Dragon

Dragon is a fully autonomous spacecraft with both pressurized and unpressurized sections. As shown in Figure 5.2.2-1, Dragon is composed of two main elements: the pressurized cargo module, or “capsule,” and the unpressurized cargo module, or “trunk”. The capsule performs all the functions of a service module. It includes a pressurized section, a service section, and a nosecone, and contains all the structure and subsystems required for transport of pressurized cargo to the ISS.

The trunk is an aluminum welded and riveted structure that provides external cargo capability in addition to serving as the adapter between the Dragon capsule and the Falcon 9 launch vehicle. The trunk also serves as a platform for the Dragon solar arrays and radiator. The trunk remains attached to Dragon throughout launch and ISS attached operations but is jettisoned shortly before reentry. Unpressurized cargo that is both mounted directly or by FRAM can be accommodated by the cargo rack within the trunk. Disposal cargo may be stowed in the trunk for jettison prior to reentry.

Volume available for cargo in the trunk, accounting for static and dynamic clearances, is shown in Figure 5.2.2-1. Total volume is 601 cu ft (17.0 m³). The volume is defined from the trunk cargo rack; the cargo rack is not included in the volume. Allowable payload volume is shown in gray. Keep-out zones are denoted in purple and inside the dashed lines. A detailed CAD model of the trunk is available for volumetric assessments on NASA CAD Model team website/Enterprise Document Management System (EDMS).

Dragon trunk will nominally provide no more than an average of 200W for the total external cargo manifest. Overall external cargo mass shall not exceed 3000 kg.

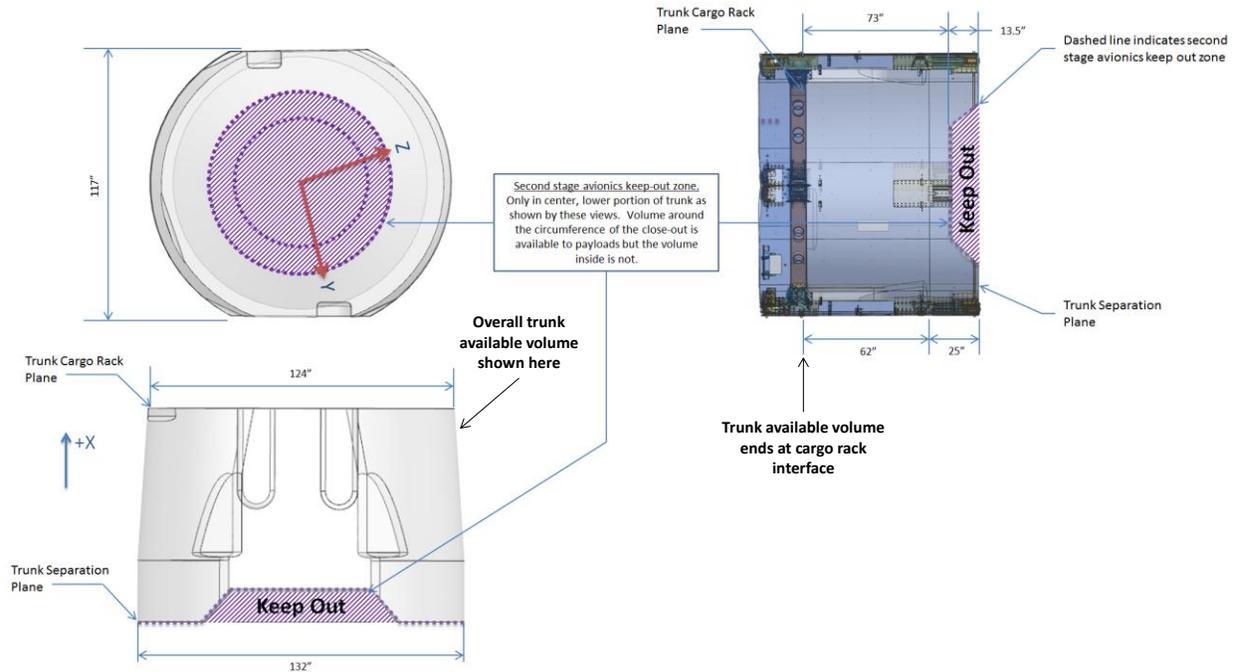


Figure 5.2.2-1. Dragon specifications.

5.2.2.1 Cargo interface

The types of Cargo interfaces to the Dragon include:

- a) ExPA and CEPA payloads.
 Example: High Definition Earth Viewing (HDEV), Optical Payload for Lasercomm Science (OPALS). The Space Test Program – Houston 4 (STP-H4)
- b) JEM-EF payloads (Direct Mount Cargo).
 Example: CATS.
- c) Other Direct mount unpressurized cargo. (See Figure 5.2.2.1-1.)
 Example: Bigelow Module

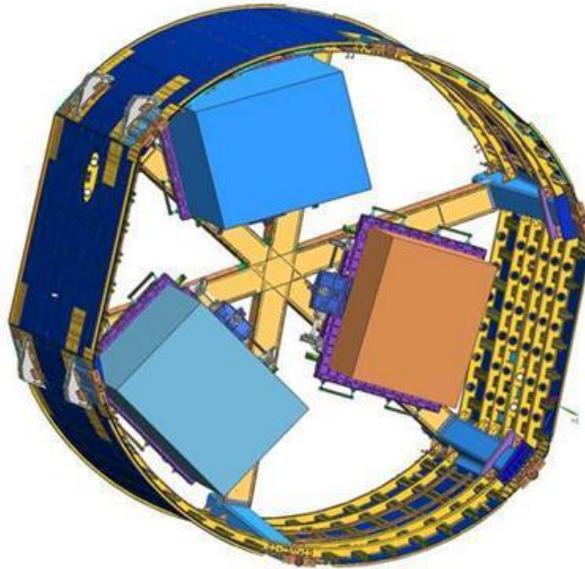


Figure 5.2.2.1-1. Three FRAM-based unpressurized cargo items.

5.2.2.2 Customer Accommodations (Power):

The Dragon visiting vehicle provides power accommodations in the trunk area of the visiting vehicle. This is where external payloads will be located. The power available, primarily for survival power, is up to 100 W continuous for all payloads at 120 Vdc (+6/7 V) and 500 W peak at 28 Vdc. This accommodation is defined in SSP 57012.³⁹

Thermostats for power/payload heaters will only turn on when the temperature is below $-30\text{ }^{\circ}\text{C}$ ($-22\text{ }^{\circ}\text{F}$), based on accommodations defined in SSP 57012.³⁹

5.2.2.3 Integration Schedule (Example)

A typical Falcon 9 integration schedule (starts at $\sim\text{L}-59$ days) is contained below:

1. Payload is vertically integrated at a Payload Processing Facility (PPF) — Cape Canaveral, for example.
2. Payload and vehicle adaptor are attached over a “break-over” fixture at the PPF.
3. The payload is rotated horizontally and encapsulated by the fairing. At this point, the payload is no longer fully accessible, and can only be accessed by an 18 x 18-inch door on the bottom of the fairing.
4. The encapsulated payload is transported to the launch complex. During this process, it remains completely horizontal for easier integration.
5. The payload and launch vehicle/visiting vehicle are integrated mechanically inside the hangar. Mechanical integration is done first to ensure payload meets the physical constraints of the vehicle and launch procedure.
6. Once the mechanical mating is confirmed, the electrical umbilical connections are established between the payload and the second stage. If necessary, electrical connections are also made between the ground support equipment and the payload.
7. The electrical interface is reassessed. Once it is verified again, the launch vehicle/visiting vehicle is transferred to the launch pad and put into a vertical position. At this point, the

- payload can no longer be accessed. For a short time during this process, conditioned air will be unavailable as new connections are made.
8. At L-7 days, the Flight Readiness Review takes place to ensure no major problems in the launch procedure.
 9. Final connections between ground support, vehicle, and launch pad are verified.
 10. The pre-launch environmental controls are terminated and the vehicle is launched.

Note: The most recent SpaceX user's guide was issued in 2009, and gives technical details for the Falcon 9 v1.0 rocket. That rocket has since been retired and replaced by the Falcon 9 v1.1 rocket. However, these two vehicles have similar configurations and processes which the user's guide outlines. Vehicle-specific information, such as load characteristics, is inaccurate and should be obtained from your PIM or SpaceX directly.

PLEASE NOTE: The PD should contact the ISS Research Integration Office for the latest Interface Definition Document (IDD) to obtain up-to-date launch loads and other interface and accommodations parameters.

For example, the following, Table 5.2.2-1, provides interface launch loads with the launch vehicle/visiting vehicle, as contained within SSP 57012³⁹. Also, the random vibration environment, Table 5.2.2-2, is provided below, as contained within SSP 57012³⁹.

Table 5.2.2-1. Interface launch loads.

Event	Interface	Force (lbs) ⁽¹⁾	Direction
Liftoff/Landing Limit Loads	Interface Brackets, Mechanism Pins	±10,000 ⁽²⁾ (±44482.2 N)	Normal to FSE Adapter Plate
	Fixed Shear Pin	±11,000 ⁽²⁾ (±48930.4 N)	Transverse Shear
	Slotted Shear Pins	±10,000 ⁽²⁾ (±44482.2 N)	Transverse Shear
Flight Quasi-Static Limit Loads	Interface Brackets, Mechanism Pins	±2,850 ⁽²⁾ (±12677.4 N)	Normal to FSE Adapter Plate
	Shear Pins	±2,950 ⁽²⁾ (±13122.3 N)	Transverse Shear
SSV On-Orbit Transient Limit Loads	Interface Brackets, Mechanism Pins	±450 ⁽²⁾ (±2001.7 N)	Normal to FSE Adapter Plate
	Shear Pins	±500 ⁽²⁾ (±2224.1 N)	Transverse Shear
ISS On-Orbit Transient Limit Loads	Interface Brackets, Mechanism Pins	±300 ⁽²⁾ (±1334.5 N)	Normal to FSE Adapter Plate
	Shear Pins	±400 ⁽²⁾ (±1779.3 N)	Transverse Shear
ISS On-Orbit Robotic Translation Limit Loads (Modified Micro-Square interface restraint)	Modified Micro-Square interface centroidal forces	±50, with a ±1,500 in-lb moment (±222.4 N, with a ±169.5 N-m)	Moments are bending and torsion, forces are shear and tensile/compressive, resultant forces and moments can be applied simultaneously in any direction.

Event	Interface	Force (lbs) ⁽¹⁾	Direction
ISS On-Orbit Robotic Translation Limit Loads (SGI interface restraint)	SGI centroidal forces	±500, with a ±12,000 in-lb moment ⁽⁴⁾ (±2224.1 N, with a ±1355.8 N-m)	Moments are bending and torsional, forces are shear and tensile/compressive, resultant forces and moments can be applied simultaneously in any direction, loads are measured at the OTP to passive SGI plane, centered on the passive SGI mounting plate.
ISS On-Orbit Transient Limit Loads (SGI interface restraint)	SGI centroidal forces	±280, with a ±5,700 in-lb moment ⁽³⁾ (±1245.5 N, with a ±644 N-m)	Any direction, one direction at a time.
ISS On-Orbit Crew-Applied Limit Loads (SGI interface restraint)	SGI centroidal forces	±125, with a ±6,250 in-lb moment (±556 N, with a ±706.2 N-m)	Any direction, one direction at a time.
Emergency Landing Ultimate Loads	Interface Brackets, Mechanism Pins	±2,650 (±11787.8 N)	Normal to FSE Adapter Plate
	Shear Pins	±4,200 (±18682.5 N)	Transverse Shear

Table 5.2.1-2. Random vibration environment.

Frequency	Level
20 - 38 Hz	0.0025 g ² /Hz
30 - 80 Hz	+7.94 dB/oct
80 - 400 Hz	0.0178 g ² /Hz
400 - 2000 Hz	-3.55 dB/oct
2000 Hz	0.00267 g ² /Hz
Composite	4.01 g root mean square (rms)
Duration	60 seconds

Notes:

1. This environment envelopes the maximum flight random vibration environment for HTV and the equivalent random vibration fatigue level for 60 second duration for SpaceX Dragon. The equivalent fatigue level accounts for the SpaceX Dragon random vibration environment whose duration is not 60 seconds. A fatigue exponent of 4 was used in the equivalent fatigue level computations.
2. These values envelope the HTV and SpaceX Dragon random vibration environments and flight durations, which can be found in Table D.2.3-1, Maximum Random Vibration Environments, and Figure D.2.3-1, Maximum Random Vibration Environments.
3. Criteria are the same for all directions (X, Y, Z).

5.2.3 Analytical Data Products

Range safety requirements are contained in the Flight Facility Range Safety Manual and in AFSPCMAN 91-710. In addition, the SpaceX safety process consists of (primarily) a “checklist,” which requires the PD to provide proof that safety requirements have been met. It is considerably less complex than that required for ISS flight and ground safety, but uses inputs from the safety packages as further proof of a payload’s launch readiness. However, the safety

process for Falcon 9 launches is still evolving, so the PD should meet with their PIM and SpaceX personnel, as soon as the PD's schedule allows, discuss more specific Falcon 9 safety information and requirements. Some additional information for safety requirements can also be found in Section 7 of Reference 41.⁴¹

In addition, the PD must develop a launch vehicle/visiting vehicle ICD with the launch services provider. In current cases, the ICD will be written between the PD and SpaceX. Specific, detailed interfaces will be contained within the LV ICD. At this time, the PDs should also continue to use SSP 57012³⁹ for interface requirements to the launch vehicle/visiting vehicle. In addition, the PD must develop a launch vehicle/visiting vehicle ICD with the launch services provider. In current cases, the ICD will be written between the PD and SpaceX. Specific, detailed interfaces will be contained within the Launch Vehicle ICD. At this time, the PDs should also continue to use SSP 57012³⁹ for interface requirements to the launch vehicle/visiting vehicle.

5.2.4 Mission Profile (Example)

The following represents a typical mission profile of the Falcon/Dragon launch to ISS.

1. Delivery: Delivery of integrated Payload to SpaceX PPF or launch site LC-40.
2. Launch Processing: After handover to SpaceX, all Launch Site Processing for Payload including integration to Cargo Rack in Dragon Trunk.
3. Launch: Launch of Dragon carrying Payload on Cargo Rack in Dragon Trunk.
4. Free Flight to ISS: Dragon Free Flight carrying Payload to ISS on Cargo Rack in Dragon Trunk.
5. Berthing: Dragon berthing to ISS carrying Payload on Cargo Rack in Dragon Trunk.
6. On-orbit Extraction: SPDM or SSRMS extracting Payload from Cargo Rack in Dragon Trunk.
7. Cargo Work Site Operation: Payload transported to ISS location by SPDM or SSRMS. (This phase does not involve SpaceX or Dragon.)
8. Payload Disposal: TBD Payload transported back in Dragon Trunk for Disposal.
9. De-berthing: Dragon de-berthing from ISS.
10. Departure: Dragon departure from ISS and Free Flight.
11. Reentry: Dragon reentry Disposal of Dragon Trunk.
12. Recovery: Dragon recovery.

6.0 Flight and Ground Safety Process

The single most important process that a PD will utilize and go through is the Safety Process. It is the safety process, both ground and flight, that ensures the safety for the human-rated ISS, and the safety of ground support personnel, and equipment. This section outlines the ISS safety process, while Section 5.0, launch and visiting vehicles, describes the launch vehicle/visiting vehicle process for safety.

Once a PD starts the Payload Integration Process, the PD should request, through their PIM, an early safety assessment for their payload. Subsequent to that meeting, a series of ground and flight safety reviews will be negotiated with the PD that will be dependent on the complexity, technical maturity, and hazard potential of the PD's payload. During the payload design/development process, the PD will be required to produce a Safety Data Package (SDP) by performing a safety analysis of their payload and GSE in parallel with their design review milestones. During the safety process, a Payload Safety Engineer (PSE) will be assigned to the payload that will assist the PD through the process.

The PD has two major safety elements that will require a substantial amount of resources. These elements are flight safety and ground safety. Finally, Reference 34 provides additional primer information on the safety process.

6.1 Flight Safety

As pointed out in Section 4.1, Integration Template, a PD will provide a flight SDP, with updates, at each of the following milestones:

- Concept design phase (Phase 0): PDs should include a Phase 0 safety review (hazard causes defined) in their payload's schedules.
- PDR (Phase I): PDs should complete the Phase I safety review activities (hazard causes refined and hazard controls defined with preliminary verifications). In many cases, Phase 0 and Phase I may be combined into a single package, due at PDR, based on specifics of the payload design.
- CDR (Phase II): PDs should complete the Phase II safety review to identify any design changes, new hazard causes and controls, and well-defined safety verification methods.
- 30 days prior to delivery (Phase III): The final flight safety review is called Phase III and must be completed 30 days prior to delivery of the payload, Airborne Support Equipment, and GSE to the launch site. The flight certificate of safety compliance shall be submitted at least 10 days prior to the Stage Operations Readiness Review. The Phase III review is focused on completion of hazard report safety verification method.
- Post Phase III Safety Review and Safety Verification Tracking Log: When the flight certification statement is submitted, it shall be included with an updated payload safety verification tracking log that documents the closeout of all required safety verification, and includes all post-Phase III safety activity.

Guidelines for the development of the critical Safety Packages can be found in JSC 26943⁴².

6.2 Ground Safety

The PD's major effort for ground safety will be developing the Ground Safety Data Package. The Ground Safety Review Panel (GSRP) recognizes that the initial submission may be difficult,

but the PSE will work with the PD to correct any errors and missing information in order for the PD to complete the review process successfully.

The timing of ground safety review is critical to the start of ground operations. According to NSTS/ISS 13830, Rev. C⁴³, submission of the SDP is due 45 days prior to the safety review. The safety review must be completed 30 days prior to the start of ground operations. This means that the package must be submitted 75 days prior to ground operations.

Additionally, the payload's flight safety reviews must be completed 30 days prior to the start of ground operations. Failure to do so will result in the delay of operations and possible launch date impact.

6.3 Safety Data Packages

The SDP contains two parts:

- Part one describes the payload, its systems, sub-systems, and interfaces, as well as flight and ground operations. It also summarizes hazard analyses used in the identification and control of payload hazards.
- Part two contains a series of hazard reports. The hazard report is used to summarize controls and verifications to ensure compliance to safety requirements. Elements of a hazard report include technical requirement references, description, and category of each hazard, and the hazard's cause, controls, and safety verification methods.

The primary objectives of the safety review process are to identify the potential payload hazards, including its flight, GSE, and ground operations, and to assure that the hazard controls and verifications (including on-orbit verification of hazard controls where applicable) comply with the safety requirements. The following list provides examples of previously identified flight hazards: contamination, corrosion, electrical shock, explosion, fire, injury and illness, loss of vehicle capability, etc. This list is NOT exhaustive, but provides a sense of the importance of flight safety.

The following represent examples of ground hazards: The following are basic hazard groups applicable to ground operations: structural failure of support structures and handling equipment; collision during handling; inadvertent release of corrosive, toxic, flammable, or cryogenic fluids; inadvertent activation of ordnance devices; ignition of flammable atmosphere/material; electrical shock/burns; personnel exposure to excessive levels of ionizing or nonionizing radiation; use of hazardous GSE materials; etc.

In all cases, the PD should use established analytical techniques, such as preliminary hazard identification, sneak circuit, fault tree, operational hazard, and failure modes and effects analyses to obtain the data necessary to complete, present, and support payload hazard reports. The analysis is comprehensive and must consider hardware design, verification, testing, and flight/ground operations.

The PD's assigned PSE will assist throughout the development of the SDPs and corresponding analysis.

6.4 Requirements

6.4.1.1 Flight

The Payload Safety Process requirements are contained in Reference 43, which further defines the deliverables and safety review activities, and provides excellent detailed information of developing your SDPs, and submittal requirements. The safety requirements are too numerous to be listed in their entirety, so more specific technical requirements for flight safety can be found in the following:

- ISS Addendum, Safety Policy and Requirements for Payloads Using the ISS, NSTS 1700.7B ISS Addendum.⁴⁴
- Interpretation of NSTS/ISS Payload safety Requirements, NSTS/ISS 18798⁴⁵
- ISS Safety Requirements Document, SSP 50021⁴⁶
- Safety Review Process, SSP 30599⁴⁷

Reference 44 contains the ISS payload safety technical requirements, and provides ISS references for the applicable core NSTS 1700.7B requirements.

6.4.2 Ground

KHB 1700.7⁴⁸, details the ground processing requirements for ISS payloads. There are two key data packages that must be completed by the PD, related to the ground safety process. The first is the Ground Certificate of Safety Compliance, where the PDs certify that they are in compliance with the Ground Safety Requirements and the second is the Ground SDP, which provides the data supporting the certification.

Upon approval of the SDP and acceptance of the Certificate of Compliance, the GSRP will issue letters of approval for the start of operations. Sometimes, at this point, the PD will have a Safety Verification Tracking Log with open items that are constraints to ground operations. The PD needs to work closely with the PSE to ensure the proper operations are constrained and that the items are closed in a timely manner.

6.5 Safety Identification and Analysis

The PD is responsible for performing safety analysis for each payload, its GSE, related software, and ground and flight operations to identify hazardous subsystems and functions. A safety assessment report documents the results of this analysis, including hazard identification, classification, resolution, and a record of all safety-related failures. This report shall be prepared, maintained, and submitted in support of the safety assessment reviews.

Hazards are classified as either critical or catastrophic:

- Critical hazards can result in damage to ISS equipment, or a non-disabling personnel injury.
- Catastrophic hazards can result in the potential for a disabling or fatal personnel injury, loss of the ISS, ground facilities, or equipment.

Fault tolerance may also be a cost driver for the PD. It should be noted that catastrophic hazard mitigation requires 2-fault tolerance for the system, subsystem, or elements identified as such. In all cases, the PD's major goal throughout the design phase shall be to ensure safety through

the selection of appropriate design features. Damage control, containment, and isolation of potential hazards shall be included in design considerations.

Actions for reducing hazards should be conducted in the following order of precedence:

- Design for Minimum Hazard – The PD should develop appropriate design features, including damage control, containment, and isolation of potential hazards.
- Safety Devices – Hazards which cannot be eliminated through design shall be made controllable through the use of automatic safety devices as part of the system, subsystem, or equipment.
- Warning Devices – When it is not practical to preclude the existence or occurrence of known hazards or to use automatic safety devices, devices shall be employed for the timely detection of the condition and the generation of an adequate warning signal, coupled with emergency controls of corrective action for operating personnel to safe or shut down the affected subsystem. Warning signals and their application shall be designed to minimize the probability of wrong signals or of improper reaction to the signal.
- Special Procedures – Where it is not possible to reduce the magnitude of a hazard through design or the use of safety and warning devices, special procedures shall be developed to counter hazardous conditions for enhancement of personnel safety.

It must be stressed that designing for hazard elimination or reduction is the preferred method, by far, of hazard mitigation.

6.6 Roles and Responsibilities

The PD's key to the overall process is early contact with Payload Safety Review Panel (PSRP) Executive Officers. Once a candidate payload is identified and requesting PSRP review, a PSE is assigned, who will advise the PD through the entire safety process. The PD is responsible for completion of all SDP submittals, and providing verification that all safety data requirements have been met.

6.6.1 Flight Safety

The PSRP is an ISS Safety Review Panel (SRP) located at JSC, and performs the following functions:

- Assist the PD in the interpretation of safety requirements
- Conduct safety reviews during appropriate phases of the payload development
- Evaluate hazard assessment revisions resulting from payload modifications that may affect a safety-critical subsystem or create potential hazards to the crew, ISS, or ISS visiting vehicles
- Evaluate the safety analyses, safety reports, and waiver/deviation requests
- Ensure the resolution of payload safety issues

Although it is the responsibility of the PSRP to audit the PD safety assessments, the PD is responsible for the safety of the payload for all mission phases. The PD is also responsible for not compromising the safety of other payloads, launch vehicle/visiting vehicle that are used to transport the payload, ISS transfer, and ISS operations.

6.6.2 Ground Safety

The GSRP is the ISS SRP located at KSC. The purpose of the GRSP is to ensure that all ground safety requirements are met and that all applicable ground requirements are met.

The GSRP process parallels that of the PSRP and is described in Reference 45. The significant difference between the panels is that the GSRP generally only requires one safety review. The key to this process is early contact with the Lead Ground PSE for the flight assigned.

6.6.3 Ground Safety at International Partners Launch Sites

The process described in this section applies to processing at KSC only. If the payload is to be launched to the ISS from an international partner launch site (French Guiana, Japan, Russia), compliance with those launch site's process is required. NASA will assist the PD in completing the appropriate process.

6.6.4 Ground Safety at Non-KSC U.S. Launch Sites

For launch at non-KSC U.S. launch sites, the PD shall provide data to the launch vehicle/visiting vehicle provider. If the PD requires processing at a non-KSC launch site, NASA will assist in completing the appropriate process.

6.6.5 Summary

The PD should always have the safety process foremost in their minds and in their designs for the payload, ground equipment, airborne support equipment, etc. The safety process is rigorous and necessary, and no payload will be flown until ALL safety requirements, data submittals, and certification has been completed and approved by the respective safety boards.

7.0 ISS Program-Provided Hardware

7.1 ISS-provided Hardware

The ISS program provides accommodation hardware to PDs for attaching and interfacing to the ISS.

The ISS payload adapters and interface units are GFE and are part of the standard services provided by ISS, and are generally provided to the PD at no cost.

These include:

- ExPA for flight on the ISS truss (ELC), including the FRAM
- CEPA – similar to the ExPA
- PIU for flight on JEM-EF

Coordinated Services are specific services negotiated between the PD and the ISS program. This requires an agreement documented in a PIA. The PIA documents what services are provided and who funds those services.

Coordinated Services (examples):

- GSE shipped for use at PD's location
 - low-profile dollies
 - lifting slings
 - hooks
- Payload interface connectors
 - electrical (power and data) – all sites
 - fluid (JEM-EF)
- Payload interface hardware from payload to ExPA adapters or interface units (or for Columbus, CEPA adapters or interface hardware)

7.2 ISS-Provided Simulators

To assist in the development of the payload, the ISS program can provide small test equipment that emulates the ISS side of the interface, specifically for payload development. These systems are portable and can be shipped to the PD's location. These systems are referred to as the Suitcase Test Environment for Payloads (STEP) and may be shipped to the PDs facility (see Figure 7.2-1 and Figure 7.2-2). The STEPs are generally utilized at no cost to the PD.



Figure 7.2-1. Portable test environment for development and testing of ELC payloads.



Figure 7.2-2. ELC suitcase simulator.

An ELC Functional Equivalent Unit is a rack-mounted, high-fidelity functional unit of the ELC, and is located at MSFC. It is used by the Payload Software Integration Test Facility to verify ELC flight software and payloads during the verification phase of your project schedule. It may be used at no cost to the PD.

The ELC Simulator simulates the ELC interfaces and is located at KSC (see Figure 7.2-3). This unit is used for final verification of a payload's C&DH and power to the ISS. The ELC is used at no cost to the PD, and is operated primarily by ISS personnel.



Figure 7.2-3. ELC simulator.

7.3 International Partner Facilities

Columbus payloads MAY elect to be tested using the standard U.S. facilities or the Rack Level Test Facility in Bremen Germany. However, the Rack Level Test Facility will not generally be used by NASA PDs.

JEM-EF payloads are tested using the standard U.S. facilities or at the JAXA launch facility in Tanegashima, Japan. As is the case with Columbus payloads, the JAXA facility will not generally be used by NASA PDs.

In addition, various commercial options are available or in development for use in place of the ELC STEP (suitcase simulators). The Boeing Company has also developed a software simulator known as Remote Advanced Payload Test Rig, which is available for use by PDs.

7.4 Summary

When ISS test facilities or capabilities are requested, approval and scheduling the use of these valuable assets must be negotiated with your assigned PIM.

All in all, the ISS program now provides excellent interfacing tools, hardware, and software simulators, and on-orbit interfacing equipment that will prove to be invaluable in your payload life cycle. Consider the use of these valuable tools during your payload's schedule of events, both during development and verification.

8.0 Payload Developer-Provided Hardware

Items typically the responsibility of the PDs are items that are generally considered unique to the payload and are not included in either the ISS standard or coordinated services outline in Section 7.0. These items are generally the responsibility of the PDs to provide, and examples are outlined below.

Typical PD-funded Items for ISS interfacing include:

- Simple attach plates, and fasteners, connecting a payload to the ExPA or CEPA
- Hardware that attaches the payload to fluid lines from the JEM-EF PIU, including payload-internal plumbing, valves, quick/disconnects, and other connections on the payload-side of the PIU interface. The PD must procure the electrical, mechanical, fluid loop connectors on their side of the interface.
- JEM-EF platform or “box” containing the experiment, containing ISS funded PIU, trunnions, and latching mechanisms included
- Payload shipping containers
- Blankets (thermal)
- Battery charging
- Payload processing in other than SSPF, or international partner facilities, to accommodate payload requirements that cannot be met at the Space Station Processing Area.
- Specific requirements for environmental and/or contamination control that are non-standard for each of the KSC processing facilities
- Large quantities of consumables (i.e., gaseous nitrogen)
- On-orbit recording capability – data latency capability – with subsequent downlink, if required

This list is not intended to be exhaustive, and should be used for initial planning purposes only as a rule of thumb. As your payload design and development matures, coordination with your PIM is essential. During this process, the details of what you may require outside of ISS-provided hardware, and the example items list above, will become more focused.

Obviously, the “typical” hardware listed above may vary widely in costs. However, the fluid connections, on the payload side of the JEM-EF interface, could be substantial. These items should be carefully evaluated and included in your proposal.



Figure 8.0-1. FRAM stack-up with the heater power connector.

9.0 Acronyms

AFRAM	Active Flight Releasable Attach Mechanism
AMS	Alpha Magnetic Spectrometer
API	Advanced Publication Information
APM	Attached Pressurized Module
APS	Automated Payload Switch
ATCS	Active Thermal Control System
BAD	Broadcast Ancillary Data
BC	Bus Controller
C&C	Command and Control
C&DH	Command and Data Handling
C.G.	Center of Gravity
CAD	Computer Aided Design
CATS	Cloud-Aerosol Transport System
CCSDS	Consultative Committee for Space Data Systems
CDR	Critical Design Review
CEPA	Columbus External Payload Adapter
COF-EPF	Columbus Orbiting Facility-External Payload Facility
CoFR	Certificate of Flight Readiness
Columbus-EPF	Columbus-External Payload Facility
COTS	Commercial Off the Shelf
CSA	Canadian Space Agency
CST	Customer Service Team
CTRS	Conventional Terrestrial Reference System
CV	Controlled Volatility
DDCU	DC to DC Converter Unit
DMC	Data Management Coordinator
ECR	Engineering Change Request
EDMS	Enterprise Document Management System
EEPROM	Electrically Erasable Programmable Read Only Memory
EEU	Equipment Exchange Unit
EF	Exposed Facility
EFU	Exposed Facility Unit
ELC	ExPRESS Logistics Carrier
ELM	Experiment Logistics Module
EMC	Electro-magnetic Compatibility
EMEP	Electromagnetic Effects Panel
EMF	Electro-magnetic Force
EMI	Electro-magnetic Interference
EMR	Electro--magnetic Radiation
EOTP	Enhanced ORU Temporary Platform
EP	Exposed Pallet - Multi-Purpose
EPF	External Payload Facility
EPMP	Environmental Protection and Management Plan
ES	Exposed Section
ESD	Electrostatic Discharge

ESA	European Space Agency
ESP	External Stowage Platform
ESSP	Earth System Science Pathfinder
EV	Earth Venture
EVA	Extravehicular Activity
EVR	Extravehicular Robotic
EWC	External Wireless Communication
ExPA	ExPRESS Payload Adapter
ExPCA	ExPRESS Pallet Control Assembly
ExPRESS	Expedite the Processing of Experiments to the Space Station
FCT	Flight Control Team
FDDI	Fiber Distributed Data Interface
FEM	Finite Element Model
FGB	Functional Cargo Block
FOD	Flight Operations Directorate
FOV	Field of View
FRAM	Flight Releasable Attach Mechanism
FRGF	Flight Releasable Grapple Fixture
GCP	Ground Command Procedures
GDS	Ground Data Services
GF	Grapple Fixture
GFE	Government Furnished Equipment
GN&C	Guidance, Navigation and Control
GR&C	Ground Rules and Constraints
GSE	Ground Support Equipment
GSFC	Goddard Space Flight Center
GSRP	Ground Safety Review Panel
H&S	Health and Safety
HDEV	High Definition Earth Viewing
HIF	Horizontal Integration Facility
HOSC	Huntsville Operations Support Center
HRDL	High-Rate Data Link
HUB	Ethernet LAN Repeater
ICD	Interface Control Document
ICS	Inter-Orbit Communication System
IDD	Interface Definition Document
IP	International Protocol
IRD	Interface Requirements Document
IRP	Increment Research Plan
ISPR	International Standard Payload Rack
ISS	International Space Station
ITA	Integrated Truss Assembly
ITS	Integrated Truss Segment
IVT	Interface Verification Tests
JAXA	Japan Aerospace Exploration Agency
JEM-EF	Japanese Experiment Module-Exposed Facility

JEM-PM	Japanese Experiment Module-Pressurized Module
JSC	Johnson Space Center
JSL	Joint Station LAN
KSC	Kennedy Space Center
L	Launch
LAN	Local Area Network
LaRC	Langley Research Center
LEHX	Layer 2 Ethernet Switch and Multiplexer
LOS	Loss of Signal
LRDL	Low-Rate Data Link
LTA	Launch-To-Activation
LVLH	Local Vertical Local Horizontal
MA	Main Arm
MAC	Media Access Control
MAGIK	Manipulator Analysis Graphics and Interactive Kinematics
MBS	Mobile Base System
MCAS	MSS Common Attach System
MCF	Micro-Conical Fitting
MDM	Multiplexer/Demultiplexer
MDP	Maximum Design Pressures
MIMO	Multiple-Input and Multiple-Output
MISSE	Materials International Space Station Experiment
MLI	Multi-Layer Insulation
MOD	Mission Operations Directorate
MRDL	Medium Rate Data Link
MRPWG	Multi-lateral Research Planning Working Group
MSFC	Marshall Space Flight Center
MSS	Mobile Servicing System
MT	Metric Ton
NIC	Network Interface Card
OLR	Outgoing Long-Wave Radiation
OPALS	Optical Payload for Lasercomm Science
ORU	Orbital Replacement Units
OTCM	ORU/Tool Change Out Mechanism
OZ	ISS Research Office
P/L	Payload
PAS	Payload Attach System
PCS	Portable Computer System
PD	Payload Developer
PD-POC	Payload Developer Payload Operations Center
PDR	Preliminary Design Review
PDU	Power Drive Unit
PEHG	Payload Ethernet Hub Gateway
PFRAM	Passive Flight Releasable Attach Mechanism
PIA	Payload Interface Agreement
PIM	Payload Integration Manager

PIRN	Preliminary Intermediate Revision Notices
PIU	Payload Interface Unit
PM	Pressurized Module
POC	Payload Operations Center
POIC	Payload Operations and Integration Center
POIF	Payload Operations Integration Function
PPF	Payload Processing Facility
PPM	Payload Planning Manager
PPSB	Payload Power Switching Box
PRCU	Payload Rack Checkout Unit
PSE	Payload Safety Engineer
PSRP	Payload Safety Review Panel
PTCS	Passive Thermal Control System
PVGF	Power and Video Grapple Fixture
RF	Radio Frequency
RIO	Research Integration Office
RMCT	Robot Micro-Conical Tool
RMS	Remote Manipulator System
RNP	Rotation, Nutation and Precession
RPCM	Remote Power Controller Module
RS	Radiated Susceptibility
RT	Remote Terminal
RTD	Resistance Temperature Detectors
RTV	Room Temperature Vulcanization
SAGE III	Stratospheric Aerosol and Gas Experiment III
SALMON	Stand Alone Missions of Opportunity Notice
SDP	Safety Data Package
SFA	Small Fine Arm
SLM	Structural Latch Mechanism
SPDA	Secondary Power Distribution Assembly
SPDM	Special Purpose Dexterous Manipulator
SPOC	SAGE-III Payload Operations Center
SRP	Safety Review Panel
SSPF	Space Station Processing Facility
SSRMS	Space Station Remote Manipulator System
STEP	Suitcase Test Environment for Payloads
STP	Space Test Program
TBS	To Be Supplied
TDRSS	Tracking and Data Relay Satellite System
TEA	Torque Equilibrium Attitude
TQCM	Thermally controlled Quartz Crystal Microbalances
U.S.	United States
UCCAS	Unpressurized Cargo Carrier Attachment System
UCM	Umbilical Connect Mechanism
UMA	Umbilical Mechanism Assembly
USB	Universal Serial Bus

USL	United States Laboratory
VDPU	Video and Data Processing Unit
VTL	Verification Tracking Log
WAP	Wireless Application Protocol
WSC	White Sands Complex
XCMU	External Command and Monitoring Unit

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